U.S Department of Transportation Federal Aviation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

MAMES TERRORIES CO

Administrati	ion									WI	9= F(S) D)(O)=1		
and dis		form. This	s repo	rt is required t	y law (49	U.S.C		2 43.9-1 (or subsequence report can result in		thered	of) for instructions		
	Natior USA	nality and R	-	ation Mark				Serial No.	0F				
1. Aircraft Make MD HELICOPTERS INC.								Model 369			Series E		
Name (As shown on registration certificate)								Address (As showr	on registra	istration certificate)			
2. Owner CITY OF OAKLAND						_		Address 4557		ET			
	POI	ICE DE	PT. I	HELICOPT	ER UNI	T		City OAKLAN			State CA		
								Zip 94607-394	10	Cou	ntry USA	_	
						PLIC		THINESS REQU			D IS APPROVED ONLY FOR IZED IN FAR 43.7.		
FAA INS	PECTOR, V	AN NUYS	S FSI	oo: Sern	16 B	es	Pu		[DATE:	SEP 2 6 2018		
4	. Туре						5. 1	Init Identification					
Repair	Alteration		Ur	nit		ľ	∕/ake	Mod	el		Serial Number		
	□ ⊠ AIRFRAME			_			(As described in Item 1 above)						
	POWERPLANT						-						
		PROPE	LLE	R									
		APPLIA	ANCE		Type Manufactu	Type Manufacturer							
	-				l	6	. Conformity Sta	tament			·	_	
A. Agency	's Name and A	ddress					and of Agency	ement					
Name_RC	TORCRAF	T SUPP	OR	Γ, INC.			U.S. Certificated	Mechanic	C	Man	nufacturer	_	
	425 HART	STREE	Ĺ				Foreign Certifica	ed Mechanic	C. Certificate No.				
	N NUYS			State_CA		⊠	Certificated Repa	air Station Y7			T2R331L		
Zip91	<u>406</u>	Country_	USA	<u> </u>			Certificated Main	tenance Organizatio	on				
have		n accordan	ce wit	h the requirem	ents of Pa	irt 43 (and described on the Aviation Regulation					
Extended raper 14 CFF App. B				Signature/I 26 SEP			ed Individual PHILLIP G. DIF	FIORE	110	ιŢ	FQ		
·						7. Ap	proval for Return	to Service	10_				
	t to the authori rator of the Fe			•	_	_		inspected in the ma EJECTED	nner prescri	bed by	the		
ВУ	FAA FIt Sta	indards		Manufacture	r 		Maintenance Org	anization		Dep	son Approved by Canadian partment of Transport		
	FAA Desig	nee	х	Repair Statio	on		Inspection Autho	rization	Othe	er (Spe	cify)		
Certificate of Designation YT2R33	n No.			sture/Date of SEP 2018	Authorized		idual LLIP G. DIFIC	PRE W	1()	Tu			

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

	Description of Work Accomplished If more space is required, attach additional sheets. Ide	ntify with ai	roraft nationality and registra	tion mark and date work completed)						
	n more space is required, allacin additional sheets. Tue.	U.S.A.	N510PD	26 SEP 2018						
		L								
	·	Nationali	ty and Registration Mark	Date						
1)	REMOVED THE FOLLOWING EQUIPMENT FROM THE AIR									
	a) AEROCOMPUTERS ULTICHART CPU MODULE S/N 2									
	b) AEROCOMPUTERS KEYBOARD P/N AKA-4 S/N 2151	2 WITH MOU	IN I							
	c) AEROCOMPUTERS GPS ANTENNAd) AEROCOMPUTERS HAND CONTROLLER INTERFAC	E LINUT								
	e) AEROCOMPUTERS X3 DVR S/N 73323	CONT								
	f) LO-JACK RECEIVER & ANTENNA ARRAY; THE DISP	LAY WAS PE	REVIOUSLY REMOVED							
2)	A CHURCHILL NAVIGATION, AUGMENTED REALITY SYST			CONSISTS OF AN:						
-,	a) ARS 600 PROCESSOR S/N 160411, INSTALLED BELC BULKHEAD WALL, IN THE PASSENGER COMPARTM	OW THE EXI								
	b) NOVATEL IMU-H62, INERTIA MEASUREMENT UNIT, P/N 01017931 S/N NCG17180004, MOUNTED ON TOP OF THE EOUIPMENT SHELF.									
	c) NOVATEL FLEXPAK6, GNSS RECEIVER, S/N 805007		• •							
	d) GPS ANTENNA INSTALLED ON THE EXISTING MOUI									
3)	INSTALLED A VIDEO ROUTING SYSTEM WHICH INTERFA FOLLOWING UNITS INSTALLED ON TOP OF THE EOUIPM	ENT SHELF:		OR. THE SYSTEM CONSISTS OF THE						
	a) AJA VIDEO SYSTEMS HD10A-PLUS - HD/SD ANALOG TO DIGITAL CONVERTER.									
	 b) TWO DECIMATOR DESIGN MD-HX, - (3G/HD/SD)-SDI / HDMI CROSS CONVERTERS; c) AJA VIDEO SYSTEMS ROI-DVI – DVI-D TO 3G-SDI WITH REGION OF INTEREST SCALING CONVERTER. 									
4\	,		•							
4)	A LO-JACK POLICE TRACKING COMPUTER 3 WAS INSTA MOUNTED TO THE BELLY, LO-JACK TRACKING INFORMA			NNEL AND AN OPDATED ANTENNA ARRAY						
5)	POWER IS PROVIDED FROM THE MISSION BUSS, THROU			IF UNITS						
6)	A 50 AMP RELAY WAS MOUNTED TO THE LOWER OUTBO									
7)	THE INSTALLATION WAS PERFORMED REFERENCING: C									
•	SCHEMATIC; CHURCHILL NAVIGATION, PASADENA ARS6	00 VIDEO RO	OUTING DIAGRAM, REVISION N	IOV 13 2017; CHURCHILL NAVIGATION, BASI						
	ARS600 / FLIR 8500 SYSTEM DIAGRAM, REVISION 01/07/20	015; CHURC	HILL NAVIGATION, BASIC ARSE	600 / FLIRU8XXX VIDEO DIAGRAM, REVISION						
	23 FEB 2016; AJA VIDEO SYSTEMS HD10A-PLUS INSTALL	ATION & OP	ERATION GUIDE, VERSION 1.0	r1; AJA VIDEO SYSTEMS ROI-DVI						
	INSTALLATION & OPERATION GUIDE, VERSION 1.5; DECI									
	DRAWING NO. D0114395400, REVISION C; NOVATEL OEM	•	•							
	REVISION 12; MD HELICOPTERS, BASIC HANDBOOK OF N									
	96 - ELECTRICAL AND CHAPTER 98 - WIRING DIAGRAMS; WEIGHT AND BALANCE; CHAPTER 11, PARA 11-30 TO 11-									
	TO 11-51 – CIRCUIT BREAKERS; PARA 11-66 TO 11-69, 11-									
	INSTALLATION; PARA 11-106 & 11-107 - EMI & INTERFERE									
	MOVEABLE CONTROLS PRECAUTIONS; PARA 11-135 TO		·							
	STRIPPING AND LACING; PARA 11-167 - SPLICING; PARA									
	PARA 11-230 TO 11-233 CONNECTORS; AC 43.13-2B, DAT	ED MARCH 2	2008, CHAPTER 1, PARA 100 TO	O 113 (STRUCTURAL DATA); CHAPTER 2,						
	PARA 200 TO 207 (COMM / NAV INSTALLATIONS), AND CH	IAPTER 3, PA	ARA 300 TO 309 (ANTENNA INS	STALLATION).						
8)	SATISFACTORY GROUND CHECKS WERE PERFORMED.									
9)	WEIGHT & BALANCE AND EOUIPMENT LIST WAS REVISED	D.								
10)	THE MAXIMUM CONTINUOUS ELECTRICAL LOAD DOES N									
11)	INSTRUCTIONS FOR CONTINUED AIRWORTHINESS: PER									
	ASSOCIATED WIRING, CABLES, CONNECTORS, HARDWA			'						
	SECURITY, WEAR, CHAFING, ETC. SPECIAL ATTENTION S		SIVEN TO THE AIRCRAFT PRIM	MAKY STRUCTURE WITH						
131	REGARDS TO FATIGUE AND STRESS, CRACKING, CORRO		DDED 10050							
12)	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC. UNDE									
	NOTHIN	O FOLLOWS	, 							

U.S Department of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Transportatio	on	()	AIITI	ame, Pov	verpian	it, P	'ro	peller, or	Appliance)			For FAA Use Only	
Federal Aviation													
and disp		form. This	repo	rt is required t	y law (49	U.S.C			C 43.9-1 (or subseque o report can result in a				
	egistr 10PE	ation Mark					Serial No. 0400E						
1. Aircraft	Make MD H	HELICOF	PTE	RS INC.					Model 369E			Series	
				egistration cert					Address (As shown on registration certificate)				
2. Owner		OF OAL		ND IELICOPTE	- D				Address 455 7 TH S	TREET		••	
	UNIT		1.1	ILLIOO! II	_1\			City <u>OAKLAND</u> State <u>CA</u> Zip <u>94607-3940</u> Country <u>USA</u>					
				 		3	3. F	or FAA Use	zip <u>94607-3940</u> Only		ountry	<u>/ USA</u>	
4,	Type	1			1			5.	Unit Identification		Т-		
Repair	Alteration	<u> </u>	Ur	nit		٨	Mak	ke	Model			Serial Number	
	\boxtimes	AIRFRAME							(As described in It	em 1 above)		<u></u>	
		POWE	RPL/						_				
		PROPE	LLE	R									
		APPLIA	ANCE		Type Manufacturer			ne					
					Manada								
		•				6	6. C	onformity St	atement				
	s Name and A		_			В. К	Kind	of Agency					
	TORCRAF			<u>r, inc.</u>			U	.S. Certificate	Mechanic			ufacturer	
	425 HART	STREE	<u>[</u>			☐ Foreign Certificated			ated Mechanic			icate No.	
	N NUYS			State_ <u>CA</u>		Ø	┿	ertificated Rep			2R3	31L	
Zip914	<u>406</u>	Country	<u>USA</u>	1			l C	ertificated Mai	ntenance Organization				
have		accordan	ce wit	h the requirem	ents of Pa	rt 43 c			e and described on the al Aviation Regulations				
Extended ra per 14 CFR App. B				Signature/I 26 JUNE				ndividual P.G. DIFIO	RE U	10=	Ţ(\mathcal{I}	
				٠		7. Ap	ppro	oval for Retur	n to Service	()	<u>u</u> _		
i e	to the authorit					_			s inspected in the man	per prescribe	d by	the	
	FAA Fit Sta	ndards		Manufacture	ı	·	м	aintenance O	ganization			on Approved by Canadian artment of Transport	
BY —	FAA Design	nee	х	Repair Statio	on		Inspection Authorization			Other	(Spec	sify)	
Certificate of Designation	n No.		1 -	nature/Date of JUNE 201					0001	() =	2/)	

8.	Description of Work Accomplished		
	(If more space is required, attach additional sheets. Ide	ntify with aircraft nationality and registration	mark and date work completed.)
		USA N510PD	26 JUNE 2017
		Nationality and Registration Mark	Date
1.	REPLACED AFT ENGINE COMPARTMENT HEAT BI CASCADE STC, LLC SUPPLEMENTAL TYPE CERTI NUMBER CAR-5045 REVISION 4 DATED 06/25/12.		
2.	AN ACTUAL WEIGHT AND BALANCE WAS PERFORM	RMED AND THE EQUIPMENT LIST UPDAT	ED.
3.	A LOGBOOK ENTRY HAS BEEN MADE.		
4.	CONTINUED AIRWORTHINESS IS ASSURED THRO INC DOCUMENT NUMBER CAR-5045 REVISION 4 D		PER CASCADE AIRFRAME REPAIR,
5.	DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN		
		HING FOLLOWS	^^
l			
	□A	dditional Sheets Are Attached	

U.S Department of
Transportation
Federal Aviation

Form Approved OMB No. 2120-0020 11/30/2007

Federal A Administ	Aviation		(24	XII I I I	anie, Pov	rei þiai:	ıı, r	ropener, or	Арриансе)		For FAA Use Only		
and	disposition	of this fo	orm. This	repo		y law (49	U.S.C		C 43.9-1 (or subsequen o report can result in a c				
	National USA	ity and Re N51		ation Mark				Serial No. 0400E					
1. Aircra		Make MD HE	ELICOF	TEF	RS INC.		-		Model 369E				
2. Own	or i	CITY	OF OAK	(LAI	egistration cert ND IELICOPTE				Address (As shown on registration certificate) Address 455 7 TH STREET CityOAKLAND State CA				
		OIVII						3. For FAA Use (zip <u>94607-3940</u>	Со	untry <u>USA</u>		
	., ., .,												
	4. Type							5.	Unit Identification				
Repa	ir Alte	ration		Ur	nit		N	<i>M</i> ake	Model		Serial Number		
		XI	AIRFRAME			_			(As described in Item 1 above)				
]	POWERPLANT										
			PROPE	LLEI	R								
	[APPLIA	NCE		Type Manufactu	ırer						
							6	. Conformity Sta	tement				
	ncy's Name						B. K	ind of Agency			· · · · · · · · · · · · · · · · · · ·		
	ROTORO				<u>r, INC.</u>		므	U.S. Certificated			Manufacturer		
	<u>16425 H</u> VAN NU`		IKEEI	•	chair CA		무	Foreign Certifica			ertificate No. R331L		
	91406		Country	USA	State <u>CA</u>			Certificated Rep	air Station Intenance Organization		KJJIL		
h	ave been n	nade İn a	ccordanc	e with		ents of Pa	art 43 (and described on the rate Aviation Regulations a				
	d range fue FR Part 43							ed Individual LIP G. DIFIOR	RE DI	T. C	i)		
							7. Ap	proval for Retur	n to Service				
					specified belo Iministration a				s inspected in the manner REJECTED	er prescribed	by the		
ВУ	FAA Inspe	Fit Stand	dards		Manufacture	ŗ		Maintenance Or	ganization		Person Approved by Canadian Department of Transport		
	FAA	Designe	е	х	Repair Statio	n		Inspection Author	prization	Other (S	Specify)		
Certifica Designa YT2R3	tion No.			_	ature/Date of JUNE 2017				DQ1 1	Q-4	20		

8.	Description of Work Accomplished		
	(If more space is required, attach additional sheets. Id	entify with aircraft nationality and registration ma	ark and date work completed.)
		USA N510PD	26 JUNE 2017
		Nationality and Registration Mark	Date
			j
1.	INSTALLED LEFT HAND SEAT PAN P/N 261023 AN SUPPLEMENTAL TYPE CERTIFICATE SR01960LA NUMBER INST-MDSP-01 REVISION IR DATED 10/	AND PLATINUM AVIATION GROUP INSTALL	
2.	AN ACTUAL WEIGHT AND BALANCE WAS PERFO	RMED AND THE EQUIPMENT LIST UPDATE	D.
3.	A LOGBOOK ENTRY HAS BEEN MADE.		
4.	CONTINUED AIRWORTHINESS IS ASSURED THR DOCUMENT NUMBER DL-3-ICA REVISION IR DAT		RATED FLIGHT SYSTEMS
5.	DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN	NC. UNDER WORK ORDER 17737.	
	NO	THING FOLLOWS	
			'
l			
l			
		Additional Sheets Are Attached	

U S Department of Transportation Federal Aviation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

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WP-FSDO-1

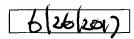
Administra	tion									AAL-Lann-I		
and di	sposition o	of this form. T	his repo	ries. See FAR ort is required be ederal Aviation	y law (49	U.S.C	ppendix B, and A0 . 1421). Failure to	C 43.9-1 (or subsection report can result in	quent revision the a civil penalty	ereof) for instructions not to exceed \$1,000		
4 0:	L	lationality and JSA N5	Registr	ation Mark				Serial No. 0400E				
1. Aircraft	I M	_{lake} ИD HELIC(OPTEI	RS INC.				Model 369E		Series		
2. Owner	_ C	CITY OF O	AKLA	egistration cert ND IELICOPTE		_		Address (As shown on registration certificate) Address 455 7 TH STREET City OAKLAND State CA				
								zip 94607-394		Country USA		
DESCRI	BED AIRCF		CT TO		(INSPEC	E AIR		EQUIREMENTS A JTHORIZED IN FA	R 43.7.	VED ONLY FOR THE ABOVE		
	4. Type						5. l	Jnit Identification				
Repair	Altera	ation	Unit			N	Make	Мо	del	Serial Number		
	×	AIRF	AIRFRAME -					(As described in	ı Item 1 above)			
	Г	POW	ERPLA	ANT								
		PROI	PELLE	R								
		APPLIANCE Type Manufac				Type Manufacturer						
					l .	6	. Conformity Stat	 tement				
A. Agenc	y's Ñame a	and Address				В. К	ind of Agency		7			
Name_R	OTORCI	RAFT SUF	PORT	Γ <u>, INC.</u>			U.S. Certificated	Mechanic		Manufacturer		
Address 1	6425 HA	RT STRE	<u>=T</u>				Foreign Certificat	ed Mechanic	t e	Certificate No.		
CityV	AN NUY	<u>s</u>		State_CA		☑ Certificated Repair Station			YT	2R331L		
Zip <u>9</u>	<u> 1406</u>	Country	<u>USA</u>	<u>\</u>			Certificated Main	tenance Organizat	ion			
hav	re been ma	ade in accorda	nce wit		ents of Pa	rt 43 d		and described on the Aviation Regulation		ntachments hereto information		
Extended per 14 CF App. B				Signature/I	185	PH	ed Individual ILLIP G. DIFIC	<u> </u>	y 16)	- \$2		
				specified belo	w, the unit	identi		inspected in the m	anner prescribe	ed by the		
	FAA FI	It Standards tor		Manufacture	r		Maintenance Org	anization		Person Approved by Canadian Department of Transport		
BY	FAA D	esignee	х	Repair Statio	n Inspection A		Inspection Author	rization	Other	(Specify)		
Certificate Designation	n No.		Sign	nature/Date of	•		idual IP G. DIFIORI	= 1Q1	(18)	Fil		

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

0	Description	of Mork	Accomplished
Ö.	Describtion	or work	Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N510PD



Nationality and Registration Mark

Date

- 1. REMOVED THE FOLLOWING ITEMS FROM THE AIRCRAFT: A TECHNISONIC INDUSTRIES LTD, TDFM 7300 TRANSCEIVER FROM THE TOP OF THE PEDESTAL; FIVE ANTENNAS FROM THE BELLY INCLUDING - TWO EACH CI-285, ONE EACH CI-275, ONE EACH CI-292-3 AND ONE FOXTRONICS FLX3050B TUNEABLE ANTENNA SYSTEM.
- 2. INSTALLED A TECHNISONIC INDUSTRIES LIMITED, TDFM-9300 S/N FTC10175 MULTIBAND DIGITAL AIRBORNE TRANSCEIVER INTO THE TOP OF PEDESTAL. THE TDFM HAS AN EXISTING SUPPLEMENTAL TYPE CERTIFICATE (NUMBER SR03448NY) FOR INSTALLATION INTO A BELL HELICOPTER 206 SERIES ROTORCRAFT, AND IS THE REFERENCE FOR THIS INSTALLATION.
- 3. FIVE NEW ANTENNAS WERE INSTALLED USING "MS" HARDWARE: TWO CI-285 ANTENNAS S/N'S 536932 & 536933 WERE MOUNTED TO THE FORWARD BELLY; ONE CI-285 S/N 536941 ANTENNA AND ONE CI-295-300 S/N 539098 ANTENNA WERE MOUNTED TO THE AFT BELLY AND A FOXTRONICS FLX3050B TUNEABLE ANTENNA SYSTEM S/N 2668 WAS INSTALLED IN THE SAME POSITION AS THE PREVIOUS SYSTEM, DOUBLERS WERE FABRICATED AND INSTALLED FOR THE LEFT FORWARD. ANTENNA LOCATION AND BOTH AFT ANTENNA LOCATIONS. THE OTHER TWO ANTENNA LOCATIONS HAD EXISTING DOUBLERS INSTALLED.
- 4. NEW WIRING WAS INSTALLED FOR THE TDFM TRANSCEIVER, TUNEABLE ANTENNA, INCLUDING ALL NEW COAX CABLE TO THE ANTENNAS.
- 5. THE INSTALLATION WAS PERFORMED REFERENCING: TECHNISONIC INSTALLATION INSTRUCTIONS, DOCUMENT NUMBER 13RE471, REVISION D. ISSUE 3. DATED 05 OCT 2016; MD HELICOPTERS HANDBOOK OF MAINTENANCE INSTRUCTIONS - CSP HMI-3, REVISION 14 DATED 15 OCT 2016 AND TEMP REVISION 16-001 DATED 07 NOV 2016; FOXCART GSE, LLC, INSTALLATION INSTRUCTIONS FOR THE FLX3050B, DRAWING 300-0002-013, REVISION A; AC43.13-1B, CHANGE 1, DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1 - WEIGHT AND BALANCE; CHAPTER 11, PARA 11-30 TO 11-32 - ELECTRICAL INSTALLATION; PARA 11-35 & 11-36 - ELECTRICAL LOAD; PARA 11-47 TO 11-51 - CIRCUIT BREAKERS; PARA 11-66 TO 11-69, 11-76 TO 11-77, 11-85 TO 11-87, & 11-89 - WIRE SELECTION; PARA 11-96 & 11-97 - WIRE INSTALLATION; PARA 11-106 & 11-107- EMI & INTERFERENCE TESTING; PARA 11-115 TO 11-118 ENVIRONMENTAL PROTECTION; PARA 11-125 - MOVEABLE CONTROLS PRECAUTIONS; PARA 11-135 TO 11-139 - SERVICE LOOP; PARA 11-146 - WIRE CLAMPING; PARA 11-155 TO 11-158 - STRIPPING AND LACING; PARA 11-167 - SPLICING; PARA 11-185 TO 11-187 GROUNDING AND BONDING; PARA 11-205 & 11-206 WIRE MARKING; PARA 11-230 TO 11-233 CONNECTORS; AC 43.13-2B, DATED MARCH 2008, CHAPTER 1, PARA 100 TO 113 - STRUCTURAL DATA; CHAPTER 2, PARA 200 TO 207- COMM INSTALLATIONS, AND CHAPTER 3, PARA 300 TO 309 - ANTENNAS.
- 6. AN ACTUAL WEIGHT AND BALANCE WAS PERFORMED AND THE EQUIPMENT LIST UPDATED.
- POWER IS PROVIDED FROM THE AVIONICS BUSS THROUGH A TEN AMP CIRCUIT BREAKER TO THE TDFM UNIT.
- 8. PERFORMED EMI, INTERFERENCE AND OPERATIONAL TESTS WITH NO ADVERSE AFFECTS ON PREVIOUSLY INSTALLED COMPONENTS AND EQUIPMENT.
- 9. THE EXISTING ELECTRICAL LOAD DOES NOT EXCEED 80% OF THE TOTAL GENERATOR CAPACITY.
- 10. TDFM-9300 OPERATING INSTRUCTIONS, DOCUMENT NO. 13RE470, REVISION F, WAS PROVIDED TO THE CUSTOMER.
- 11. PERFORM ON AN ANNUAL BASIS AN INSPECTION OF THE EQUIPMENT MOUNTINGS, ASSOCIATED WIRING, CABLES, CONNECTORS, HARDWARE, AND RELATED AIRCRAFT STRUCTURE FOR INTEGRITY, SECURITY, WEAR, CHAFING, ETC. SPECIAL ATTENTION SHOULD BE GIVEN TO THE AIRCRAFT PRIMARY STRUCTURE WITH REGARDS TO FATIGUE AND STRESS, CRACKING, CORROSION, ETC.

12.	DETAILS	ON FILE A	I ROTORCKAF	T SUPPORT INC	UNDER WORK ORDE	R 17737

۷.	BETALLS ON THE AT NOTOKOTAL TOUT INCOMBEN WORK ONDER 17707.
	NOTHING FOLLOWS
	Additional Sheets Are Attached

U.S Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

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WP-FSDO-1

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and dis	position of thi	s form. Thi	s repo	ries. See FAR rt is required be ederal Aviation	y law (49	U.S.C	ppendix B, and A0 . 1421). Failure to	C 43.9-1 (or subse- report can result i	quent revisi in a civil per	on there	eon for instructions it to exceed \$1,000	
	Natio US/	nality and F	-	ation Mark				Serial No. 0400E				
1. Aircraft	Make MD	HELICO	PTE	RS INC.				Model 369E			Series	
2. Owner	CIT	Y OF OA	KLAI	egistration cert ND IELICOPTE				Address (As shown on registration certificate) Address 455 7TH STREET City OAKLAND State CA Zip 94607-3940 Country USA				
DESCRIB	ED AIRCRAF	T SUBJEC	T TO (E AIR		EQUIREMENTS A JTHORIZED IN FA	AR 43.7.		D ONLY FOR THE ABOVE	
FAA INSP	PECTOR, VAN	I NUYS FS	DO: _	- July	Jerry	N. 1	Dees	DA	TE:&	JUIN	70 20	
4	. Туре			42214	· ·		**************************************	Jnit Identification	1			
Repair	Alteration Unit					λ	Make	Мо	odel		Serial Number	
								(As described in Item 1 above				-
		POWE	RPLA	ANT								
		PROPE	ELLEI	R								
		APPLIA	ANCE	•	Type Manufacturer			-				
								1				
						1	. Conformity Sta	tement				
	's Name and		000	T INC			ind of Agency			<u> </u>		
	OTORCRA			I, IINC.		므	U.S. Certificated				anufacturer	
-	3425 HART	STREE	<u>!</u>	0.4			Foreign Certifica				rtificate No.	
	N NUYS			State_ <u>CA</u>		Ø	Certificated Repa	ir Station		Y 12F	R331L	
Zip <u>91</u>	<u>406</u>	Country_	USA	!			Certificated Main	tenance Organizat	tion	-		
have	e been made i	n accordan	ce witl		ents of Pa	rt 43 d		and described on Aviation Regulation				
Extended r per 14 CFF App. B				1.1	6/201) PH	ed Individual ILLIP G. DIFIO	P	400), 4	EP .	
				•			proval for Return		0			
				specified belo Iministration a				inspected in the m EJECTED	narmer pres	cribed t	by the `	
ВУ	FAA Fit St Inspector	andards		Manufacture	r		Maintenance Org	anization		De	erson Approved by Canadian epartment of Transport	
	FAA Desig	nee	X Repair Station		n		Inspection Autho			ther (Sp	pecify)	
Certificate Designation YT2R33	n No.		Sign	ature/Date of			^{idual} IP G. DIFIOR	= NO	18)	7		

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

	Description of Work Accomplished
((If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
	USA N510PD - 26 261
	Nationality and Registration Mark Date
	Nationality and Registration Mark Date
1.	REMOVED THE GARMIN GNS 430 GPS / NAV / COMM FROM THE INSTRUMENT PANEL AND THE GA 56 GPS ANTENNA FROM
	THE MOUNT LOCATED ON THE TOP OF THE AFT FUSELAGE.
2.	INSTALLED A GARMIN GTN 650 GPS / NAV / COMM TRANSCEIVER P/N 010-00813-A0 S/N 1Z8022881 IN THE INSTRUMENT
	PANEL AND GA 35 S/N 135785 GPS ANTENNA ON AN EXISTING MOUNT ON TOP OF THE AFT FUSELAGE REFERENCING
	SUPPLEMENTAL TYPE CERTIFICATE NUMBER SR02120SE AND MASTER DRAWING LIST NO. 005-00533-H0, REVISION 2.
	THE GTN WAS INTERFACED TO THE EXISTING GI-106A NAV INDICATOR. THERE IS A DEVIATION FROM THE STC WHICH
	SHOWS THE GTN 650 INSTALLED IN THE TOP OF THE PEDESTAL AND THE GA 35 INSTALLED ON TOP OF THE UPPER AFT
	FUSELAGE ENCLOSURE.
3.	THE INSTALLATION WAS PERFORMED REFERENCING: GARMIN GTN 6XX/7XX PART 27 AML STC INSTALLATION MANUAL NO.
	190-01007-B3, REVISION 2; MD HELICOPTERS HANDBOOK OF MAINTENANCE INSTRUCTIONS – CSP HMI-3, REVISION 14
	DATED 15 OCT 2016 AND TEMP REVISION 16-001 DATED 07 NOV 2016; AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001:
	CHAPTER 10, PARA 10-1 - WEIGHT AND BALANCE; CHAPTER 11, PARA 11-30 TO 11-32 - ELECTRICAL INSTALLATION; PARA 1
	35 & 11-36 - ELECTRICAL LOAD; PARA 11-47 TO 11-51 - CIRCUIT BREAKERS; PARA 11-66 TO 11-69, 11-76 TO 11-77, 11-85 TO 11
	87, & 11-89 - WIRE SELECTION; PARA 11-96 & 11-97 - WIRE INSTALLATION; PARA 11-106 & 11-107- EMI & INTERFERENCE
	TESTING; PARA 11-115 TO 11-118 ENVIRONMENTAL PROTECTION; PARA 11-125 - MOVEABLE CONTROLS PRECAUTIONS;
	PARA 11-135 TO 11-139 - SERVICE LOOP; PARA 11-146 - WIRE CLAMPING; PARA 11-155 TO 11-158 - STRIPPING AND LACING
	PARA 11-167 - SPLICING; PARA 11-185 TO 11-187 GROUNDING AND BONDING; PARA 11-205 & 11-206 WIRE MARKING; PARA 1 230 TO 11-233 CONNECTORS; AC 43.13-2B, DATED MARCH 2008, CHAPTER 1, PARA 100 TO 113 - STRUCTURAL DATA;
	CHAPTER 2, PARA 200 TO 207- COMM INSTALLATIONS, AND CHAPTER 3, PARA 300 TO 309 – ANTENNAS.
4	GTN 625/635/650 COCKPIT REFERENCE GUIDE NO. 190-01004-04, REVISION K AND GTN 625/635/650 PILOT'S GUIDE NO. 190-
7.	01004-03, REVISION L, WERE PROVIDED TO THE CUSTOMER.
5.	GARMIN STC PERMISSION LETTER WAS FILED IN THE AIRCRAFT RECORDS.
	ROTORCRAFT FLIGHT MANUAL SUPPLEMENT NO. 190-01007-JB, REVISION 2 WAS FILED IN THE ROTORCRAFT FLIGHT
	MANUAL.
7.	AN ACTUAL WEIGHT AND BALANCE WAS PERFORMED AND THE EQUIPMENT LIST UPDATED.
8.	POWER IS PROVIDED FROM THE AVIONICS BUSS THROUGH TWO FIVE AMP CIRCUIT BREAKERS TO THE GTN UNIT.
9.	PERFORMED EMI, INTERFERENCE AND OPERATIONAL TESTS WITH NO ADVERSE AFFECTS ON PREVIOUSLY INSTALLED
	COMPONENTS AND EQUIPMENT.
0.	THE EXISTING ELECTRICAL LOAD DOES NOT EXCEED 80% OF THE TOTAL GENERATOR CAPACITY.
1.	REFER TO GARMIN GTN 6XX/7XX PART 27 AML STC MAINTENANCE MANUAL NO. 190-01007-B1, REVISION 2 FOR
	MAINTENANCE AND TROUBLESHOOTING INSTRUCTIONS. REFER TO GARMIN INSTRUCTIONS FOR CONTINUED
	AIRWORTHINESS NO. 190-01007-KB, REVISION 2 FOR INSPECTION REQUIREMENTS WHICH INCLUDES 100 HOUR / 12 MONTI
	INSPECTION AND A 10 YEAR / 2000 HOUR ELECTRICAL BONDING CHECK.
_	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC UNDER WORK ORDER 17737.
2.	

U.S Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

WP-FSDO-1

and o	dispositio	n of this f	orm. This	repo		y law (49 l	U.S.C	ppendix B, and A0 . 1421). Failure to				off for instructions	
		Nationa USA	ality and Re		ation Mark				Serial No. 0400E				
1. Aircra	aft	Make MD H	ELICOF	TE	RS INC.				Model 369E			Series	
2. Owne	er	CITY	OF OA	ΚLΑΙ	egistration cert ND IELICOPTE		-		Address (As shown on registration certificate) Address 455 7 TH STREET City OAKLAND State CA Zip 94607-3940 Country USA				
							3	. For FAA Use C	nly				
								WORTHINESS R BY A PERSON AI	JTHORIZED II	N FAR 43.7,		ONLY FOR THE ABOVE	
FAA IN	SPECTOI	R, VAN N	NUYS FSD	00: _	-JA	Jerry K	ľ. D	<i>- Ull</i> 9es		DATE:	IN Z	<u>6 2017 </u>	
	4. Type						•	_	Jnit Identifica	ition			
Repai	Repair Alteration Unit				nit		N	/lake		Model		Serial Number	
		AIRFRAME							(As described in Item 1 above)				
			POWER	RPLA	ANT								
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	APPLIANCE _				Type Manufacturer								
			l				6	. Conformity Sta	ement				
A. Agen	icy's Nam	e and Ad	Idress				1	ind of Agency					
Name	ROTOR	CRAF	T SUPP	ORT	T, INC.			U.S. Certificated	Mechanic		☐ Mar	nufacturer	
Address_	16425 F	HART S	STREET	-				Foreign Certifica	ed Mechanic		C. Certi	ificate No.	
City\	/AN NU	<u>JYS</u>			State_CA		Ø	Certificated Repa	ir Station		YT2R331L		
Zip	91406		Country_[<u>USA</u>	<u>.</u>			Certificated Main	lenance Orgai	nization			
ha	ave been	made in	accordanc	e with		ents of Pa	rt 43 c	ed in item 5 above of the U.S. Federa					
	d range fu FR Part 4				Signature/E	Date of Aut		ed Individual ILLIP G. DIFIO	DRE X	2118	平	P	
							7. Ap	proval for Return	to Service	1100			
					specified belo Iministration a			fied in item 5 was PROVED R	inspected in th EJECTED	ne manner preso	cribed by	the	
DV.		A FIt Star pector	ıdar d s		Manufacture	r		Maintenance Org	anization			son Approved by Canadian partment of Transport	
BY	FAA	\ Design	ee	X Repair Station		n		Inspection Authorization			her (Spe	ecify)	
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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8.	Descrip	otion	of	Work	Accom	plished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N510PD

Nationality and Registration Mark

Date

- 1. REMOVED THE GARMIN GTX 330 TRANSPONDER FROM THE PEDESTAL, THE KA-60A TRANSPONDER ANTENNA FROM THE AFT BELLY AND A SSD120-30N BLIND ENCODER FROM THE SHELF FORWARD OF THE PEDESTAL.
- 2. INSTALLED A GARMIN GTX 345R REMOTE TRANSPONDER P/N 010-01217-01 S/N 3EH002165 WITH ADS-B OUT AND IN FUNCTIONALITY UNDER THE PILOT SEAT, REFERENCING GARMIN INTERNATIONAL, INC. SUPPLEMENTAL TYPE CERTIFICATE NUMBER SR02124SE AND MASTER DRAWING LIST 005-00734-A4, REVISION 1. THIS IS A DEVIATION FROM THE STC WHICH SHOWS THE GTX 345R LOCATED BEHIND THE INSTRUMENT PANEL, A MOUNT FABRICATED FROM 2024-T3 ALUMINUM SHEET .040" THICK WAS USED TO MOUNT THE TRANSPONDER UNDER THE PILOT SEAT PANEL.
- 3. A CI 105 ANTENNA WAS INSTALLED ON THE AFT BELLY AND A SSD120-30N-RS232 ENCODER S/N SRN23237 INSTALLED ON THE SHELF FORWARD OF THE PEDESTAL.
- 4. THE GTX 345R IS INTERFACED TO THE GTN 650 FOR DISPLAY.
- 5. PILOT'S GUIDE, NO. 1490-01499-00, REVISION C WAS PROVIDED TO THE CUSTOMER.
- 6. THE INSTALLATION WAS PERFORMED REFERENCING: GARMIN GTX 3X5, PART 27, AML STC INSTALLATION MANUAL NO. 190-00734-20, REVISION 4; MD HELICOPTERS HANDBOOK OF MAINTENANCE INSTRUCTIONS - CSP HMI-3, REVISION 14 DATED 15 OCT 2016 AND TEMP REVISION 16-001, DATED 07 NOV 2016; TRANS-CAL INDUSTRIES, INC. MODEL SSD120-(XX)N-RS232 ALTITUDE ENCODER OWNER / INSTALLATION MANUAL NO. 882189, REVISION G; AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1 - WEIGHT AND BALANCE: CHAPTER 11, PARA 11-30 TO 11-32 - ELECTRICAL INSTALLATION: PARA 11-35 & 11-36 - ELECTRICAL LOAD; PARA 11-47 TO 11-51 - CIRCUIT BREAKERS; PARA 11-66 TO 11-69, 11-76 TO 11-77, 11-85 TO 11-87, & 11-89 - WIRE SELECTION; PARA 11-96 & 11-97 - WIRE INSTALLATION; PARA 11-106 & 11-107 - EMI & INTERFERENCE TESTING; PARA 11-115 TO 11-118 ENVIRONMENTAL PROTECTION; PARA 11-125 - MOVEABLE CONTROLS PRECAUTIONS; PARA 11-135 TO 11-139 - SERVICE LOOP; PARA 11-146 - WIRE CLAMPING; PARA 11-155 TO 11-158 -STRIPPING AND LACING; PARA 11-167 - SPLICING; PARA 11-185 TO 11-187 GROUNDING AND BONDING; PARA 11-205 & 11-206 WIRE MARKING; PARA 11-230 TO 11-233 CONNECTORS; AC 43.13-2B, DATED MARCH 2008, CHAPTER 1, PARA 100 TO 113 -STRUCTURAL DATA; CHAPTER 2, PARA 200 TO 207- COMM INSTALLATIONS, AND CHAPTER 3, PARA 300 TO 309 - ANTENNAS.
- 7. AN ACTUAL WEIGHT AND BALANCE WAS PERFORMED AND THE EQUIPMENT LIST UPDATED.
- 8. POWER IS PROVIDED FROM THE AVIONICS BUSS THROUGH A FIVE AMP CIRCUIT BREAKER TO THE GTX 345R.
- PERFORMED EMI, INTERFERENCE AND OPERATIONAL TESTS WITH NO ADVERSE AFFECTS ON PREVIOUSLY INSTALLED COMPONENTS AND EQUIPMENT.
- 10. THE EXISTING ELECTRICAL LOAD DOES NOT EXCEED 80% OF THE TOTAL GENERATOR CAPACITY.
- 11. GARMIN STC PERMISSION LETTER WAS FILED IN THE AIRCRAFT RECORDS.
- 12. FLIGHT MANUAL SUPPLEMENT NO. 190-00734-22, REVISION 1, WAS FILED IN THE ROTORCRAFT FLIGHT MANUAL.
- 13. THE SYSTEM IS APPROVED FOR SERVICE AFTER SATISFACTORY CHECK FLIGHT VALIDATED BY THE ADS-B COMPLIANCE MONITOR.
- 14. REFER TO GARMIN GTX 3X5 PART 27 AML MAINTENANCE MANUAL NO. 190-00734-21, REVISION 1 FOR INSTRUCTIONS ON

	CONTINUED AIRWORTHINESS WHICH INCLUDES A 12 MONTH VISUAL INSPECTION AND A 10 YEAR / 2000 HOUR BONDING
	TEST.
15.	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC UNDER WORK ORDER 17737.

	TEST.
15.	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC UNDER WORK ORDER 17737.
	NOTHING FOLLOWS
	_
	Additional Sheets Are Attached

U.S Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

WP-FSDO-1

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and c	disposition	of this f	orm. This	repo		y law (49 l	U.S.C		43.9-1 (or subseque report can result in a			of) for instructions	
		Nationa USA	lity and Ro	-	ation Mark				Serial No. 0400E				
1. Aircra	ft	Make MD H	ELICOF	TEF	RS INC.				Model 369E			Series	
2. Owne	er	CITY	OF OAK	ΚLΑΙ	egistration cert ND IELICOPTE		-		Address (As shown on registration certificate) Address 455 7 TH STREET City OAKLAND State CA Zip 94607-3940 Country USA				
							3	B. For FAA Use O	nly				
									EQUIREMENTS AND ITHORIZED IN FAR		OVED	ONLY FOR THE ABOVE	
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	4. Type	-						5. l	Unit Identification				
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		AIRFRAME					-		(As described in Item 1 above)				
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			Country_					l	tenance Organization				
ha	ive been r	nade in a	accordanc	e with		ents of Pa	rt 43 c		and described on the Aviation Regulations				
	f range fu FR Part 4				Signature/I	(201)	PH	ed Individual ILLIP G. DIFIC	W.	0	J)	
					specified belo	w, the unit	identi	ified in item 5 was	inspected in the mann	ner prescri	bed by	the	
	FAA Inspe	FIt Stan	dards		Manufacture	r		Maintenance Org	ganization			son Approved by Canadian artment of Transport	
BY	FAA	Designe	ee	Х	Repair Statio	tion		Inspection Authorization		Othe	er (Spe	cify)	
Certificate Designati	ion No.			Sign	ature/Date of			idual IP G. DIFIORI	1 Qa	10	7		

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

	a destruction and the second s								
8.	Description of Work Accomplished								
	(If more space is required, attach additional sheets. tdentify with aircraft nationality and registration mark and date work completed.)								
	USA N510PD 6 26 201)								
	Nationality and Registration Mark Date								
1	REMOVED TWO CIGARETTE LIGHTER ADAPTERS ONE FROM THE COCKPIT AND ONE FROM THE PASSENGER COMPARTMENT.								
2	INSTALLED A MID-CONTINENT INSTRUMENT CO., INC., TRUE BLUE POWER, TA102 DUAL USB CHARGING PORT P/N 6430102-1 S/N C17-10899 BELOW THE INSTRUMENT PANEL. THE UNIT MEETS THE PERFORMANCE STANDARDS OF TSO-C71 AND ENVIRONMENTAL QUALIFICATION DO-160G.								
3	THE INSTALLATION WAS PERFORMED REFERENCING: TRUE BLUE POWER, INSTALLATION MANUAL AND OPERATING INSTRUCTIONS NO. 9017942, REVISION J; MD HELICOPTERS HANDBOOK OF MAINTENANCE INSTRUCTIONS – CSP HMI-3, REVISION 14 DATED 15 OCT 2016 AND TEMP REVISION 16-001 DATED 07 NOV 2016; AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1 – WEIGHT AND BALANCE; CHAPTER 11, PARA 11-30 TO 11-32 – ELECTRICAL INSTALLATION; PARA 11-35 & 11-36 - ELECTRICAL LOAD; PARA 11-47 TO 11-51 - CIRCUIT BREAKERS; PARA 11-66 TO 11-69, 11-76 TO 11-77, 11-85 TO 11-87, & 11-89 - WIRE SELECTION; PARA 11-96 & 11-97 - WIRE INSTALLATION; PARA 11-106 & 11-107- EMI & INTERFERENCE TESTING; PARA 11-115 TO 11-118 ENVIRONMENTAL PROTECTION; PARA 11-125 - MOVEABLE CONTROLS PRECAUTIONS; PARA 11-135 TO 11-139 – SERVICE LOOP; PARA 11-146 - WIRE CLAMPING; PARA 11-155 TO 11-158 – STRIPPING AND LACING; PARA 11-167 - SPLICING; PARA 11-185 TO 11-187 GROUNDING AND BONDING; PARA 11-205 & 11-206 WIRE MARKING; PARA 11-230 TO 11-233 CONNECTORS; AC 43.13-2B, DATED MARCH 2008, CHAPTER 1, PARA 100 TO 113 – STRUCTURAL DATA; AND CHAPTER 2, PARA 200 TO 207- COMM INSTALLATIONS.								
4.	INSTALLATION MANUAL AND OPERATING INSTRUCTIONS WAS PROVIDED TO THE CUSTOMER.								
5.	AN ACTUAL WEIGHT AND BALANCE WAS PERFORMED AND THE EQUIPMENT LIST UPDATED.								
6.									
	PERFORMED EMI, INTERFERENCE AND OPERATIONAL TESTS WITH NO ADVERSE AFFECTS ON PREVIOUSLY INSTALLED COMPONENTS AND EQUIPMENT.								
8.	THE EXISTING ELECTRICAL LOAD DOES NOT EXCEED 80% OF THE TOTAL GENERATOR CAPACITY.								
9.	PERFORM ON AN ANNUAL BASIS AN INSPECTION OF THE EQUIPMENT MOUNTINGS, ASSOCIATED WIRING, CABLES,								
	CONNECTORS, HARDWARE, AND RELATED AIRCRAFT STRUCTURE FOR INTEGRITY, SECURITY, WEAR, CHAFING, ETC.								
	SPECIAL ATTENTION SHOULD BE GIVEN TO THE AIRCRAFT PRIMARY STRUCTURE WITH REGARDS TO FATIGUE AND								
	STRESS, CRACKING, CORROSION, ETC.								
10.	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC UNDER WORK ORDER 17737.								
	NOTHING FOLLOWS								

U.S Department of Transportation

Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

WP-FSDO-1

and disp	JCTIONS: Print position of this a such violation	form. This	s repo	rt is required b	y law (49 l	U.S.C	ppendix B, and AC . 1421). Failure to	43.9-1 (or subsequent i report can result in a civ	evision thill penalty	nereof) for instructions not to exceed \$1,000		
	Nation USA	ality and R N510		ation Mark				Serial No. 0400E				
1. Aircraft	Make MD F		PTE	RS INC.				Model 369E		Series		
2. Owner	CITY	OF OA	KLAI	egistration cert ND IELICOPTE	ER UNIT			Address (As shown on registration certificate) Address 455 7 TH STREET City OAKLAND State CA Zip 94607-3940 Country USA				
						E AIR	BY A PERSON AU	EQUIREMENTS AND IS THORIZED IN FAR 43.	7.	VED ONLY FOR THE ABOVE		
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A. Agency's	s Name and A	ddress					and of Agency	ement	_	4.8		
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Address 164	425 HART	STREE	Γ				Foreign Certificat	ed Mechanic	C.	C. Certificate No.		
CityVAI	N NUYS			State_ <u>CA</u>		Ø	Certificated Repa	ir Station	YT	YT2R331L		
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have		accordan	ce with	h the requirem	ents of Pa	rt 43 (and described on the rev Aviation Regulations an				
Extended ra per 14 CFR App. B				11.1	20()	PH	ed Individual ILLIP G. DIFIC	MIC	7	ke		
	to the authorit			•	w, the unit	ident		to Service Inspected in the manner EJECTED	prescribe	ed by the		
	FAA Fit Star	ndards		Manufacture			Maintenance Org	anization	T	Person Approved by Canadian Department of Transport		
BY	FAA Design	ee	X Repair Station		n l		Inspection Author	Inspection Authorization		(Specify)		
Certificate of Designation	No.		Sign	alure/Date of			idual IP G. DIFIORE	10018) =	\$0		

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

	(If more space is required, attach additional sheets. Iden	nt <mark>ify with aircraft</mark> nationality and registration m	ark and date work completed.)
		USA N510PD	6(26(201)
		Nationality and Registration Mark	Date
	REMOVED THE TECHNISONIC INDUSTRIES LIMITE CONTROL PANELS – ONE FROM THE INSTRUMEN (PILOT).	T PANEL (CO-PILOT) AND ONE FROM THE	BOTTOM OF THE PEDESTAL
2	. INSTALLED A TECHNISONIC INDUSTRIES LIMITED, CONTROL PANELS P/N 961072-60 S/N'S MJ1310 & M THE RIGHT HAND SIDE OF THE INSTRUMENT PANEL	MJ1309 - ONE INTO THE BOTTOM OF THE	
3.	THE INSTALLATION WAS PERFORMED REFERENCE REVISION A; MD HELICOPTERS HANDBOOK OF MAY AND TEMP REVISION 16-001 DATED 07 NOV 2016; WEIGHT AND BALANCE; CHAPTER 11, PARA 11-30 LOAD; PARA 11-47 TO 11-51 - CIRCUIT BREAKERS; SELECTION; PARA 11-96 & 11-97 - WIRE INSTALLATED 11-118 ENVIRONMENTAL PROTECTION; PARA 11-118 ENVIRONMENTAL PROTECTION; PARA 11-118 TO 11-1187 GROUNDING AND BONDING CONNECTORS; AC 43.13-2B, DATED MARCH 2008, SPARA 200 TO 207- COMM INSTALLATIONS.	AINTENANCE INSTRUCTIONS – CSP HMI-3, AC43.13-1B, CHANGE 1,DATED SEPT 27, 2 TO 11-32 – ELECTRICAL INSTALLATION; P PARA 11-66 TO 11-69, 11-76 TO 11-77, 11-8 TION; PARA 11-106 & 11-107- EMI & INTERF I1-125 - MOVEABLE CONTROLS PRECAUTI PARA 11-155 TO 11-158 – STRIPPING AND G; PARA 11-205 & 11-206 WIRE MARKING; I	REVISION 14 DATED 15 OCT 20: 2001: CHAPTER 10, PARA 10-1 – PARA 11-35 & 11-36 - ELECTRICAL 25 TO 11-87, & 11-89 - WIRE FERENCE TESTING; PARA 11-115 21 JONS; PARA 11-135 TO 11-139 – LACING; PARA 11-167 - SPLICING PARA 11-230 TO 11-233
4.	OPERATING INSTRUCTIONS MANUAL, DOCUMENT AN ACTUAL WEIGHT AND BALANCE WAS PERFOR		
6. 7.	POWER IS PROVIDED FROM THE AVIONICS BUSS TWO AUDIO NOISE REDUCTION (ANR) HEADPHONI OF THE COCKPIT AND ARE PROTECTED BY A ONE PERFORMED EMI, INTERFERENCE AND OPERATION	THROUGH A ONE AMP CIRCUIT BREAKER E ADAPTERS P/N HCA-PPM24 WERE INSTA AMP CIRCUIT BREAKER.	S TO EACH AUDIO PANEL. ALLED IN THE AFT UPPER AREA
9.	COMPONENTS AND EQUIPMENT. THE EXISTING ELECTRICAL LOAD DOES NOT EXCE	EED 80% OF THE TOTAL GENERATOR CA	PACITY.
	PERFORM ON AN ANNUAL BASIS AN INSPECTION CONNECTORS, HARDWARE, AND RELATED AIRCF SPECIAL ATTENTION SHOULD BE GIVEN TO THE ASTRESS, CRACKING, CORROSION, ETC.	OF THE EQUIPMENT MOUNTINGS, ASSOC RAFT STRUCTURE FOR INTEGRITY, SECU	CIATED WIRING, CABLES, RITY, WEAR, CHAFING, ETC.
11.	DETAILS ON FILE AT ROTORCRAFT SUPPORT INC	•	
11.	SPECIAL ATTENTION SHOULD BE GIVEN TO THE A STRESS, CRACKING, CORROSION, ETC. DETAILS ON FILE AT ROTORCRAFT SUPPORT INC	AIRCRAFT PRIMARY STRUCTURE WITH RI	EGARDS TO FATIGUE ANI

U.S Department of
Transportation
Federal Aviation

Form Approved OMB No. 2120-0020

U.S Departm	nent of	,	A :	omo Dov	vernier	4 D	ropollo	- 0 - A	ppliance)	11/30/2007				
Transportation	on	(4	AIITI	ame, Pov	verpian	it, P	ropellei	r, or A	ppliance)		For FA	VA Use Only		
Federal Avia Administrati														
and dis	position of ti	nis form. Thi	s repo	ries. See FAR ort is required be ederal Aviation	y law (49	U.S.C	ppendix B, . 1421). Fa	and AC	43.9-1 (or subseque eport can result in a	nt revision to civil penalty	nereof) for ins not to exceed	structions d \$1,000		
		ionality and F		ation Mark		Serial No. 0400E								
1. Aircraft	Mal			OUGLAS				1	Model 369E	del Series				
		egistration cert	ificate)			1	Address (As shown o	-	n certificate)					
2. Owner	PC	LICE HEI	ICO	PTER UNI	т		cityOAKLAND StateCA					State <u>CA</u>		
							B. For FAA		ip <u>94607-3940</u>	(country <u>USA</u>			
							S. FOFFAA	Use Un	ıy					
4.	. Type				r			5. U	nit Identification		Т			
Repair	Alteration	en	Ur	nit		V	<i>l</i> lake		Model			Serial Number		
	\boxtimes	AIRFR						(As described in Ite	em 1 above)					
		POWE	RPLA	ANT					*					
		PROP	ELLE	R										
		APPLIA	ANCE		Type Manufacturer									
	<u> </u>						. Conform	ity State	ment					
A. Agency'	's Name and	Address					ind of Ager							
		AFT SUPE	ORT	Γ, INC.			U.S. Certi	ficated N	lechanic		Manufacture	r		
Address 16	425 HAR	T STREE	Ī				Foreign C	ertificate	d Mechanic	C.	Certificate No).		
city <u>VA</u>	N NUYS			State_CA		Ø	Certificate	d Repair	Station	YT	2R331L			
Zip914	406	Country_	USA				Certificate	d Mainte	nance Organization					
have	been made	in accordan	ce witi	ion made to the h the requirem to the best of	ents of Pa	rt 43 c	ed in item 5 of the U.S. I	above a	nd described on the Aviation Regulations	reverse or a and that the	ttachments h information	ereto		
Extended ra per 14 CFR App. B				Signature/I 26 APRI		thorize	ed Individua	ai	PHILLIP G. DIF	IORE ((ND)=	30		
						7. Ap	proval for	Return t	o Service		10			
				specified belo Iministration a	K.	_	ified in item PROVED		spected in the mann	ner prescribe	d y the			
ву	FAA Fit S Inspector			Manufacture	r		Maintenar	nce Orga	nization		Department	oved by Canadian of Transport		
	FAA Des	ignee	х	Repair Statio	on		Inspection	Authoria	zation	Other	(Specify)			
Certificate of Designation YT2R331	n No.		1 ~	ature/Date of APRIL 201		Indiv	idual	Pŀ	IILLIP G. DIFIO	RE	(A) 3h	l		

8. Description of Work Accomplished	•	
(If more space is required, attach additional she	ets. Identify with aircraft nationality and registration mark	and date work completed.)
	USA N510PD	26 APRIL 2016
	Nationality and Registration Mark	Date
THE ORIGINAL TURBINE OUTLET TEMPER	ATURE (TOT) INSTRUMENT WAS REMOVED.	
	DIGITAL TOT INDICATOR WITH EXCEEDANCE LOGGIN	IG CAPABILITY.
	L, WAS INSTALLED IN THE INSTRUMENT PANEL.	
·	ENCING SUPPLEMENTAL TYPE CERTIFICATE NUMBE	R SR00031LB AND DRAWING
NO. 21001, REV C, DATED 20 APR 2010.		
5. INSTRUCTIONS FOR CONTINUED AIRWOR	THINESS ARE CONTAINED IN MANUAL NO. 87041, RE	VISION B, DATED 01 JUN 1999.
6. FLIGHT MANUAL SUPPLEMENT NO. 87012,	REV B, DATED 07 MAY 1999 WAS FILED IN THE ROTO	ORCRAFT FLIGHT MANAUL.
7. OWNERS MANUAL, DOCUMENT NO. 87040	, REVISION E, DATED 08 MAY 2012, WAS PLACED IN T	THE AIRCRAFT RECORDS.
8. COMPLETED A COMPASS SWING AND UPI	DATED COMPASS CORRECTION CARD.	
9. A LOGBOOK ENTRY HAS BEEN MADE.		
10. EQUIPMENT LIST REVISED.		
11. WEIGHT AND BALANCE REVISED.		
 PERMISSION LETTER FOR USE OF THE ST RECORDS. 	C PROVIDED BY JASON HARRIS OF DIAMOND J, INC.	AND FILED IN THE AIRCRAFT
13. PERFORMED EMI, INTERFERENCE AND O	PERATIONAL TESTS WITH NO ADVERSE AFFECTS ON	N PREVIOUSLY INSTALLED
COMPONENTS AND EQUIPMENT.	• • • • • • • • • • • • • • • • • • •	
14. THE MAXIMUM CONTINUOUS ELECTRICAL	LOAD DOES NOT EXCEED 80% OF THE TOTAL GENE	RATOR CAPACITY.
15. DETAILS ON FILE AT ROTORCRAFT SUPPO	ORT, INC. UNDER WORK ORDER 17064.	
	NOTHING FOLLOWS	84
	,	
•		
	••	
	W.E.	•
	Additional Sheets Are Attached	·

U.S Department of
Transportation
Endoral Aviation

Form Approved OMB No. 2120-0020 11/30/2007

U.S Departm	nent of		/ A :	Da.		4 D	ranallan an	Annlianas	11/00/20	<u> </u>		
Transportatio	ransportation (All Hallie, Fower plant, Fropener, or Appliance) ederal Aviation										For FAA Use Only	
Federal Avia Administration												
0.050	UOTIONO I				10.0.51							_
and dis	sposition of th	is form. Th	nis repo		by law (49 l	U.S.C.		C 43.9-1 (or subseque report can result in a				
4.4	US.	onality and A N510		ation Mark				Serial No. 0400E				
1. Aircraft	Mak		LL D(DUGLAS				Model 369E			Series	
2. Owner	CIT	Y OF OA	٩KLAI	egistration cert ND PTER UNI		Address (As shown on registration certificate) Address 455 7 TH ST CityOAKLAND StateCA				_		
						3. For FAA Use O	zip <u>94607-3940</u> Pnly		Courn	<u> </u>		
4.	. Type				т—		5, (Unit Identification				\dashv
Repair	r Alteration Unit					N	Make	Model			Serial Number	
	☐ AIRFRAME -				_			(As described in Ite	em 1 abo	ve)		
		POWE	ERPL⁄	ANT								
		PROP	'ELLE'	R								
		APPLI	ANCE	Ē	Type Manufactu	ırer						
						6	. Conformity Sta	tement		٠		
A. Agency	's Name and	Address					Kind of Agency					-3
Name_RC	OTORCRA	FT SUP	POR	Γ, INC.			U.S. Certificated	Mechanic		☐ Mar	nufacturer	
Address 16	425 HAR	<u>r stree</u>	<u>:T</u>				Foreign Certificat	ted Mechanic			ificate No.	_
CityVA	<u>IN NUYS</u>			State <u>CA</u>		Ø	Certificated Repa	air Station		YT2R3	331L	_
zip914	<u>406</u>	Country	<u>USA</u>	<u>\</u>			Certificated Main	tenance Organization				
have	e been made	in accorda	nce with		nents of Pa	art 43 c		and described on the I Aviation Regulations				
Extended raper 14 CFR App. B	ange fuel				Date of Aut		ed Individual	PHILLIP G. DIF	IORE	Da	10 Fel	
						7. Ap	proval for Return	to Service				
				specified belo dministration a	· -	_		inspected in the man	ner preso	ribed by	the	
ву —	FAA Flt S Inspector			Manufacture	ır		Maintenance Org	ganization		Dep	son Approved by Canadian partment of Transport	
	FAA Desi	gnee	Х	Repair Statio	on	Inspection Authorization			Other (Specify)			
Certificate or Designation No. Signature/Date of Authorized Individual PHILLIP G. DIFIORE On the Phillip G. DIFIORE PHILLIP G. DIFIORE												

3.	Description of Work Accomplished		
	(If more space is required, attach additional sheets. Id	lentify with aircraft nationality and registrat	ion mark and date work completed.)
		USA N510PD	26 APRIL 2016
		Nationality and Registration Mark	Date
۱.	INSTALLED NEW OIL CAP ASSEMBLY P/N AM369/ CERTIFICATE SR01996SE AND AIRCRAFT MANUF NUMBER AMC MI-08002 REVISION NC DATED 09/	FACTURING COMPANY, LLC INSTALLAT	
2.			ANCE MANUAL CSP-HMI-2, SECTION
3.	A LOGBOOK ENTRY HAS BEEN MADE.		
	EQUIPMENT LIST REVISED.	• • •	• • •
	WEIGHT AND BALANCE NEGLIGIBLE.		
	DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN	IC LINDER WORK ORDER 17064	
· ·			
	·	101111101 0220110	
	-		
		Additional Sheets Are Attached	

MAJOR REPAIR AND ALTERATION (Airframe Powernlant Propeller or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Transport			()	VII II	aine, row	rei þlati	ι, Γ	opener, or	Appliance)		For FAA Use Onty		
Federal Administ													
and	dispositio	n of this f	form. This	repo	ries. See FAR ort is required b ederal Aviation	y law (49 l	J.S.C.	ppendix B, and A(1421). Failure Io	2 43.9-1 (or subsequent report can result in a ci	revision thereovil penalty not	of) for instructions to exceed \$1,000		
		Nationa U.S.A	•	egistri 10Pl	ation Mark	•		-	Serial No. 0400E				
1. Aircra	aft	Make MCD0	 ONNELI	L DC	OUGLAS				Model 369E	· · · · · · · · · · · · · · · · · · ·			
					egistration certi	ificate)	-		Address (As shown on	registration ce	rtificate)		
2. Own	er		OF OAK			r			Address455 7 TH ST				
		POLIC	TE UEL	100	PTER UNIT	ı			city <u>OAKLAND</u> state <u>CA</u> zip94607-3940 country				
		<u> </u>					3	. For FAA Use C		Count	ıry		
	4. Type	-							Jnit Identification				
Repa	ir Alt	teration	on Unit					1ake 	Model		Serial Number		
			AIRFRAME					(As described in Ite.		n 1 above)			
			POWER	₹₽∟₽	ANT								
			PROPE	LLE	R								
			APPLIA	NCE	<u>:</u>	Type Manufacturer			-				
								Conformity Sta	tement				
	ncy's Nam		ort, Inc					ind of Agency U.S. Certificated	Mochanic	IT Ma	nufacturer		
	16425 H			4			믐	Foreign Certifica			ificate No.		
_	Van Nu		<u> </u>		State_CA			Certificated Repa		YT2R:	331L		
	91406		Country_	US <u>A</u>				Certificated Main	tenance Organization				
h	ave been	made in a	accordanc	e with	ion made to th h the requirem to the best of i	ents of Pa	rt 43 c	ed in item 5 above of the U.S. Federa	and described on the re Aviation Regulations a	everse or attac nd that the info	hments hereto ormation		
	d range fu FR Part 4					Date of Aut		d Individual	PHILLIP G. DIFIC	RE D	10 Feb		
							7. Ap	provat for Return	to Service	1	1		
					specified belo Iministration a				inspected in the manne	r prescribed by	y the		
BV.	ľ	A Flt Stan	dards		Manufacture	ſ		Maintenance Org	anization	Der	son Approved by Canadian partment of Transport		
BY	FAA	A Designe	e	Х	Repair Statio	n		Inspection Autho	rization	Other (Spe	ecify)		
Certification Designation YT2R3	tion No.			1 -	nature/Date of 0		Indivi		LIP G. DIFIORE	80118	(). Il		

8. Descri	otion of Work Accomplished	<u></u>										
(If more	space is required, attach additional sheets. Ide	ntify with a	ircraft nationality and registrati	on mark and date	work completed.)							
		U.S.A.	N510PD		06 JUL 2015							
		National	ity and Registration Mark		Date							
	OVED THE XENON STROBE LIGHT FROM THE ER THE COCKPIT FLOOR.	E FORWA	RD BELLY AND THE STROBE	E LIGHT POWER	SUPPLY FROM							
2. INST	2. INSTALLED A WHELEN RED LED STROBE LIGHT P/N 01-0771080-01 AT THE SAME LOCATION ON THE BELLY REFERENCING											
WHELEN SUPPLEMENTAL TYPE CERTIFICATE NUMBER SA615EA AND WHELEN INSTALLATION AND SERVICE MANUAL -												
DOC	DOCUMENT NUMBER 05131, REVISION D, DATED MAY 2015.											
3. A L0	3. A LOGBOOK ENTRY HAS BEEN MADE.											
4. NO	NTERFERENCE WAS NOTED UPON COMPLE	TION OF	SYSTEM TESTING.									
	ITINUED AIRWORTHINESS IS ASSURRED US			LATION AND SE	RVICE MANUAL.							
	ACTUAL WEIGHT AND BALANCE WAS PERFO											
	ISFACTORY OPERATIONAL TESTS WERE PE	RFORME	D.									
9. DET	AILS ON FILE AT ROTORCRAFT SUPPORT, IF	NC UNDER	WORK ORDER 15993.									
	□A	dditional Si	neets Are Attached									

Form Approved OMB No. 2120-0020

								ALIERAI		11/30/2	007		
U.S Departm Transportation		(Airfi	rame, Pov	verplan	ıt, P	, rol	peller, or A	ppliance)		Ī	For FAA Use Only	
Federal Avia													
and dis	sposition of thi	s form. Th	is repo	tries. See FAF ort is required I Federal Aviation	oy law (49	U.S.C	Арре С. 14	endix B, and AC (21). Failure to	43.9-1 (or subseque port can result in	ent revisi a civil per	on thereof) naity not to	exceed \$1,000	
4 4:6	U.S	-	Regist 510P	ration Mark PD				Serial No. 0400E					
1. Aircraft	Make		LL DOUGLAS						Model Series Series				
				egistration cen	ificate)				Address (As shown	•	ration certil	icate)	
2. Owner	1	Y OF OA		NU PTER UNI	т				Address 455 7TH S	<u>ST</u>		0.4	
		ICE HE	LICC	SOLITER SIMI					CityOAKLAND			State <u>CA</u>	
							3 5	or FAA Use Or	2ip <u>94607-3940</u>		Country		
	 						J. F	OI FAA OSE OI	<u>.</u>			-	
									and the said and				
4	I. Type							5. 0	nit Identification				
Repair	Alteration	ו	Unit			V	Make	е	Model			Serial Number	er
	\boxtimes	AIRFF	RAME		_				(As described in Item 1 above)		ove)		
		POWE	RPL	ANT									
		PROP	ELLE	ER .									
		APPLI	IANCI	E	Type Manufactu	ype							
						6	6. C	onformity State	ement				
	's Name and					В. к	Kind	of Agency					
	otorcraft Su		<u>1c.</u>				-	S. Certificated N				facturer	
	6425 Hart S	Street		0.4		무	-	oreign Certificate	·		C. Certific		
	n Nuys			State <u>CA</u>				ertificated Repai			YT2R33) I L	
	406	Country							enance Organizatio				
have	e been made	in accordar	nce wit	tion made to the th the requirent to the best of	ents of Pa	rt 43 (of th	n item 5 above a ne U.S. Federal	and described on th Aviation Regulation	e reverse as and tha	or attachm t the inform	nents hereto nation	
Extended roper 14 CFF App. B				Signature/ 06 J	Date of Au UL 2015		ed Ir	ndividual	PHILLIP G. DIF	IORE	121	(0. Frl)
							` 	val for Return			TY		
				s specified belo dministration a					nspected in the mai JECTED	nner pres	cribed by th	ne	
ВУ	FAA Fit St Inspector	andards		Manufacture	ſ		Ma	aintenance Orga	nization		Depar	n Approved by Canad tment of Transport	lian
	FAA Desig	gnee	X	Repair Station	on		Ins	spection Authori	zation		ther (Specif		
Certificate of Designation	n No.		1	nature/Date of 06 JUL 20		1 Indiv	vidua		IP G. DIFIORE	10	W/) Fl	

8. Description of Work Accomplished										
(If more space is required, attach additional sheets. Ide	entify with a	aircraft nationality and registration m	ark and date work completed.)							
	U.S.A.	N510PD	06 JUL 2015							
	Nationa	lity and Registration Mark	Date							
REMOVED THE INCANDESCENT AFT POSITION L		M THE HORIZONTAL STABILIZE	R ASSEMBLY.							
2. INSTALLED A WHELEN WHITE LED POSITION LIG										
SUPPLEMENTAL TYPE CERTIFICATE NUMBER SA										
NUMBER 05131, REVISION D, DATED MAY 2015.										
3. A LOGBOOK ENTRY HAS BEEN MADE.										
A LOGBOOK ENTRY HAS BEEN MADE. NO INTERFERENCE WAS NOTED UPON COMPLETION OF SYSTEM TESTING.										
5. CONTINUED AIRWORTHINESS IS ASSURRED US			ION AND SERVICE MANUAL							
6. AN ACTUAL WEIGHT AND BALANCE WAS PERFO		150 TE MENTHONED INOTALENT	JOHN SERVICE WINTONE.							
7. THE EQUIPMENT LIST WAS REVISED.	ZI MAIED,									
SATISFACTORY OPERATIONAL TESTS WERE PE		n								
SATISFACTORY OPERATIONAL TESTS WERE PE DETAILS ON FILE AT ROTORCRAFT SUPPORT, II										
9. DETAILS ON FILE AT ROTORCRAFT SUPPORT, II										
	NOTHING	5 TOLLOWS								
·										
•										
Па	ddition <i>a</i> l S	heets Are Attached								

Form Approved OMB No. 2120-0020

	•							ND ALTERA		11/30/2007		
U.S Departo Transportati Federal Avi	ion		(A	\irfr	ame, Pov	erplan/	it, P	ropeller, or <i>l</i>	Appliance)		For F	AA Use Only
Administrat												
and dis	sposition of	f this fo	orm. This	repo		y law (49	U.S.C		43.9-1 (or subseque report can result in a			
	Įυ	ational J.S.A.		egistr 10P	ation Mark				Serial No. 0400E			
1. Aircraft	M	lake ICDC	NNELI	_ DC	OUGLAS			Model 369E			Series	
			As shown OF OAl		egistration cert	ificate)			Address (As shown o	•	on certificate)	
2. Owner					ND PTER UNI	Г		Address455 7 TH ST CityOAKLAND				State <u>CA</u>
								zip <u>94607-3940</u>		Country	51818 <u>071</u>	
							3	3. For FAA Use O	nly			
	4. Type								Init Identification			O. J. I. W. and D. a.
Repair	Altera	ition		Ur	11t			Make	Model			Serial Number
	×]	AIRFRA	ME		_			(As described in Ite	em 1 above)	
]	POWER	RPLA	ANT							
			PROPE	LLE	R							
]	APPLIA	NCE	Ē	Type Manufactu	rer					
		-					6	. Conformity Stat	ement			
A. Agency							_	ind of Agency			r	
Name RC Address 16	otorcraft			<u>).</u>				U.S. Certificated			Manufacture Certificate N	
	an Nuys	<u>it Sut</u>	===		State CA			Foreign Certificat Certificated Repa			2R331L	<u> </u>
	1406		Country_	USA				<u> </u>	enance Organization			
D. I cer have	e been ma	e repair de in a	r and/or a	Iterati e with	ion made to th	ents of Pa	lentifie rt 43 c	ed in item 5 above	and described on the Aviation Regulations			
Extended r per 14 CFF App. B	range fuel				Signature/D		horize	ed Individual	PHILLIP G. DIFI	ORE	lik)	
					<u> </u>		7. Ap	proval for Return	to Service	 }	1	
					specified belo	_	_		nspected in the mann	ner prescrib	ed by the	
FAA Flt Standards Manufacturer					Maintenance Org	anization		Department	roved by Canadian of Transport			
	FAA De	esigne	e	X	Repair Statio	n		Inspection Author	ization	Other	(Specify)	4
Certificate or Designation No. VT2P3311 Certificate or Signature/Date of Authorized Individual Designation No. Designation No												

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

		ent ify with airc raft nationality and registration ma	ark and date work completed.)
		U.S.A. N510PD	06 JUL 2015
		Nationality and Registration Mark	Date
1.	THE MONITOR & MOUNT WERE FOUND PREVIOU	JSLY REMOVED.	
2.	FABRICATED A MOUNT FOR THE INSTALLATION	OF A MONITOR ON THE RIGHT SIDE OF TH	E INSTRUMENT PANEL.
	a. THE BASE OF THE MOUNT IS MADE USING:	2024-T3 ALUMINUM SQUARE TUBING, 1.0" >	(1.0" x .065" THICK, 2024-T3
	ALUMINUM PLATE .125" THICK, AND 7075-T6	ALUMINUM ANGLE EXTRUSION 1.25" x 1.25	5" x .125" THICK.
	b. THE RIGHT AND LEFT UPRIGHTS ARE MADE	FROM 2024-T3 ALUMINUM ANGLE EXTRUS	SION 1.0" x 1.0" x .190" THICK.
	c. A SUPPORT MADE FROM 2024-T3 ALUMINUI	M EXTRUSION .065" THICK IS ATTACHED TO	O THE RIGHT UPRIGHT.
	 d. THE MOUNT IS ASSEMBLED WITH AIRCRAF WITH FLAT BLACK PAINT. 	T HARDWARE AND WAS ALODINED, EPOXY	Y PRIMED AND TOP COATED
3.	THE MOUNT IS INSTALL ON EXISTING SUPPORTS	S FROM THE PREVIOUS MONITOR INSTALL	ATION, USING TWO .1875"
	DIAMETER STEEL PIT PINS. THE RIGHT UPRIGHT	T SUPPORT IS ATTACHED TO AN EXISTING	BRACKET ON THE RIGHT DOOR
	FRAME USING AIRCRAFT HARDWARE.		
4.	AN AIRBORNE DISPLAYS, MODEL CBO, REV 1; PA	/N AB-12-W-HD-SDI-N-T, 12.1", HD, NVG, TO	UCH SCREEN MONITOR WAS
	INSTALLED INTO THE FABRICATED MOUNT USIN	IG AIRCRAFT HARDWARE.	
5.	POWER TO THE MONITOR IS PROVIDED FROM T	THE BATTERY BUSS, THROUGH A LABELLE	D CIRCUIT BREAKER TO THE
	UNIT.		
6.	THE INSTALLATION OF THE MONITOR AND FABR	RICATION OF THE MOUNT WAS PERFORME	D REFERENCING: AB-12-W
	AIRBORNE DISPLAY MONITOR - USER'S GUIDE ,	REVISION E, DATED MARCH 2015; MD HELI	COPTERS INC., MAINTENANCE
	MANUAL, CSP-HMI-3, REVISION TR15-001, DATED	O 04 MAY 2015, CHAPTER 96; AEROCOMPU	TERS AND FLIR SYSTEMS
	INTERFACE DIAGRAMS; AC43.13-1B, CHANGE 1,	DATED 27 SEP 2001: CHAPTER 7, PARA 7-1	& 7-2; PARA 7-14 & 7-15; PARA 7-
	63 & 7-64; PARA 7-85 & 7-86; CHAPTER 10, PARA	10-1; CHAPTER 11, PARA 11-30 TO 11-32; PA	ARA 11-35 & 11-36; PARA 11-47 TC
	11-52; PARA 11-66 TO 1-68; PARA 11-76 & 11-77; P	PARA 11-85 TO 11-87; PARA 11-96 & 11-97; P	ARA 11-106 & 11-107; PARA 11-
	115 TO 11-118; PARA 11-146; PARA 11-155 TO 11-	158; PARA 11-167; PARA 11-185 TO 11-187 F	PARA 11-205 & 11-206; PARA 11-
	230 TO 11-233; AC43.13-2B, DATED MARCH 2008,	CHAPTER 1, PARA 100 TO 113 AND AC 20-1	168.
7.	A LOGBOOK ENTRY HAS BEEN MADE.		
8.	NO INTERFERENCE WAS NOTED UPON COMPLE	ETION OF SYSTEM TESTING.	
9.	INSTRUCTIONS FOR CONTINUED AIRWORTHINE		
	MOUNTINGS, ASSOCIATED WIRING, CABLES, CO	·	
	INTEGRITY, SECURITY, WEAR, CHAFING, ETC. S		THE AIRCRAFT PRIMARY
	STRUCTURE WITH REGARDS TO FATIGUE AND	STRESS, CRACKING, CORROSION, ETC.	
10.	AN ACTUAL WEIGHT AND BALANCE WAS PERFO	ORMED.	
11.	THE EQUIPMENT LIST WAS REVISED.		
12.			
	DETAILS ON FILE AT ROTORCRAFT SUPPORT, I		
		NOTHING FOLLOWS	
	·		

U.S Department of
Transportation
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Form Approved OMB No. 2120-0020

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U.S Departme Transportation		(/	۱irfr	ame, Pov	erplar/	ıt, P	ropeller, or	Appliance)		For F	AA Use Only	
Federal Aviat Administration												
and disp	position of thi	s form. This	геро		y law (49	U.S.C		C 43.9-1 (or subseque o report can result in a				
lor each		nality and R		ration Mark	TACE 1900	·/		Serial No. 0400E				
1. Aircraft	Make							Model 369E		Series		
2. Owner	CIT	Y OF OA	KLA	egistration cert ND PTER UNI	-	Address (As shown on registration certificate) Address 455 7 TH ST CityOAKLAND StateCA zip94607-3940 Country					State <u>CA</u>	
						3	3. For FAA Use	Only				
4,	Туре	_			r		5.	Unit Identification		1		
Repair	Alteration	n	Ur	nit			Make 	Mode	ľ		Serial Number	
		AIRFR	AME		_			(As described in It	tem 1 above)			
		POWE	RPLA	ANT								
		PROPE	LLE	R						1		
		APPLIA	NCE	.	Type Manufactu	ırer		_				
						. 6	. Conformity St	atement				
	s Name and					В. К	(ind of Agency					
· · · · · · · · · · · · · · · · · · ·	torcraft Su		<u> 2.</u>				U.S. Certificated	Mechanic	1 1	Manufacture		
	425 Hart S	Street		0.4			Foreign Certifica			Certificate N	0.	
	n Nuys			State_ <u>CA</u>		Ø	Certificated Rep	air Station	\ ^Y ''	2R331L		
Zip 914 D. I certi		Country_ pair and/or a			e unit(s) id	dentific		ntenance Organization a and described on the		tachments I	nereto	
have	been made	in accordan	ce wit		ents of Pa	art 43 d		al Aviation Regulations			3	
Extended ra per 14 CFR App. B	ange fuel Part 43				Date of Au JL 2015		ed Individual	PHILLIP G. DIF	IORE	216) 7	
							proval for Retur	· · · · · · · · · · · · · · · · · · ·				
				specified below dministration ar	_	_		s inspected in the man REJECTED	ner prescribe	d by the		
ву	FAA Fit St Inspector	andards		Manufacture	ſ		Maintenance Organization			Department	roved by Canadian of Transport	
D1	FAA Desig	nee	х	Repair Statio	n		Inspection Auth	orization	Other (Specify)		
Certificate or Signature/Date of Authorized Individual Designation No. VT2P2311								30				

8.	Description of Work Accomplished				
	(If more space is required, attach additional sheets. Iden	tify with a	ircraft nationality and registration	on mark and date work	completed.)
		U.S.A.	N510PD	0	6 JUL 2015
		National	ty and Registration Mark	D	ate
1	I. REMOVED THE INCANDESCENT LANDING LIGHT F	:POM THI	NOSE OF THE AIRCRAFT		
	2. INSTALLED A WHELEN PAR-46 SUPER-LED REPLA			-20 AT THE SAME LO	CATION
-	REFERENCING SEAPLANES NORTH, LLC., SUPPLE				
	LLC., INSTALLATION INSTRUCTIONS - DOCUMENT				20711-071-1011
3	B. A LOGBOOK ENTRY HAS BEEN MADE.		, ,,, ,, _		
	I. NO INTERFERENCE WAS NOTED UPON COMPLET	TION OF	SYSTEM TESTING.		
5	6. CONTINUED AIRWORTHINESS IS ASSURED USING			ONS FOR CONTINUE)
	AIRWORTHINESS FOR LED AIRCRAFT LIGHT SYS				
6	6. AN ACTUAL WEIGHT AND BALANCE WAS PERFOR				
7	7. THE EQUIPMENT LIST WAS REVISED.				
8	3. SATISFACTORY OPERATIONAL TESTS WERE PER	RFORME).		
g	DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN	C UNDER	WORK ORDER 15993.		
	······································	NOTHING	FOLLOWS		
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	∏ Adı	ditional SI	neets Are Attached		

Form Approved OMB No. 2120-0020

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U.S Depar Transporta			(A	Airfr	ame, Pov	/erplar	it, P	ropeller, or <i>i</i>	Appliance)		For FA	AA Use Only	
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and d	disposition	of this f	orm. This	repo	ies. See FAR rt is required be ederal Aviation	y law (49	u.s.c	ppendix B, and AC . 1421). Failure to	: 43.9-1 (or subsequer report can result in a	nt revisior civil pena	thereof) for ins Ity not to exceed	structions ± \$1,000	
		Nationa U.S.A		egistra 10P	ation Märk D				Serial No. 0400E				
1. Aircra		Make MCD(ONNELI	L DC	UGLAS				Model 369E		Series		
					gistration cert	ificate)		Address (As shown on registration certificate)					
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D. I ce	ertify that to	nade in a	ir and/or a	Iterati e with	ion made to th	ents of Pa	i dentific irt 43 c	ed in item 5 above	and described on the Aviation Regulations	reverse o and that t	r attachments h he information	ereto	
	rnished he		ue and co	nrect	to the best of			ed Individual			A 1 /	γ	
	FR Part 43				, -	JL 2015		o morridadi	PHILLIP G. DIFI	ORE	Ry 1	C.FU	
								proval for Return			11	7	
					specified belo ministration a				nspected in the mann EJECTED	er prescri	bed by the		
DV	FAA Inspe	Fit Stan	dards		Manufacture	-		Maintenance Org	anization		Department	oved by Canadian of Transport	
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Certificate Designati YT2R3	ion No.	-			ature/Date of 2010		Indiv		IP G. DIFIORE	Q	110	30	

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished		
(If more space is required, attach additional sheets. Idea	ntify with aircraft nationality and registration m	ark and date work completed.)
	U.S.A. N510PD	06 JUL 2015
	Nationality and Registration Mark	Date
 VERIFIED CORRECT INSTALLATION OF THE PREVENTION OF THE GNS-430 CONFORMS TO TOO TOO TOO TOO TOO TOO TOO TOO TOO	CING: GARMIN INSTALLATION MANUAL NU ISTALLATION MANAUL NUMBER 190-00180 ANUAL NUMBER 190-00094-00, REVISION I IUAL, CSP-HMI-3, REVISION TR15-001, DAT APTER 10, PARA 10-1; CHAPTER 11, PARA ARA 11-76 TO 11-77; PARA 11-85 TO 11-87;	IMBER 190-00140-02, REVISION T, 0-00, REVISION C, DATED 20 APR F, DATED 07 DEC 2004, FOR THE TED 04 MAY 2015, CHAPTER 96; 11-30 TO 11-32; PARA 11-35 & 11- PARA 11-89; PARA 11-96 TO 11-
100; PARA 11-106 & 11-107; PARA 11-115 TO 11-118 146; PARA 11-155 TO 11-158; PARA 11-167; PARA 1 12, PARA 12-37; AC 43.13-2B, DATED MARCH 2006 PARA 209; CHAPTER 3, PARA 300 TO 309; CHAPT 3. AN ACTUAL WEIGHT AND BALANCE WAS PERFORM 4. THE EQUIPMENT LIST WAS REVISED. 5. PERFORMED A COMPASS SWING AND UPDATED 6. A LOGBOOK ENTRY HAS BEEN MADE. 7. PERFORMED EMI, INTERFERENCE AND OPERATION COMPONENTS AND EQUIPMENT. 8. PERFORM ON AN ANNUAL BASIS AN INSPECTION CONNECTORS, HARDWARE, AND RELATED AIRCE SPECIAL ATTENTION SHOULD BE GIVEN TO THE STRESS, CRACKING, CORROSION, ETC. 9. DETAILS ON FILE AT ROTORCRAFT SUPPORT, INC.	I1-174 TO 11-175; PARA 11-185 TO 11-187; I 8, CHAPTER 1, PARA 100 TO 113; CHAPTE ER 11, PARA 1100 TO 1104 AND AC 20-138 RMED. COMPASS CORRECTION CARD. IONAL TESTS WITH NO ADVERSE AFFECT N OF THE EQUIPMENT MOUNTINGS, ASSO RAFT STRUCTURE FOR INTEGRITY, SECU- AIRCRAFT PRIMARY STRUCTURE WITH R C UNDER WORK ORDER 15993.	PARA 11-230 TO 11-233; CHAPTER 2, PARA 200 TO 203; PARA 207; D. TS ON PREVIOUSLY INSTALLED CLATED WIRING, CABLES, JRITY, WEAR, CHAFING, ETC. REGARDS TO FATIGUE AND
	NOTHING FOLLOWS	

Form Approved OMB No. 2120-0020 11/30/2007

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Federal Adminis	Aviation stration											
and	dispositio	n of this f	form. This	repo	ries. See FAR ort is required b ederal Aviation	y law (49 l	U.S.C.	ppendix B, and AC , 1421). Failure to (43.9-1 (or subsequent report can result in a c	it revision there civil penalty not	eof) for instructions to exceed \$1,000	
		Nationa U.S.A		egistr 10P	ration Mark			T I	Serial No. 0400E			
1. Aircr	raft	Make			OUGLAS				Model 369E		Series	
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Certifica Designa	ation No.				nature/Date of a		Indivi		IP G. DIFIORE	Md.	A Bul	

8.	Description of Work Accomplished			
	(If more space is required, attach additional sheets. Iden	tify with	aircraft nationality and registration i	mark and date work completed.)
		U.S.A.	N510PD	06 JUL 2015
		Nationa	ality and Registration Mark	Date
	 VERIFIED CORRECT INSTALLATION OF THE FOLL TWO TECHNISONIC ACCESS/A A711 SERIES A INSTRUMENT PANEL AND ONE INSTALLED IN DELIVERY OF THE PORT AVIONICS TRANS-CAL, MODEL SSD120-30N, ALTITUDE EN PANEL PEDESTAL. TECHNISONIC A790 LOAD HAILER CONTROLLE POWER SONIX PSAMP600 AMPLIFIER AND PSAFE. ASTRON N2412-24 VOLTAGE CONVERTER, MODEL THE INSTALLATION WAS VERIFIED REFERENCING 96RE189, REVISION 2.2, DATED JUL 2009, FOR THOUSE 16 DEC 2002 FOR THE BLOWER; TRANS-CAL DOCUMENTS. 	OWING, AUDIO PA THE LOV BLOWER NCODER ER IN TH AIR12A S JNTED L S: TECHN EE AUDIO IE PA CO	PREVIOUSLY INSTALLED EQUIF ANELS, ONE INSTALLED IN THE VER SECTION OF THE TILT PANE R, ON THE SHELF FORWARD OF I DIGITIZER ON THE FORWARD E LOWER SECTION OF THE TILT PEAKER UNDER THE CO-PILOT INDER THE CO-PILOTS FLOOR. UISONIC INSTALLATION AND OP D PANELS; TECHNISONIC INSTAL INTROLLER; LONESTAR DRAWII	PMENT: RIGHT LOWER PORTION OF THE EL. THE INSTRUMENT PANEL PORTION OF THE INSTRUMENT PANEL SEAT. ERATION MANUAL NUMBER LATION INSTRUCTIONS NUMBER NG NUMBER CRB6457-05, DATED ATED 08 DEC 2008 FOR THE
£ 5	ENCODER; POWER SONIX INSTALLATION AND US DATA SHEET FOR THE CONVERTER; MD HELICOP 04 MAY 2015, CHAPTER 96; AC43.13-1B, CHANGE 1 TO 11-32; PARA 11-35 & 11-36; PARA 11-47 TO 11-5 11-89; PARA 11-96 TO 11-100; PARA 11-106 & 11-10 135 TO 11-139; PARA 11-146; PARA 11-155 TO 11-15 230 TO 11-233; CHAPTER 12, PARA 12-37; AC 43.1 PARA 200 TO 203; PARA 207; PARA 209; AND AC 2 AN ACTUAL WEIGHT AND BALANCE WAS PERFOR 5. THE EQUIPMENT LIST WAS REVISED. 5. PERFORMED A COMPASS SWING AND UPDATED A LOGBOOK ENTRY HAS BEEN MADE. 7. EMI, INTERFERENCE AND OPERATIONAL TESTS NOT INSTALLED COMPONENTS AND EQUIPMENT. 8. PERFORM ON AN ANNUAL BASIS AN INSPECTION CONNECTORS, HARDWARE, AND RELATED AIRCUSPECIAL ATTENTION SHOULD BE GIVEN TO THE STRESS, CRACKING, CORROSION, ETC. 9. DETAILS ON FILE AT ROTORCRAFT SUPPORT, INCOMPONENTS.	TER'S IN I, DATEE 2; PARA 7; PARA 58; PARA 0-168. EMED. COMPA: WERE CI I OF THE RAFT ST AIRCRAI	IC. MAINTENACE MANUAL, CSP- 0 27 SEP 2001; CHAPTER 10, PAR 11-66 TO 11-69; PARA 11-76 TO 11-115 TO 11-118; PARA 11-120; 11-167; PARA 11-174 TO 11-175 ITED MARCH 2008, CHAPTER 1, SS CORRECTION CARD. OMPLETED WITH NO ADVERSE I EQUIPMENT MOUNTINGS, ASS RUCTURE FOR INTEGRITY, SEC IT PRIMARY STRUCTURE WITH	HMI-3, REVISION TR15-001, DATED RA 10-1; CHAPTER 11, PARA 11-30 11-77; PARA 11-85 TO 11-87; PARA PARA 11-123 TO 11-125; PARA 11- ; PARA 11-185 TO 11-187; PARA 11- PARA 100 TO 113; CHAPTER 2, AFFECTS ON PREVIOUSLY OCIATED WIRING, CABLES, CURITY, WEAR, CHAFING, ETC. REGARDS TO FATIGUE AND
	☐ Ac	ditional S	Sheets Are Attached	

Form Approved OMB No. 2120-0020 11/30/2007

Transpor Federal	U.S Department of Transportation (Airframe, Powerplant, F Federal Aviation Administration							ropeller, or <i>l</i>	Appliance)	1130/2007	For FAA Use Only		
and	disposition	n of this f	orm. This	repo		y law (49	U.S.C.		43.9-1 (or subsequer report can result in a				
		Nationa U.S.A		egistr 10Pl	ation Mark D				Serial No. 0400E				
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	ed range fu CFR Part 4					Date of Aut JL 2015		ed Individual	PHILLIP G. DIFI	ORE W	07		
							7. Ap	proval for Return	to Service	1/1			
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p.v.		Fit Stan	dards		Manufacture			Maintenance Org	anization		rson Approved by Canadian partment of Transport		
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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. C	Description of Work Accomplished		
(If more space is required, attach additional sheets. Iden	tify with aircraft nationality and registration	mark and date work completed.)
		U.S.A. N510PD	06 JUL 2015
	·	Nationality and Registration Mark	Date
1. 2. 3. 4.	CERTIFICATE NUMBER SR03265AT AND HELI-TEC ORIGINAL ISSUE, DATED 10 APR 2006. THE RIGHT HAND EXHAUST TAILPIPE WAS REPLA THIS STC. A LOGBOOK ENTRY HAS BEEN MADE. SATISFACTORY ENGINE RUN AND OPERATIONAL	CH INSTALLATION INSTRUCTIONS, DOC ACED WITH A SERVICEABLE UNIT, WHIC CHECK WAS PERFORMED.	CH WAS PREVIOUSLY MODIFIED BY
5.	REFER TO HELI-TECH DOCUMENT NUMBER 0121	04 FOR INSTRUCTIONS FOR CONTINUE	D AIRWORTHINESS.
6.	AN ACTUAL WEIGHT AND BALANCE WAS PERFO	RMED.	
7.	THE EQUIPMENT LIST WAS REVISED.	0 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	
8.	DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN		
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U.S Department of
Transportation
Federal Aviation

Form Approved OMB No. 2120-0020 11/30/2007

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1. Aircraft	Make		L DO	DUGLAS				Model 369E		Series		
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2. Owner	POLI	CE HEL	.ICO	PTER UNI	Τ		Address 455 7 TH ST CityOAKLAND StateCA Zip94607-3940 Country					
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Address 164	425 Hart St	reet					Foreign Certi	icated Mechanic		Certificate No.		
city <u>Var</u>	n Nuys			State <u>CA</u>		Ø	Certificated F	epair Station	YT:	2R331L		
Zip914	<u> 106</u>	Country_	<u>USA</u>	7			Certificated M	laintenance Organization				
have		accordan	e wit	h the requirem	ents of Pa	rt 43 d		ove and described on the eral Aviation Regulations				
Extended ra per 14 CFR App. B					Date of Aut JL 2015		ed Individual	PHILLIP G. DIF	ORE (11071		
						7. A p	proval for Ret	urn to Service		4000		
	to the authority ator of the Fed			•		_	ified in item 5 v	vas inspected in the mann	ner prescribe	d by the		
pv l	FAA Flt Star Inspector	ndards		Manufacture	r		Maintenance	Organization		Person Approved by Canadian Department of Transport		
BY	FAA Design	ee	Х	Repair Statio	on		Inspection Au	thorization	Other (Specify)		
Certificate on Designation	No.		1 -	nature/Date of 06 JUL 201		Indiv		IILLIP G. DIFIORE	(A7) A	10)	į	

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) Nationality and Registration Mark Date 1. REMOVED THE WHITE XENON STROBE LIGHT FROM THE TAIL AND THE RIGHT HAND & LEFT HAND INCANDESCENT POSITION LIGHTS FROM THE SIDES OF THE FUSELAGE. 2. INSTALLED A WHELEN WHITE LED STROBE LIGHT P/N 01-0771410-01 AT THE EXISTING LOCATION ON THE TAIL REFERENCING WHELEN INSTALLATION GUIDE - FORM NO. 14353B, DATED 05-29-2014. THE STROBE CONFORMS TO TSO C96a CLASS III. 3. WHELEN LED POSITION LIGHTS P/N'S 01-0771105-03 (R/H), AND 01-0771105-04 (L/H) WERE INSTALLED AT THEIR EXISTING LOCATIONS REFERENCING INSTALLATION GUIDE - FORM NO.14650B, DATED 07-24-2014. THE LIGHTS CONFORM TO TSO C30c TYPE I & II (R/H & L/H). 4. ADDITIONAL DATA REFERENCED FOR THIS INSTALLATION: MD HELICOPTERS INC. MAINTENANCE MANUAL, CSP-HMI-3, REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-35 & 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; P/11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA 11-115 TO 11-1118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-206; PARA 11-230 TO 11-2 CHAPTER 12, PARA 12-8; AC 43.13-2B, DATED MARCH 2008, CHAPTER 4, PARA 400 TO 406.	
Nationality and Registration Mark Date 1. REMOVED THE WHITE XENON STROBE LIGHT FROM THE TAIL AND THE RIGHT HAND & LEFT HAND INCANDESCENT POSITION LIGHTS FROM THE SIDES OF THE FUSELAGE. 2. INSTALLED A WHELEN WHITE LED STROBE LIGHT P/N 01-0771410-01 AT THE EXISTING LOCATION ON THE TAIL REFERENCING WHELEN INSTALLATION GUIDE - FORM NO. 14353B, DATED 05-29-2014. THE STROBE CONFORMS TO TSO C96a CLASS III. 3. WHELEN LED POSITION LIGHTS P/N'S 01-0771105-03 (R/H), AND 01-0771105-04 (L/H) WERE INSTALLED AT THEIR EXISTING LOCATIONS REFERENCING INSTALLATION GUIDE - FORM NO.14650B, DATED 07-24-2014. THE LIGHTS CONFORM TO TSO C30c TYPE I & II (R/H & L/H). 4. ADDITIONAL DATA REFERENCED FOR THIS INSTALLATION: MD HELICOPTERS INC. MAINTENANCE MANUAL, CSP-HMI-3, REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; P/11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-206 TO 11-206; PARA 11-230 TO 11-2	
 REMOVED THE WHITE XENON STROBE LIGHT FROM THE TAIL AND THE RIGHT HAND & LEFT HAND INCANDESCENT POSITION LIGHTS FROM THE SIDES OF THE FUSELAGE. INSTALLED A WHELEN WHITE LED STROBE LIGHT P/N 01-0771410-01 AT THE EXISTING LOCATION ON THE TAIL REFERENCING WHELEN INSTALLATION GUIDE - FORM NO. 14353B, DATED 05-29-2014. THE STROBE CONFORMS TO TSO C96a CLASS III. WHELEN LED POSITION LIGHTS P/N'S 01-0771105-03 (R/H), AND 01-0771105-04 (L/H) WERE INSTALLED AT THEIR EXISTING LOCATIONS REFERENCING INSTALLATION GUIDE - FORM NO.14650B, DATED 07-24-2014. THE LIGHTS CONFORM TO TSO C30c TYPE I & II (R/H & L/H). ADDITIONAL DATA REFERENCED FOR THIS INSTALLATION: MD HELICOPTERS INC. MAINTENANCE MANUAL, CSP-HMI-3, REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-35 & 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; P/11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-187 TO 11-187; PARA 11-200 TO 11-206; PARA 11-230 TO 11-206. 	
 POSITION LIGHTS FROM THE SIDES OF THE FUSELAGE. INSTALLED A WHELEN WHITE LED STROBE LIGHT P/N 01-0771410-01 AT THE EXISTING LOCATION ON THE TAIL REFERENCING WHELEN INSTALLATION GUIDE - FORM NO. 14353B, DATED 05-29-2014. THE STROBE CONFORMS TO TSO C96a CLASS III. WHELEN LED POSITION LIGHTS P/N'S 01-0771105-03 (R/H), AND 01-0771105-04 (L/H) WERE INSTALLED AT THEIR EXISTING LOCATIONS REFERENCING INSTALLATION GUIDE - FORM NO.14650B, DATED 07-24-2014. THE LIGHTS CONFORM TO TSO C30c TYPE I & II (R/H & L/H). ADDITIONAL DATA REFERENCED FOR THIS INSTALLATION: MD HELICOPTERS INC. MAINTENANCE MANUAL, CSP-HMI-3, REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; PARA 11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-205 TO 11-206; PARA 11-230 TO 11-2 	
REFERENCING WHELEN INSTALLATION GUIDE - FORM NO. 14353B, DATED 05-29-2014. THE STROBE CONFORMS TO TSO C96a CLASS III. 3. WHELEN LED POSITION LIGHTS P/N'S 01-0771105-03 (R/H), AND 01-0771105-04 (L/H) WERE INSTALLED AT THEIR EXISTING LOCATIONS REFERENCING INSTALLATION GUIDE - FORM NO.14650B, DATED 07-24-2014. THE LIGHTS CONFORM TO TSO C30c TYPE I & II (R/H & L/H). 4. ADDITIONAL DATA REFERENCED FOR THIS INSTALLATION: MD HELICOPTERS INC. MAINTENANCE MANUAL, CSP-HMI-3, REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; PARA 11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-205 TO 11-206; PARA 11-230 TO 11-2	
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REVISION TR15-001, DATED 04 MAY 2015, CHAPTER 96; FAR 91.205 (c)(3); AC43.13-1B, CHANGE 1,DATED SEPT 27, 2001: CHAPTER 10, PARA 10-1; CHAPTER 11, PARA 11-30 TO 11-32; PARA 11-35 & 11-36; 11-47 TO 11-52; PARA 11-66 TO 11-68; PARA 11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-205 TO 11-206; PARA 11-230 TO 11-2	
11-76 TO 11-77; PARA 11-85 TO 11-87; PARA 11-96 TO 11-97; PARA11-106 & 11-107; PARA 11-115 TO 11-118; PARA 11-125; PARA 11-146; PARA 11-155 TO 11-158; PARA 11-167; PARA 11-185 TO 11-187; PARA 11-205 TO 11-206; PARA 11-230 TO 11-2	\ D A
CHAPTER 12, PARA 12-8; AC 43.13-2B, DATED MARCH 2008, CHAPTER 4, PARA 400 TO 406.	
5. AN ACTUAL WEIGHT AND BALANCE WAS PERFORMED.	
6. THE EQUIPMENT LIST WAS REVISED. 7. SATISFACTORY OPERATIONAL TESTS WERE PERFORMED.	
8. A LOGBOOK ENTRY HAS BEEN MADE. 9. NO INTERFERENCE WAS NOTED UPON COMPLETION OF SYSTEM TESTING.	5 14
10. CONTINUED AIRWORTHINESS IS ASSURRED USING THE CURRENT MD HELICOPTERS, INC. INSPECTION PROGRAM AND REFERENCING THE ABOVE MENTHONED INSTALLATION GUIDES.	BY
11. DETAILS ON FILE AT ROTORCRAFT SUPPORT, INC UNDER WORK ORDER 15993 NOTHING FOLLOWS	. .

9

MAJOR REPAIR AND ALTERATION

Form Approved OMB No. 2120-0020

Electronic Tracking Number

II S Door	artment of						4 5				11/30/2	JU <i>1</i>	- 1			
Transpor	tation		(A	urtr	ame, Pow	erpian/	it, Pi	ropelle	opeller, or Appliance)					For FAA Use Only		
Fadaral Aviation Administration INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B,																
and	disposition	of this f	orm. This	repo		y law (49	U.S.C.			43.9-1 (or subsequer report can result in a						
			lity and Ro N510P		ation Mark					Serial No. 0400E						
1. Aircra	aft	Make MCD(ONNEL	DOL	JGLAS HE	LICOPT	ER			Model 369	Series E					
Name (As shown on registration certific CITY OF OAKLAND						ificate)	Address 455 7 TH STREET City OAKLAND							Stete_ <u>CA</u>		
						··	3	B. For FAA	Use Or	nly						
													· · · · · · · · · · · · · · · · · · ·			
	4. Type					ſ			5. U	nit identification						
Repa	ir Atte	eration		Ur	ıit			/lake		Model				Serial Numbe	er	
		\boxtimes		_				(As described in Ite	em 1 abe	ove)	-					
		POWERPLANT														
	PROPELLER					A										
			APPLIA	NCE	:	Type Menufacturer										
							6.	. Conform	ity State	ement						
•	ncy's Name						В. К	ind of Age	ncy				-			
Nama	ROTOR	CRAF	<u>r supp</u>	<u>OR1</u>	<u>ſ, INC.</u>			U.S. Cert	ificated N	Mechanic			nufacturer			
_	16425 H		STREET	-				Foreign C	ertificate	ed Mechanic		ificate No.				
	<u>VAN NU</u>	<u>YS</u>			State_ <u>CA</u>		Ø	Certificate	ed Repai	r Station		YT2R	331L			
Zip	<u>91406</u>		Country_	USA	<u>.</u>			Certificate	ed Mainte	enance Organization						
h	ave been r	made in	accordanc	e with		ents of Pa	rt 43 c			and described on the Aviation Regulations				reto		
Extended range fuel per 14 CFR Part 43 App. B Signature/Date of Authorized 07-06-2015 PHILLIP C									Wa	100), ;	FC.	_	·		
							7. Ap	proval for	Return	to Service						
			-		specified below Iministration are		_	ified in iten PROVED		nspected in the mann JECTED	ner pres	cribed by	y the			
BY		FIt Stan	dards		Manufacture	r		Maintenance Organization						ved by Canadi f Transport	an	
Dī	FAA	Designe	ee	Х	Repair Statio	ın		Inspection	n Authori	zation	Ot	her (Spe	ecify)		,	
Certificate or Signature/Date of Authorized Individual O7-06-2015 PHILLIP G. DIFIORE																

B. Description of	of Work Accomplished		
(If more space	e is required, attach additional sheets. Ic	dentify with aircraft nationality and registration n	nark and date work completed.)
		USA N510PD	07-06-2015
		Nationality and Registration Mark	D <i>a</i> te
COMPANY		REVISION E INSTALLED IN ACCORDANCE SR09074RC BY BIG VALLEY AVIATION INC.	
. MD MAIN R	OTOR BLADES CURRENTLY INSTALL	ED.	
. WEIGHT A	ND BALANCE NOT AFFECTED.		
. EQUIPMEN	T LIST PREVIOUSLY REVISED.		
	D AIRWORTHINESS IS ASSURED THE NO. 46 DATED 19 SEPTEMBER 2014 C	ROUGH SCHEDULED INSPECTIONS PER ME OR LATER REVISION	HELICOPTERS CSP-HMI-2
	NO	THING FOLLOWS	
	•		
			•
	П	Additional Sheets Are Attached	
	, <u></u>	A WARRANGE OFFICE A TELEVISION	

U.S Department of
Transportation
Federal Aviation
A -11 - 1 - 4 44

MAJOR REPAIR AND ALTERATION

Form Approved **Electronic Tracking Number** OMB No. 2120-0020 11/30/2007

(Airframe, Powerplant, Propeller, or Appliance) For FAA Use Only INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958) Serial No. Nationality and Registration Mark USA 0400E N510PD 1. Aircraft Model Make Series MCDONNELL DOUGLAS HELICOPTER 369 Ε Address (As shown on registration certificate) Name (As shown on registration certificate) CITY OF OAKLAND Address455 7TH STREET 2. Owner CityOAKLAND State CA zip94607-3940 Country USA 3. For FAA Use Only 5. Unit Identification 4. Type Repair Alteration Unit Make Model Serial Number \boxtimes (As described in Item 1 above) **AIRFRAME POWERPLANT PROPELLER** Type APPLIANCE П Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency Name ROTORCRAFT SUPPORT, INC. □ U.S. Certificated Mechanic ☐ Manufacturer C. Certificate No. Address 16425 HART STREET Foreign Certificated Mechanic YT2R331L VAN NUYS State CA X Certificated Repair Station Country_USA 91406 Zip Certificated Maintenance Organization D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. Signature/Date of Authorized Individual Extended range fuel per 14 CFR Part 43 07-06-2015 PHILLIP G. DIFIORE App. B 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED. REJECTED Person Approved by Canadian FAA Flt Standards Manufacturer Maintenance Organization Department of Transport Inspector BY Other (Specify) FAA Designee Repair Station Inspection Authorization Certificate or Signature/Date of Authorized Individual Designation No. 07-06-2015 PHILLIP G. DIFIORE YT2R331L

8.	Description of Work Accomplished
	(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
	USA N510PD 07-06-2015
	Nationality and Registration Mark Date
1.	REMOVED HELICOPTER TECHNOLOGY COMPANY TAIL ROTOR BLADES P/N 369D21641-503 S/N 6084-A943, A944, A945, AND A946 AND INSTALLED NEW HELICOPTER TECHNOLOGY COMPANY TAIL ROTOR BLADES P/N 500P3100-305 S/N C630, C631, C632 AND C633 IN ACCORDANCE WITH STC SRO1282LA.
2.	WEIGHT AND BALANCE NOT AFFECTED.
3.	EQUIPMENT LIST REVISED.
4.	NO FLIGHT MANUAL SUPPLEMENT REQUIRED FOR THIS INSTALLATION.
5.	CONTINUED AIRWORTHINESS IS ASSURED THROUGH SCHEDULED INSPECTIONS PER HELICOPTER TECHNOLOGY COMPANY MAINTENANCE MANUAL HTCM-002 REVISION LETTER C DATED 22 NOVEMBER 2004 OR LATER REVISION.
	NOTHING FOLLOWS
	·
	Additional Sheets Are Attached

U.S Depertment of
Transportetion
Federal Aviation
Adminintendina

MAJOR REPAIR AND ALTERATION (Airframe Powernlant Propeller or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

Transportetion Federal Avia Administrati	ation	()	AIFIF	aine, Pow	rerpian	il, Pi	ropeller, or	Арр папсе)		For FAA Use Only			
and dis	sposition of th	is form. This	s repo		y law (49	U.S.C.		C 43.9-1 (or subseque report can result in a					
	US/	onality and R	egistr 510					Serial No. 0400E					
1. Aircraft	Mak	е		DUGLAS H	ELICOF	TER		Model 369		Series E			
2. Owner		ne (As shown Y OF OAI		egistration certi ND	ificate)		Address (As shown on registration certificate) Address 455 7 TH STREET City OAKLAND Stete CA Zip 94607-3940 Country USA						
						3	3. For FAA Use (<u> </u>		<u>,</u>			
4	I. Type	+					5.	Unit Identification	<u> </u>				
Repair	Alteration	n	Ur	nit 	<u> </u>	N	Make	Model		Serial Number	_		
		AIRFR/	AIRFRAME					(As described in Ite	em 1 above)				
		POWE	POWERPLANT										
		PROPE	PROPELLER										
		APPLIA	NCE		Type Manufactu								
					L	6	. Conformity Sta	ltement	1				
A. Agency	's Name and	Address				T	(ind of Agency						
Name_RC	OTORCRA	FT SUPP	ORT	Γ, INC.		U.S. Certificated Mechanic			☐ Mai	☐ Manufacturer			
	6425 HAR	T STREET	Ξ				Foreign Certifica	ted Mechanic		ificate No.			
	<u>AN NUYS</u>			State <u>CA</u>		☐ Certificated Repair Station			YT2R:	331L			
Zip91	406	Country	<u>USA</u>	<u> </u>			Certificated Main	ntenance Organization					
have	e been made	in accordance	ce witl		ents of Pa	art 43 (and described on the A Aviation Regulations					
Extended range fuel per 14 CFR Part 43 App. B								11	(),3	J			
						7. Ap	proval for Retur	to Service) 				
				specified belo Iministration a		_	_	inspected in the mani REJECTED	ner prescribed by	, the			
DV.	FAA FIt S Inspector			Manufacture	r		Maintenance Or	ganization		rson Approved by Canadian partment of Transport			
BY	FAA Desi	gnee	х	Repair Statio	n		Inspection Author	onzation	Other (Spe	ecify)			
Designation	ertificate or esignation No. T2R331L Signature/Date of Authorized Individual 07-06-2015 PHILLIP G. DIFIORE												

В.	Description of Work Accomplished		
	(If more space is required, attach additional sheets. Ide	entify with aircraft nationality and registration	mark and date work completed.)
		USA N510PD	07-06-2015
		Nationality and Registration Mark	Date
1.	THIS FAA FORM 337 CORRECTS FAA FORM 337 OINSTALLATION OF WECO GENERATOR KIT IN ACBY CORRECTING THE SUPPLEMENTAL TYPE CEI	CORDANCE WITH SUPPLEMENTAL TYPE	TERS, INC. DATED FEB-19-1993 FOR CERTIFICATE NUMBER SH3195NW
2.	WEIGHT AND BALANCE NOT AFFECTED.		
3.	FOUND EQUIPMENT LIST PREVIOUSLY REVISED		
4.	FLIGHT MANUAL SUPPLEMENT NO.1 REVISION 1		LIGHT MANUAL.
5.	CONTINUED AIRWORTHINESS IS ASSURED THROREVISION NO.46 DATED 19 SEPTEMBER 2014 OR	OUGH SCHEDULED INSPECTIONS PER M	
	NOT	THING FOLLOWS	
		Additional Sheets Are Attached	

MAJOR REPAIR AND ALTERATION

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

U.S Dep	partment of		12	liefe	ame Pow	ernlan	t Pi	ropeller, or A	Appliance)	7 (730/2007	F F.	10.11 0-1	
Transportation Federal Aviation Administration				VII II	ame, rov	reipian	, .	ļ	ForFA	AA Use Only			
				_									
and	l dispositio	n of this t	form. This	repo	ries. See FAR ort is required be ederal Aviation	y law (49 l	U.S.C.	ppendix B, and AC . 1421). Failure to	43,9-1 (or subseque report can result in a	nt revision civil penalty	thereof) for ins y not to exceed	structions d \$1,000	
		Nationa U.S.A		egistr 10P	ation Mark D				Serial No. 0400E				
1. Aircr	raft	Make MCD	ONNELI	_ DC	OUGLAS				Model 369E		Series		
2. Owr	ner	Name ((As shown OF OAI	on re	egistration cert	Address Address CityOAk			Address (As shown of Address 455 7 TH ST City OAKLAND 2ip 94607-3940	DAKLAND State <u>CA</u>			
	-	<u> </u>			<u> </u>		3	. For FAA Use O			Country		
	4. Type		<u> </u>					5. U	Init identification				
Repa	air Al	teration		Ur	nit		N	/lake	Model			Serial Number	
	☐ ☐ AIRFRAME					_			(As described in Ite	em 1 above)		
	POWERPLANT												
	PROPELLER						*=**						
	APPLIANCE APPLIANCE					Type	rer						
							6.	. Conformity Stat	ement				
	ncy's Nam							ind of Agency			T		
	Rotorcr			<u> 2.</u>				U.S. Certificated			Manufacture Certificate No		
	<u>16425 l</u> Van Nu		reet		State CA		⊠	Foreign Certificate Certificated Repa			T2R331L		
	91406	10	Country	USA				<u> </u>	enance Organization				
D. I	certify that	made in	nir and/or a	Iterat	ion made to the	ents of Pa	tentifie rt 43 c	ed in item 5 above	and described on the Aviation Regulations	reverse or	attachments he information	nereto	
Extended range fuel Signature/Date of per 14 CFR Part 43 App. B Signature/Date of DJUL 2						Date of Aut	horize	ed Individual	PHILLIP G. DIFI	ORE	OX 18) Ful	
					**			provat for Return					
					specified belo Iministration a	K-			inspected in the manr EJECTED	ner prescrib	ed by Ne		
вү		A FIt Star pector	ndards		Manufacture	r		Maintenance Org	anization		Department	roved by Canadian of Transport	
וט	FAV	A Design	ee	X	Repair Statio	n		Inspection Author	Inspection Authorization Other (Specify)				
Certificate or Signature/Date of Authorized Individual Designation No. 06 JUL 2015						Indivi		IP G. DIFIORE	00	11 16	7		

YT2R331L

8. Description of Work Accomplished	· · · · · · · · · · · · · · · · · · ·		
(If more space is required, attach additional sheets. Ide	entify with	aircraft nationatity and registration	mark and date work completed.)
	U.S.A.	N510PD	06 JUL 2015
	Nationa	lity and Registration Mark	Date
1. VERIFIED CORRECT INSTALLATION OF THE FOL	LOWING	PREVIOUSLY INSTALLED EQUIP	PMENT:
a. SX-16 JUNCTION BOX - ATTACHED ON THE E	BULKHEAI	D BELOW AND AFT OF THE CO-	PILOT SEAT
b. SX-16 SEARCHLIGHT - ON THE LEFT HAND S	SIDE OF T	HE AIRCRAFT. THE SX-16 IS AT	TACHED TO A FLIGHT TRAILS
HELICOPTER HARD MOUNT P/N FTH 105-7, V	VHICH IS	APPROVED UNDER SUPPLEME	NTAL TYPE CERTIFICATE NUMBER
SH6080NM. THIS STC WAS USED TO INSTAL	L THE FLI	R SENSOR ON THE RIGHT HAN	D SIDE OF THE AIRCRAFT;
REFERENCE FAA FORM 337 DATED 08 APR 2	2008 FOR	DETAILS.	
c. SX-16 CONTROLLER - IN THE COCKPIT			
2. THE INSTALLATION WAS VERIFIED REFERENCIN	G: SPECT	ROLAB SX-16 OPEARATION MA	NUAL, DOCUMENT NUMBER 031734,
REVISION E, DATED 09 JUN 2010; MD HELICOPTE	R'S INC. I	MAINTENACE MANUAL, CSP-HM	I-3, REVISION TR15-001, DATED 04
MAY 2015, CHAPTER 96; AC43.13-1B, CHANGE 1,	DATED 27	SEP 2001; CHAPTER 10, PARA	10-1; CHAPTER 11, PARA 11-30 TO
11-32; PARA 11-35 & 11-36; PARA 11-47 TO 11-52;	PARA 11-	66 TO 11-69; PARA 11-76 TO 11-7	77; PARA 11-85 TO 11-87; PARA 11-
96 TO 11-100; PARA 11-106 & 11-107; PARA 11-115	TO 11-11	8; PARA 11-123 TO 11-125; PAR/	A 11-135 TO 11-139; PARA 11-146;
PARA 11-155 TO 11-158; PARA 11-167; PARA 11-17	² 4 TO 11-1	75; PARA 11-185 TO 11-187; PAR	RA 11-230 TO 11-233; AC 43.13-2B,
DATED MARCH 2008, CHAPTER 1, PARA 100 TO 1	13; AND	AC 20-168.	
3. AN ACTUAL WEIGHT AND BALANCE WAS PERFO	RMED.		
4. THE EQUIPMENT LIST WAS REVISED.			
5. PERFORMED A COMPASS SWING AND UPDATED	COMPAS	SS CORRECTION CARD.	
6. A LOGBOOK ENTRY HAS BEEN MADE.			
7. PERFORMED EMI, INTERFERENCE AND OPERAT	IONAL TE	STS WITH NO ADVERSE AFFE	CTS ON PREVIOUSLY INSTALLED
COMPONENTS AND EQUIPMENT.			
8. REFER TO SPECTROLAB SX-16 INSPECTION AND	MAINTE	NANCE SCHEDULE FOR CONTIL	NUED AIRWORTHINESS.
9. DETAILS ON FILE AT ROTORCRAFT SUPPORT, IN			
	NOTHIN	G FOLLOWS	······································
•	•		•
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		~~	
•			
A	dditional S	Sheets Are Attached	

MAJOR REPAIR AND ALTERATION

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

II S Den	nartment of								- A I!	11/00/2						
U.S Department of Transportation Federal Aviation			(Airframe, Powerplant, Propeller, or Appliance)									For FAA Use Only				
Federal Adminis																
and	l dispositio	n of this f	orm. This	repo		y law (49	U.S.C		AC 43.9-1 (or subseque to report can result in a							
		Nationa USA		egistr	ation Mark			-	Serial No. 0400E							
1. Aircr	raft	Make MCD(ONNELI	_ DC	OUGLAS H	ELICOP	TER		Model 369			Series E				
Name (As shown on registration certificate CITY OF OAKLAND						ificate)	Address (As shown on Address 455 7 TH STF CityOAKLAND Zip94607-3940					-				
							3	. For FAA Us	Only							
	4. Type					ſ		5	. Unit Identification							
Repa	air Alt	teration		Ur	nit 			/lake	Model			Serial Number				
]	\boxtimes	ME			(As o			As described in Item 1 above)							
]		POWERPLANT													
	PROPELLER															
]		APPLIA	NCE		Type Manufactu	Type Manufacturer									
							6	. Conformity S	Statement							
A. Age	ncy's Nam	e and Ad	dress				В. К	ind of Agency								
Name	ROTOR	CRAF	<u>r supp</u>	<u>ORT</u>	<u> , INC.</u>			U.S. Certificat	ed Mechanic		☐ Ma] Manufacturer				
	16425 H		STREET	-				Foreign Certifi	cated Mechanic			ificate No.				
	VAN NL	<u>JYS</u>			State_ <u>CA</u>		⊠	Certificated R	epair Station		YT2R	331L				
Zip <u>91406</u> Country <u>USA</u>								Certificated M	aintenance Organization							
۲	have been	made in	accordanc	e with	h the requirem	ents of Pa	ırt 43 (ed in item 5 about the U.S. Fede	ive and described on the eral Aviation Regulations	reverse and tha	or attac it the info	hments hereto ormation				
Extended range fuel per 14 CFR Part 43 App. B Signature/Date of Authorize O6 JULY 2015 PHILL									ORE DRE	NA	(0)	341				
							7. Ap	proval for Ret	urn to Service							
			_		specified belo Iministration a			ified in item 5 w PROVED	as inspected in the man	nergres	cribed by	y the				
DV		A Fit Stan	dards		Manufacture	Г <u></u>		Maintenance (Organization			rson Approved by Canadian partment of Transport				
BY	FAA	A Designe	e	Х	Repair Statio	n		Inspection Au	horization	0	ther (Spe	ecify)				
Certificate or Signature/Date of Authorized Individual Designation No. O6 JULY 2015 PHILLIP G. DIFIORE								VO 1-1	$\overline{()}$, 0					

8. Description of Work Accomplished		
(If more space is required, attach additional sheets. Ide	entify with aircraft nationality and registration ma	rk and date work completed.)
	USA N510PD	06 JULY 2015
	Nationality and Registration Mark	Date
	•	
 VERIFIED PREVIOUS INSTALLATION OF AERONA ACCORDANCE WITH SUPPLEMENTAL TYPE CER 	UTICAL ACCESSORIES, INC LOCKING FUEL TIFICATE SR00087AT.	CAP ASSEMBLY IN
2. WEIGHT AND BALANCE NOT AFFECTED.		
3. EQUIPMENT LIST PREVIOUSLY REVISED.		
NO٦	THING FOLLOWS	
	/	
	Additional Sheets Are Attached	

6
US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance

	or FAA Use Only
2/28/2011	
Form Approved OMB No. 2120-0020	Electronic Tracking Number

Federal Avia Administrat	ation	(Milliali	16, F	owerpiant, Proj	μŧ	E11 6 1	i, or App	iaiice)				
instructio		ition of this	form.									osequent revision thereof) for each
		y and Regis		Mark				Serial No.				-
1. Aircraft	N510PE)				0400E						<u> </u>
i. Aircraft	wake							Model				Series
			_	LAS HELICOPT	Έ	R		369E				
	Name (As	s shown on	registr	ation certificate)				Address (As shown on registration certificate) Address 455 7TH STREET				
2. Owner	ì							city Oaklar		IKEE	ـــــــــــــــــــــــــــــــــــ	State CA
	City of 0	Dakland						Zip 94607			Cou	ntry US
					3	3. Fo	r FAA Use	Only				
		Regu	lreme ect to	identified herein ints and is appropriate interest in the interest in the identified herein into a propriate interest in the interest int	VO	/ed	only for t	he above des authorized in FA	cribe AR 43	d air	craft,	
4. 1	Гуре	<u> </u>		<u> </u>	5	5. Un	nit identifica	ition				
Repair	Alteration	Unit		Ma	ske	e			Mod	el		Serial No.
	V	AIRFRAM	E			(As described in Item 1 above)						
		POWERP	LANT									
		PROPELL	.ER									
		APPLIAN	CE	Type Manufacturer								
				•	5. (Con	formity Sta	tement				
A. Agency's	s Name and A	ddress			I	B. K	ind of Agent	у	•			
Name Aeri	ial Avionics 0 john Monto	nomen/ Dr			L	1		ated Mechanic			-	nufacturer
-	o john Monto	Joinery Dr	•	State CA	1	. -		ficated Mechanic Repair Station			C. Cert	ificate No.
zip 951		untry U.S.			F	Certificated Maintenance Organization 7IAR379				79B		
D. I cert	tify that the rep	air and/or a	e with	on made to the unit(the requirements of the best of my kno	P	art 4	3 of the U.S	n 5 above and de 5. Federal Aviation	scribe n Reg	ed on t	he rever	se or attachments hereto nat the information
Extended r per 14 CFF App. B			Signa	uture/Date of Authori		1	teles (18	1.	201	/}
Pursuant	t to the authorized	ority given	perso	ns specified below	_		unit identi					he manner prescribed by the
	FAA Flt. Stand		T	ninistration and is		Mai	ntenance O	Approved rganization	<u>'</u>		ns Appro	ved by Canadian Transport
BY	FAA Designee	. /		air Station		Inst	pection Auth	orization	Othe	r (Spe		· · · · · · · · · · · · · · · · · · ·
Certificate of Designation			Signa	ature/Date of Author	ize	ed In	ndividual		-			
7IAR379			15	ful of	2	h	THE	07	_	18	/-0	7012

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

Description of Work Accomplished
 (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N510PD

7/02/2012

Nationality and Registration Mark

Date

Page 1

- 1. No equipment was removed from the aircraft.
- 2. Follow on field approval for the installation of a Technisonics TDFM7300 FM Transceiver P/N TDFM73L148 STC SR02379NY. This aircraft is not listed on the STC.

Installed Technisonics TDFM73L148 FM Transceiver in the aircraft center pedestal. Circuit protection is provided by a 7.5 amp circuit breaker labeled "FM Radio" mounted to the existing avionics circuit breaker panel. The TDFM7300 is connected to a CI292-3 VHF Antenna installed at fuselage station 45", a FLX3050B VHFLO Antenna installed at fuselage station 65, and a CI275 UHF Antenna installed at fuselage station 63.

- 3. CONTROL & OPERATION: Refer to pilot's operating handbook provided to the pilot. Refer to TDFM7300 Installation and Operating Instructions TiL Document No. 08RE389 Rev. A Issue 7, dated Aug 2010 or later revision provided in the aircraft maintenance records.
- 4. SERVICING: None Required.
- 5. MAINTENANCE INSTRUCTIONS: Periodic maintenance is not required. Refer to this FAA Form 337. Inspect in accordance with FAR 43, appendix D.
- 6. TROUBLESHOOTING INFO: Refer to this Form 337. Major troubleshooting should refer to a qualified FAA certificated avionics personnel.
- 7. REMOVAL AND REPLACEMENT: Removal and replacement should be done by trained qualified and FAA certificated avionics personnel. Refer to this Form 337.
- 8.DIAGRAMS: Refer to TDFM7300 Installation and Operating Instructions TiL Document No. 08RE389 Rev. A Issue 7, dated Aug 2010 or later revision provided in the aircraft maintenance records.
- 9. SPECIAL INSPECTION REQUIREMENTS: N/A
- 10. APPLICATION OF PROTECTIVE TREATMENTS: N/A
- 11.DATA: Refer to TDFM7300 Installation and Operating Instructions TiL Document No. 08RE389 Rev. A Issue 7, dated Aug 2010 or later revision.
- 12. LIST OF SPECIAL TOOLS: N/A
- 13. FOR COMMUTER CATEGORY AIRCRAFT: N/A
- 14. RECCOMMENDED OVERHAUL PROCEDURES: N/A

-----Last Item This Page, See Page 2------

Additional Sheets Are Attached

U.S. Department of Transportation
Federal Aviation

FAA Form 337 (10-06)

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance

Form Approved OMB No.2120-0020	Electronic	Tracking	Number
2/28/2011			
F	0 0 11-0 0	-I	

of Transpo		(Airtr	ame, Powerplant, Pro	ope	eller, or A	ppliance)				
Federal A Administr							•	1			
INSTRU instructi	JCTION ions and	S: Print or type disposition of to tion (49 U.S.C.	this for	tries. See Title CFR §43. m. This report is required 01(a)).	9, F I by	Part 43 Appe law (49 U.S	endix B, and AC 43 5.C. §44701). Fail	3.9-1 (or sul ure to repor	bsequer t can re	nt revision thereof) for sult in a civil penalty for	
1. Alrcraft Nationality and Registration Mark N510PD							Serial No. 0400E		·,·		
	Make MDHI						Model 369E			Series	
		Name (As shown	on reg	istration certificate)			Address (As show	vn on registra	tion certifi	cate)	
2. O	wner	City Of Oaklan					Address 455 7 to Oaklan			State Ca	
		Police Helicop	lei Oili	•			Zip 94607		untry US		
				3. For	F#	AA Use Only	/				
	4. Type			5. Uni	t ld	entification					
Repa	air Alt	eration U	Init	Make			Model	ļ		Serial No.	
		AIRFR	AME .			-	(As described in i	tem 1 abov	re)		
		POWE	RPLAN	п							
		PROP	ELLER								
		APPLI	ANCE	Type Manufacturer							
	l					rmity Stater	nont	 -			
A. Age	ency's N	ame and Addre	ss	0. 001	<u> </u>	B. Kind of					
Name		ley Aviation Inc.			\vdash	U.S. Certifica		· ·		Manufacturer	
Address	7535 L	ndbergh St.			一	Foreign Certif	Foreign Certificated Mechanic			ertificate No.	
City	Stockto	on		State Ca '	X	Certificated R	epair Station		WG3R944L		
Zip	95206	Countr	y <u>U</u> S	SA		Certificated M	laintenance Organizat	ion			
hei	reto hav ormation	e been made ir I furnished here	accoi	eration made to the unit(s' dance with the requireme and correct to the best	nts of	of Part 43 o my knowled	f the U.S. Federal				
	I range fue Part 43 Ap	p. B	Signati	we/Date # Suthorized Ind	خ	911		Paul McKenz	zie 03/2	20/12	
				-		or Return T					
	ne Admi	nistrator of the		persons specified below, al Aviation Administration a				REJECTE	D		
BY L		A Flt. Standards pector		Manufacturer		Maintenan	ce Organization			ved by Canadian of Transport	
		A Designee	х	Repair Station			Authorization	Other (S	pecify)	·	
Certifica	ite or Des	ignation No.		Signature/Date of Author		d Individual					
WG3R9	441			When Allha	. Ĉ	12	Paul McKenzie	03/20/12			

8. Description of Work Accomplished (If more space is required, attach additional sheet	s. Identify with aircraft nationality and registration ma	rk and date work completed.)
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N510F	סי	03/20/12
Nationa	lity and Registration Mark	Date
Horizo	ontal Stabilizer Installation	
Installed Aerometals Inc. Horizontal Stabilizer Assert List (MDL) #33 Rev. A, dated 1/23/02 or latest revision SH6015NM.	nbly P/N 421X-087-905 S/N 1563 in accordance with on, Aerometals Inc. AMI-20 Initial revision, dated Aug	Aerometals Master Drawing ust 31, 2000, and STC #
Instructions for Continued Airworthiness for stabilizer Balance change negligible.	r P/N 421X-087-905 S/N 1563 are contained in Aeron	netals AMI-20. Weight and
		:
-	Ϋ́ι	
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<u> </u>	Additional Sheets Are Attached	

6	
U.S. Departme of Transportati	

MA JOR REPAIR AND ALTERATION

Form Approved OMB No.2120-0020 11/30/2007	Electronic Tracking Num	Del
Earl	EAA Lico Only	_

U.S. Department of Transportation (Airframe, Powerplant, Pr								For FAA Use Only		
Federal Aviat Administratio	tion	<i>(,,</i>	· · · · · · · · · · · · · · · · · · ·			.pp				
instructions	s and dispos	nt or type all e sition of this f O U.S.C. 4630	entries. See Title CFR 43.9 form. This report is require 01(a)).	9, F d b	Part 43 Appe by law (49 U.	ndix B, and AC 43.9-1 S.C. 44701). Failure	1 (or sub: to report	sequer can re	nt revision thereof) for sult in a civil penalty for	
1. Alrcraft Nationality and Registration Mark N510PD				Serial No. 0400E				-		
	Make MDHI					Model . 369E			Series	
2. Own	er City of	As shown on re Oakland Helicopter Ur	gistration certificate) nit	Address (As shown on registre Address 455 7 th. Street City Oakland Zip 94607 Co				State Ca.		
	<u> </u>		3. Fo	r F	AA Use Oni	y				
			Jos data identified herein co vaquirements and is approved chinformity inspection by a purious DATE	9 8	les with the application describe carried in FAR	licable alrworthiness and aircraft, subject to 43, Section 43.7)			
4.	Туре		5. Un		dentification					
Repair						Model		Serial No.		
	AIRFRAME				(As described in item 1 above)					
y .										
		PROPELLER								
		APPLIANCE	Type Manufacturer			0				
			6 Co.	nto	rmity State	ment				
A. Agency	's Name and	d Address	0. 001	T	B. Kind of					
Name Bi	g Valley Aviatio	on Inc		T	U.S. Certifica	ted Mechanic	, _		Manufacturer	
	35 S. Lindberg	h St.			Foreign Certi	ficated Mechanic		C. C	ertificate No.	
_	ockton 206	Country U	State Ca.	×		Repair Station	pair Station		NG3R944L	
D. I certify hereto	that the rep	pair and/or alt	teration made to the unit(s	nts	lentified in its s of Part 43 c	of the U.S. Federal Av	ribed on iation Re	the rev	verse or attachments	
Extended rang	e fuel per		rue and correct to the best ure/Date/of Authorized Ind	/=		ge.			· · · · · · · · · · · · · · · · · · ·	
14 CFR Part 4	3 App. B	J 1/2	Muster	ડ	or Return T	Paul McKenzie	04/08/0	08 04	/08/08	
Pursu	ant to the a	uthority giver	a persons specified below, al Aviation Administration	the	e unit ideatifi	ed in item 5 was insp	ected in		anner prescribed by	
BY BY	FAA Fit. Stan Inspector		Manufacturer	an (ce Organization	Pers	son App	oved by Canadian of Transport	
	FAA Design	nee X	Repair Station		Inspection	Authorization	iher (S	pecify)		
Certificate or	Designation N	o.	Signature/Date of Autho	rize	ed Individual					
NG3R944L			Mus Muss	4	75	Paul McKenzie (04/08/08			

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach addition	nal sheets. Identify with aircraft nationality and r	egistration mark and date work completed.)
9 (A T)	N510PD	04/08/08
- 1 1 1 Commonwealth	Nationality and Ragistration Mark	Date

N510PD Actt. 12042.0

U-8500 Flir System Installation

Ramoved existing Flir system U2000 camera from the right hand side of hellcopter, ECU from the rear cabin compartment, harness assembly, power cables, and hand controller from the interior of the halicopter which was installed and approved on 10/05/90 and has flown on this helicopter for 18 years trouble free.

Using Flight Trails Inc. STC # SH6080NM mount P/N FTH105-8 we installed the Flir U8500 day/night/infrared camera to the airframe using the airframe jack hard point at station 96.9 on the right hand side of the helicopter. The Flir 8500 system is lighter and has a smaller profile than the U-2000 system that was removed which will reduce the loads imposed on the helicopter. The electrical harness is routed from the camera gimbal along side the extenor of the airframe to a cannon plug at station 76.5 on the right hand side of the halicopter where it enters the airframe and continues to the Flir ECU. The Flir ECU is located in the center of the rear cabin bulkhead at station 78.5 on aircraft center line using the existing U2000 ECU mount assembly. The Flir ECU is protected through a 15 amp circuit breaker labled Flir System. All Flir system electrical harness was constructed by Flir Systems and the system is a plug and play system upon installation of the hard mounted system components.

Installad Universal Searchlight LLC. Slass System to allow slaving of the Flir camera and the SX-16 Night sun in the same track at the same time. The Slass System Electronic control unit P/N XX-7120 is mounted under the co-pllot seat at station 70.0 using standard AN hardware. The system hand controler is located in the foward cabin area at station 80.0 for easy access by the observer in the right seat location. Control cables are routed from the Flir ECU and the SX-16 J box to the Slass ECU using standard AN hardware.

Installed Broadcast Microwave Services, Inc helicopter microwave downlink system to allow live video from the Flir camera system to a base station on the ground. Mounted the control panel HCP-50 P/N 800205217 in the center console of the Instrument panel at station 57.0 Installed the Antenna P/N 120103103 and antenna actuator assembly P/N 800169304 on the right hand skid tube foward of the right foward gear leg at station 56.85 longitudinal and station 48.75 lateral. The transmitter BMT75-9P P/N 801-3454800 was mounted under the right hand cabin floor at station 45.0 on an existing mounting tray using standard AN hardware. The antenna cable was routed from the transmitter to the antenna on the right skid using standard AN hardware and MS adel clamps to secure the cable assembly. Routed the antenna actuator control harness from the control panel to the antenna actuator located on the right skid tube using standard AN hardware and MS adal clamps. The BMS transmitter is protected by a 5 amp circuit breaker labeled BMS TX and the BMS antenna actuator is protected by a 2 amp circuit breaker labeled BMS ACT.

Installad Aerocomputer X3 Digital Video Recorder (DVR) above Flir ECU at station 78.5 on center line of aircraft to allow recording of the audio and video from the Flir system 8500 and the Aerocomputer LE5000 maping system. Installed in accordance with aerocomputer X3 DVR Installation Manual Revision 3.

A latter of no objection to external equipment mounting from MD Helicopters Inc. is attached for your review.

The helicopter will be test flown in accordance with FAR 91.407 (b) by a properly rated pilot to determine if this installation has appreciably changed the flight characteristics of the helicopter, and a log book entry will be made of the operational test and the results noted.

All alectrical installations were conducted in accordance with AC43.13-1B Chapter 11, Section 3, 4, 5, 6, 7, 8, 9, 10, 11, 12, 15, 16, 17, 18, Chapter 12 Sections 1, 2, and 3. Structural requirements were complated in accordance with AC43.13-2A Chapter 1 and 2.

Adjusted aircraft equipment list and waight and balance data to reflact this installation.

Instructions for continuad airworthiness are as follows:

(1) Introduction: The Flir day/night camera and BMS microwava downlink is to be used for law enforcement and crime fighting. This system allows the ground support personal commanders to make tactical decisions basad on real time video of the crima scene.

X Additiona	Sheets Are	e Attached
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3	· **;. · · · · · · · · · · · · · · · · · · ·	N510PD		111111		04/08/08				
		Nationality and Regis	tration Mark		1	Date				
	(2) Description: As described in the above	e text.								
	(3) Control: Not Applicable.									
	(4) Servicing Information: Flir System 850 attachment and operational status can by the manufacturer. Only general sec field. The Slass system can only be in equipment repairs must be completed	n be completed in the curity of attachment ar spected for general se	field. BMS micro nd operational st	wave systen atus can be	n is to be se completed in	rviced n the				
•	(5) Maintenance Instructions: All system of manufacturer. Perform visual inspection	equipment repairs and on of systems for secu	I updates must burity of attachme	e completed nt and gener	by the al condition.	green thing perfects. That the co-				
	(6) Troubleshooting Information: Reference	ce Flir System 8500 s	ervice and main	tenance mar	ual 19208-9	940				
	(7) Removal and installation: Must be installed by a properly rated mechanic with a log book entry.									
	(8) Diagrams: Reference Flir System 850	00 Installation Manual	19208-904.							
	(9) Special inspections: Not applicable:									
	(10) Application of protective treatment: N	vot applicable.			, 4.					
•	(11) Data: Not applicable.	,	·			en e	, ,			
	(12) List of special tools: Not applicable.		K® Colorphis koja KK olorphis koja K		٠.					
	(13) For commuter catagory aircraft: Not	applicable.								
	(14) Recomended Overhaul periods: Flir recomendations.	camera to be serviced	in accordance	with the man	ufacturers					
	(15) Airworthiness Limitations section: No	ot applicable.								
	(16) Revision status: To revise this ICA s an a revised ICA. Once the revision I revision and the date of the revised	has been accepted a	cal FSDO with a maintenance ent	copy of the ry will be ma	revised FAA de, identifyi	A 337 ing the				
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DESCRIPTION:

5107489559

FAA requests through BIG VALLEY AVIATION a letter of no objection to external equipment mounted on 369 Series helicopters.

BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR installation from FI IR 2000 to FI IR BIG VALLEY AVIATION upgraded the FI IR INSTALLATION
BIG VALLEY AVIATION upgraded the FLIR installation from FLIR 2000 to FLIR 8500 which is a lighter and more aerodynamic unit onto the OAKLAND Police Department helicopters.

REPLY:

MD HELICOPTERS has reviewed the request and we have no technical objection to external equipment mounted as described above in accordance with the intent of the original installation.

John M. Kerr

Field Service Representative

MD Helicopters Inc.

4555 E. McDowell Rd.

Mesa, Arizona

85215

602-284-7479

Section 1985 . . • .



DESCRIPTION:

FAA requests through BIG VALLEY AVIATION a letter of no objection to external equipment mounted on 369 Series helicopters.

BIG VALLEY AVIATION upgraded the FITE installation from FLIE 2000 to FLIE as

BIG VALLEY AVIATION upgraded the FLIR installation from FLIR 2000 to FLIR 8500 which is a lighter and more aerodynamic unit onto the OAKLAND Police Department helicopters.

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MD Helicopters Inc. 4555 E. McDowell Rd.

Mesa, Arizona

85215

602-284-7479

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U.S. Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020

For FAA Use Only

Office Identification

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

excee	d \$1,0	00 for each such v	olati	on (Section 901	Federal A	viati	ion Act of 1958).		, to ropoit t			oridity flot to	
1. Ai	ircraft	Make McDonnel D	ougl	as Helicopter		Model 369E								
		Serial No. 0400E					Nationality and Registration Mark N510PD							
2. 0	wner	Name (As shown on registration certificate) City of Oakland Police Department				,	Address (As shown on registration certificate) 455 7th Street Helicopter Aviation Unit Oakland, CA 94607						•	
					3. F	or f	FAA Use Only							
					The data requirement conformation	identents atty Ins	rified herain compliand is approved for the pection by a person at the pect	es vith	the applicable and applicable app	e ainvorthiness creff, subject to section 43.7	h	7 .		
						DATE		NATUR	RE	OAK-FEGO				
				4. Un	t Identific	atio	on					5. 1	уре	
	Unit		Mak	e			Model		s	erial No.	,	Repair	Alteration	
AIRFF	RAME	E(As described in item 1 above)									x			
POWE	ERPLAN	vr :												
PROP	ELLER										*			
APPLI	ANCE	Type , Manufacturer												
					6. Cor	ıfor	mity Stateme	nt						
A. Ag	ency's	Name and Addre	ss				B. Kind of Ag	ency			C. C	ertificate No.		
	vionics					\bigsqcup	U.S. Certificated	.S. Certificated Mechanic				WK3R948L		
	Earhar Ind, CA	t Rd \ 94621					Foreign Certificated Mechanic							
	-					Ľ	Certificated Repair Station							
he	ereto h	that the repair and/ ave been made in ion furnished herei	acco	rdance with the	requireme	nts	of Part 43 of th	e U.S	ove and de S. Federal	scribed on Aviation Re	the revegulation	erse or attac ons and that t	hments he	
Date 7/5/20	06						ire of Authorize							
				7		al fo	or Return To S	ervi						
	Pursua	ant to the authority ministrator of the F	give	n persons specif	fied below.	the	unit identified	in ite		spected in		anner prescr	ibed by	
BY	ı	FAA Fit. Standards		Manufacturer		T	inspection A		ization	Other (S		· · · · · · · · · · · · · · · · · · ·		
		FAA Designee	Х	Repair Station			Person Appoved Canada Airwort							
	f Appr	oval or Rejection		Certificate or Des	signation No.	T	Signature of Authorized Individual							

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

_			
8.	Description	of Work Acc	complished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N510PD

s/n 0400E

McDonnell

Douglas 369E

Removed the Bendix/King KY-196 COM and KT-76A transponder

Installed into this aircraft:

Garmin GTX-330 Mode S transponder w/ TIS TSO-C112 at FS+49

Garmin SL-40 COM TSO-C37d, C38d, C128 at FS+48

Utilized supplied hardware from the manufacturer, standard aircraft hardware, MS27500/18-22(shielded wire) MS22759/16 (wire), and RG/U-400 Type IX Mil-C17 (coax)

Mounted the GTX-330 in the radio stack utilizing the existing mounting rails where the previous transponder was located. Connected transponder to the existing antenna. Interfaced to the existing encoder. Interfaced to the existing Garmin GNS-430's to display the traffic information system. Connected audio to the existing audio panel. Connected to the aircraft buss through a Klixon 7277-2-5amp circuit breaker and labeled "XPON".

Mounted the SL-40 in the center radio stack. Connected COM to the existing COM antenna. Interfaced to the existing audio panel "COM 2" position. Connected to the aircraft buss through a Klixon 7277-2-5amp circuit breaker and labeled "COM 2".

Checked transponder and encoder per FAR 91.217 and FAR 91.413.

Installation referenced to Garmin GNS-430 installation manual #190-00140-02 rev Q, GTX-330 #190-00207-02 rev K, SL-40 #560-09560-03 rev B, applicable paragraphs of FAA AC43.13-1B chapter 10, chapter 11 and chapter 12, AC43.13-2A 1-1 - 1-12, 2-21 - 2-25, 2-27, 3-36, 3-38, FAR 25.1301, FAR 25.1309a, b, d, AC 20-138a. This is a follow on installation to Garmin's SL-40 STC #SA00421SE, and GTX-330 STC #ST01125WI

Revised the equipment list, airframe log, and weight & balance.

<u>AIRW</u>	VORTHINESS LIMITATION	S: The Garmin GTX-330 pilot's gui	ide p/n 190-00207-00 rev A and the Garmin
GTX-330 FAA	Approved Flight Manual Sup	pplement doc #75398 dated	or subsequent revisions must
be in the aircra	ft when utilizing this system.		
Instructions for	r continued airworthiness are	attached to this 337 and are listed in	the corresponding installation manuals.
	**********	***** NOTHING FOLLOWS ***	
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		• •	

Additional Sheets Are Attatched

A.2 Continued Airworthiness

Other than for regulatory periodic functional checks, maintenance of the GTX 330 is "on condition" only. Refer to the GTX 330 Maintenance Manual (Garmin P/N 190-00207-05). Periodic maintenance of the GTX 330 is not required.

This section provides assistance to the installing agency in preparing Instructions for Continued Airworthiness (ICA) in response to Bulletin Number HBAW 98-18, "Checklist for Instructions for Continued Airworthiness for Major Alterations Approved Under the Field Approval Process", effective 10/7/98.

Aviation Authority approved installers are hereby granted permission to reference appropriate service instructions and excerpts from this Installation Manual to accomplish the Instructions for Continued Airworthiness. This permission does not construe suitability of the documents. It is the applicant's responsibility to determine the suitability of the documents for the ICA.

Following is a suggested ICA for a Garmin GTX 330 unit installation. Some of the checklist items do not apply, in which case they should be marked "N/A" (Not Applicable).

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS, GARMIN GTX 330

1. Introduction

[Aircraft that has been altered: Registration (N-) number, Make, Model and Serial Number]

Content, Scope,

Purpose and Arrangement: This document identifies the Instructions for Continued Airworthiness

for the modification of the above aircraft by installation of a Garmin

Applies to aircraft altered by installation of the Garmin GTX 330.

GTX 330.

Applicability:

Definitions/Abbreviations: None, N/A.

Precautions:

None, N/A.

Units of Measurement:

None, N/A.

Referenced Publications:

Garmin GTX 330 Installation Manual, P/N 190-00207-02

Garmin GTX 330 Maintenance Manual, P/N 190-00207-05

Garmin STC # ST01125WI.

Garmin GTX 330 Pilot's Guide, P/N 190-00207-00.

Distribution:

This document should be a permanent aircraft record.

2. Description of the Alteration

Installation of the Garmin GTX 330, with interface to Encoding Altimeter or Blind Encoder. Refer to Section 4 and Appendix C of this manual for interconnect information. Antenna installation, removal and replacement should be in accordance with applicable provisions of FAA Advisory Circulars AC 43.13-1B and AC 43.13-2A.

3. Control, Operation Information

Refer to the GTX 330 Pilot's Guide.

4. Servicing Information

N/A

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5. Maintenance Instructions

Maintenance of the GTX 330 is 'on condition' only. Periodic maintenance is not required. Refer to the GTX 330 Maintenance Manual.

6. Troubleshooting Information

Refer to the GTX 330 Maintenance Manual.

7. Removal and Replacement Information

Refer to Section 2 of this manual. If the unit is removed and reinstalled, a functional check of the equipment should be conducted in accordance with Section 5 of this manual.

8. Diagrams

Refer to Section 2, Section 4 and Appendices B and C of this manual.

9. Special Inspection Requirements

N/A

10. Application of Protective Treatments

N/A

11. Data: Relative to Structural Fasteners

Antenna installation, removal and replacement should be in accordance with applicable provisions of FAA Advisory Circulars AC 43.13-1B and AC 43.13-2A. Also, refer to Section 2 of this manual.

12. Special Tools

N/A

13. This Section is for Commuter Category Aircraft Only

- A. Electrical loads: Refer to Section 1.6.3 of this manual.
- B. Methods of balancing flight controls: N/A.
- C. Identification of primary and secondary structures: N/A.
- D. Special repair methods applicable to the airplane: Antenna installation, removal, and replacement should be in accordance with applicable provisions of FAA Advisory Circulars AC 43.13-1B and AC 43.13-2A.

14. Overhaul Period

No additional overhaul time limitations.

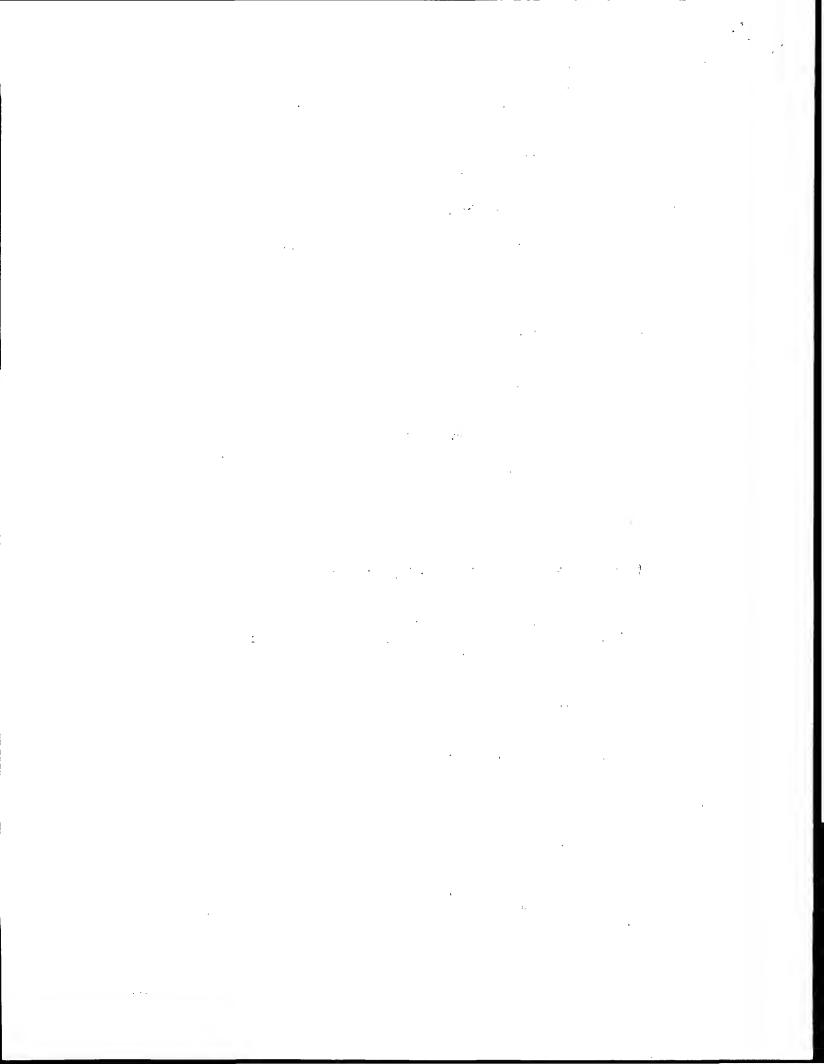
15. Airworthiness Limitation Section

N/A.

16. Revision

To revise this ICA, a letter must be submitted to the local FSDO with a copy of the revised FAA Form 337, and revised ICA. The FAA inspector accepts the change by signing Block 3 and including the following statement:

The attached revised/new instruction	is for Continue	ed Airworthiness (date) for the above
aircraft or component major alteration	have been ac	cepted by the FAA, super-	seding the Instructions
for Continued Airworthiness (date)."		



17. Assistance

Flight Standards Inspectors have the resources to respond to questions regarding the ICA.

18. Implementation and Record Keeping

For major alterations performed in accordance with FAA field approval policy, the owner/operator operating under Part 91 is responsible for ensuring that the ICA is made part of the applicable Section 91.409 inspection program for their aircraft. This is accomplished when a maintenance entry is made in the aircraft's maintenance record in accordance with Section 43.9. This entry records the major alteration and identifies the original ICA location (e.g., Block 8 of FAA Form 337, dated _____) along with a statement that the ICA is now part of the aircraft's inspection/maintenance requirements.

GTX 330 Airborne ATCRBS/Mode S Transponder Equipment

TYPE/MODEL/PART NO (Unit P/N)	TSO COMPLIANCE*	JTSO COMPLIANCE*	Diversity
010-00230-() (011-00455-00, -20)	C112 Class 2A	2C112a Level 2s	No
010-00293-() (011-00455-10, -30)	C112 Class 2A	2C112a Level 2s	Yes
010-00308-() Obsolete (011-00455-40,-50)	C112 Class 1A	2C112a Level 1s	Yes

* Notes: See Section 2.2 for part number, and complete TSO classification and description.

All versions of the 011-00455-() are class 2A except -(40), -(50) which are class 1A and are obsolete.

APPENDIX B - PERIODIC MAINTENANCE

EQUIPMENT CALIBRATION

The SL40 design requires very few adjustments or calibration to be made. In fact, there are **no** internal manual adjustments.

REFERENCE OSCILLATOR

The reference oscillator frequency should be checked approximately every 3 to 5 years to ensure the units transmit frequency is within allowable tolerance.

The oscillator frequency can be checked by connecting the transmitter output through an appropriate load to a calibrated frequency counter. The transmit frequency should be within 15ppm of the selected channel frequency. Contact the Garmin AT, Inc. factory for instructions on adjusting the frequency if required.

CLEANING THE FRONT PANEL

The front bezel, keypad, and display can be cleaned with a soft cotton cloth dampened with clean water. DO NOT use any chemical cleaning agents. Care should be taken to avoid scratching the surface of the display.

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U.S. Department of Transportation Federal Aviation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020

For FAA Use Only
Office Identification

TIA, WP-27

Administration										VII		
instructions an	d disposition of th	nis for	ntries. See FAR 43 rm. This report is r n (Section 901 Fed	equired	by	law (49 U.S.0	j. 1421	C 43.9-1 (d). Failure	r subseque to report c	ent rev an resi	ision thereof) ult in a civil pe	for enalty not to
1. Aircraft	Make MDHC				Model 369E							
	Serial No. 0400E						rk	,,,,,,				
	Name (As shor	wn on	registration certificate)				Addre	ss (As show	on registrat	on certif	ficate)	
2. Owner	City of Oakla Police Helico	nd					455 7	th. Street and Ca. 94				
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			requirements at conformity insp	nd is approved the section by a	ved i	mplies with the arrow the above described the Fin	an	aft, subject to ction 43.7 .K-FSDO				
			4. Unit k	lentifica	atio	on					5. T	ype
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POWERPLANT	r											
PROPELLER												
APPLIANCE	Type Manufacturer						7					
				6. Con	for	mity Statem	ent					
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Big Valley Avi				/		U.S. Certificate	d Mecha	anic		WG3	R944L	
7535 S. Lindb						Foreign Certificated Mechanic						
Stockton Ca.	95206				X Certificated Repair							
					Manufacturer					1		
hereto ha	ve been made in	acco	teration made to the rdance with the rec rue and correct to	quiremer	nts	of Part 43 of	the U.S	ve and de S. Federal	scribed on Aviation Re	the reve	verse or attac ons and that	hments the
Date 01/17/06				Sign	nati	ure of Authori	MA.	//	,		Pa	ul McKenzie
01/1//00			-	A	-14	or Return 10	m		2			U WICKEILIE
Pursua	nt to the authority	give	n persons specified ral Aviation Admini	below,	the	e unit identifie	d in ite		spected in	the m	anner prescr	ibed by
F	AA Fit. Standards		Manufacturer			Inspection		ization	Other (S			
J	AA Designee	x	Repair Station			Person Appov Canada Airwo						
Date of Appro	val or Rejection	•	Certificate or Design	ation No.				horized In	dividual			
0//17/0	16		WG3R944L			Van	Ma	WET I	2		Pa	ul McKenzie
FAA Form 33					,		1 9	(19)				

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

01/17/06 MDHC 369E N510PD S/N 0400E Actt 11140.5

AeroComputer LE 5000 Installation

The AeroComputer UltiChart LE 5000 series airborne command and control system provides situational awareness in airborne environments such as Airborne Law Enforcement, Serveillance, Search and Rescues; Fire Fighting or Electronic News Gathering. The LE 5000 is designed to be operated by the observer, supplying moving maps and directional information regarding current position, targets, and points of interest.

The system is not an FAA approved navigation tool and is not meant to be used for navigation by the pilot of the aircraft. Installed placard on monitor that reads: Moving Map System not to be used for navigational purposes.

The user input to define destinations, map choices, display data ect. is accomplished with the keyboard. The system also uses a touch screen and mouse enable for additional methods of used input.

Installed Aero Computer Airborne Systems Ultichart LE-5000 Map System hardware in accordance with LE-5000 Installation Manual Revision 1.9 dated August 8, 2003.

The CPU Module S/N 2257 is installed in the aircraft at station 91.9 on the left hand side of the rear cabin. A support shelf was fabricated from 2024T3 x .071 sheet aluminum, with folded ends for added strength to a dimention of 10.0" x 14.5" onto which four shock mounts are attached with screws P/N MS35207-263, washers AN960PD10L, and nuts MS21042L3. The CPU Module Mount is secured to the four shock mounts with four screws P/N MS35207-263. The CPU Module is secured to the mount plate with hardware supplied with the installation kit. This shelf is attached to the left foward seat bulkhead aft side with support brackets fabricated from 2024T3 x .071 sheet aluminum with foulded flanges to allow attachment to the seat bulkhead and the CPU shelf described above. The support brackets are riveted to the shelf with AN470AD4-5 revits and the assembly is attached to the seat bulkhead with eight screws MS35207-264, four per side and doublers under the nuts MS21042L3.

The system keyboard is attached to the bottom of the monitor support mount at station 46.0 that was previously approved on a FAA 337 field approval dated 09/05/1990 for the Flir Monitor installation. The keyboard is secured to the monitor support bracket using six screws P/N MS24693S32. With the keyboard fully extended it has ample clearance from the right cycle stick through full range of travel. The keyboard is the system default and the primary method of user input to the CPU.

The system GPS antenna is mounted in the foward cabin right hand side at station 36.0. The GPS antenna allows the GPS receiver in the CPU Module to capture the signals of the GPS satellites. The system uses an active powered GPS antenna and is designed to be independent of all aircraft instrumentation and as such cannot be shared or use any other antenna.

The system CPU Module is powered by aircraft 24 volt electrical system through a 3 amp circuit breaker switch P/N 7270-1-3 located on the right hand side of the center console. This switch is labled (Computer). The necessary electrical and data harness were fabricated in accordance with the LE 5000 Installation Manual and routed through the belly of the aircraft with the existing wire bundles in accordance with Standard Practices Manual AC43.13-1B. The wire harness is routed from the CPU Module to the monitor, and the keyboard located in the foward cabin.

This system installation was completed in accordance with LE 5000 Installation Manual and AC43.13-1B Change 1 Chapter 7 Aircraft Hardware, Chapter 10 Weight and Balance Data, Chapter 11 Electrical Load Section 1,3,4,5,6,7,8,9,10,11,12,15,16, and 17, Chapter 13 Human Factors, and AC43.13-2A Chapter 1 Structural Data, Chapter 2 Radio Installations, and Chapter 3 Antenna Installations.

New equipment list and weight and balance data computed and installed into flight manual.

The aircraft was test flown in accordance with FAR 91.207 (b) by an appropriately roted pilot, and the system was tested in flight to determine its ability to operate properly and to determine its interaction with other aircraft systems. The test flight revealed that there was no interference with other communication or navigation systems in the aircraft. An log book entry was completed to reflect this flight test.

X Additional Sheets Are Attatched

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) ,
01/17/06 MDHC 369E S/N 0400E N510PD Actt: 11140.5
Instructions for Continued Airworthiness
1) Introduction: As described briefly above. Additional information is in the LE 5000 Installation Manual.
2) Description: As described briefly above, Additional information is in the LE 5000 Installation Manual.
3) Control: The system is powered and protected by a 3 amp circuit breaker and is controlled and operated by the system keyboard and touch screen.
4) Servicing information: The LE 5000 system components must be serviced and repaired by AeroComputer Inc. 716 N. Ventura Rd. # 342, Oxnard Ca. 93030 P/H# 805-985-3390.
5) Maintenance Instructions: General condition and security of the system components and electrical harness are to be inspected at each 100 hour inspection. System maintenance and data updated are to be conducted by the manufactor.
6) Troubleshooting information: Reference the LE 5000 Installation Manual and the Users Manual for troubleshooting information.
7) Removal and installation information: Reference LE 5000 Installation Manual.
8) Diagrams: Reference LE 5000 Installation Manual.
9) Special inspection requirements: None.
10) Application of protective treatment: None.
11) Data: Reference LE 5000 Installation Manual and system installation description above.
12) List of special tools. None.
13) For commuter category aircraft: Not applicable.
14) Recommended overhaul penods: Not applicable.
15) Airworthiness Limitation Section: Not applicable.
16) Revision: To revise this ICA submit a letter to the local FSDO with a copy of the revised FAA Form 337 and revised ICA Once the revision has been accepted a maintenance record entry will be made, identifying the revision, its location, and date of the revised Form 337.

Additional Sheets Are Attatched

********************* NOTHING FOLLOWS *

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US Department of Transportation

Federal Aviation Administration

Date of Approval or Rejection

12/1/03

Certificate or

Designation No.

NG3R709L

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only
Office Identification

WP-25

112

INSTRUCTIONS: Print or type all entries. See FAR 43.9 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958). Make Model McDonnell Douglas Helicopter 369E 1. Aircraft Serial No. Nationality and Registration Mark 400E N510PD Name (As shown on registration certificate) Address (As shown on registration certificate) 2. Owner City of Oakland Police Department 455 7th St Helicopter Av Unit Oakland, CA 94607 3. For FAA Use Only The data identified herein complies with the applicable airworthiness requirements and is approved only for the above described aircraft, subject to conformity inspection by a person authorized in 14 CFR part 43 section 43.7 11/28/2003 Date Larry L. Złoczewski, Principal Ayjonics Inspecto 4. Unit Identification 5. Type Unit Make Model Serial No. Repair Alteration AIRFRAME ----- (As described in Item 1 above) -----X POWERPLANT **PROPELLER** Type APPLIANCE Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency C. Certificate No. U.S. Certified Mechanic NG3R709L THE GYRO HOUSE Foreign Certified Mechanic Inst. 1, 2, 3, 4 2389 RICKENBACKER WAY Certificated Repair Station Radio, 1, 2, 3, Ltd. **AUBURN, CA 95602** Limited Air Frame Manufacturer D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge Date Signature of Authorized Individual Albert Shogren 11/29/03 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the ▼ APPROVED Administrator of the Federal Aviation Administration and is ☐ REJECTED Other (Specify) FAA Flt. Standards Manufacturer Inspection Authorization Inspector BY Person Approved by Transport Repair Station FAA Designee Canada Airworthiness Group

Signature of Authorized Individual

FAA Form 337 NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

R	Description	of Work	Accomplished

(If more space is required, attach additional sheets, Identify with aircraft nationality and registration mark and date work accomplished.)

MD Helicopter 369E

N510PD

SN: 400E

- 1. Installed LoJack Vehicle Tracking System as follows:
- 2. This Installation was previously approved on Bell Helicopter Model 206B III, Registration N211LA, SN: 3956, on FAA Form 337 dated 7/15/2002. The following components were listed in that installation:

The Receiver/ Computer Module, LoJack P/N: PTC2A-E was installed under Copilot's seat on existing access panel (fs 62) using standard AN hardware.

The system was coupled to existing 28VDC to 12VDC power converter with a 2 amp circuit breaker switch mounted at bottom of avionics stack.

The Display Controller, pn: PTC2-D, was mounted on the top of aircraft instrument panel glare shield (fs 33.0) using standard AN hardware.

Installed fabricated antenna plate to bottom of aircraft fuselage (fs 85.5). Antenna plate was fabricated IAW LoJack Installation Drawing #LJ010. Four CI 177 Antennas were mounted on the antenna plate and the Assembly was installed IAW LoJack Installation drawing # LJ005. The CI 177 Antennas are TSO'd C37B and C38B.

LoJack system is a non-flight critical item and is completely removed from aircraft power using control switch. All other systems in aircraft are not affected.

The Aircraft Weight and Balance and Equipment List have been updated IAW AC 43.13-1B, and a copy has been added to the aircraft POH.

An electrical load analysis was performed IAW AC 43.13-2A and found to be within 80% of maximum continuous alternator outputs.

Instructions for continued airworthiness require a visual inspection and operational test annually IAW manufacturers instructions and FAR 43 appendix D.
 END
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U.S. Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020

For FAA Use Only

Office Identification T.O. G. WA 27

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for

excee	d \$1,00		violati	on (Section 901	Federal A	via	ion Act of 19						enalty not t	
1. AI	rcraft	Make MD		Model 369E										
		Serial No. 0400	E					y and Re I510PD	gistration Ma	ırik				
2. 0	wner			n registration certific Police Dept.	Address (As shown on registration certif 456 7th Street Oakland Ca. 94607						ficate)			
		· · · · · · · · · · · · · · · · · · ·			3. 1	For	FAA Use On	ly						
				4. Uni	it identifi	cati	on					5. T	уре	
	Unit . Make						Model		s	erial No.		Repair	Alteration	
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Date 12-06-	01		• • • • • • • • • • • • • • • • • • • •			A)	ure of Author	us	dual သိ			Pa	ul McKenzie	
				7	7. Appro	val 1	or Keturn J	Service						
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BY	F.	AA Fit. Standards spector		Manufacturer	- 1		Inspection Authorization Other (Specify)							
Doto -		AA Designee	×	Repair Station			Canada Airw	orthiness Gro	oup					
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12-06-01 WG3R944L							Paul McKenzie							

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

Companie with air previous atterations to assure continued contornity with the applicable an worthness requirements.
8. Description of Work Accompilshed (If more space is required, attach additional sheats. Identify with aircraft nationality and registration mark and date work completed.
12-06-01
N510PD Actt. 9497.7
Engine Inlet Filter Installation
Installed engine inlet air filtration system P/N 13691N-1 in accordance with INTEC Master Drawing List No. 1369-1001, Revision A dated Feburary 2, 2001 and STC # SR00877SE.
Adjusted equipment list and weight and balance records to reflect this installation.

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RECI	EIVED					Form Ap				
		OMB No.2120-0020 For FAA Use Only								
of Transportation.	L	Office Identification								
Federal Aviation Administration	2-1	1 Wh	200							
	: Print or type all entries. See FAR 43.	O EAD A	3 /	Innendiy B. and A	C 43 Q-1 (or sub					
instructions and o	disposition of this form. This report is report each such violation (Section 901 Federal)	quired b	y la	w (49 U.S.C. 142	1). Failure to rep	port can res	sult in a civil p	enalty not to		
1. Aircraft	Make MDHC			Mode	1 369E					
	Serial No. 0400E			Natio	nality and Registrati N510PD	on Mark				
2. Owner	Name (As shown on registration certificate) City of Oakland Police Dept.		Address (As shown on registration certificate) 456 7th Street Oakland Ca. 94607							
	1	3. For	FΑ	A Use Only						
	4. Unit ide	ntificati	on				5. 1	Гуре		
Unit	Make .		М	odel	Serial N	lo.	Repair	Alteration		
AIRFRAME	(As de	scribed ir	ı ite	em 1 above)			x			
POWERPLANT				11				La C		
PROPELLER	·									
APPLIANCE	Туре .									
17	Manufacturer			N.						
0.1		. Confo	rm	Ity Statement	-		.1	1		
A. Agency's Na	me and Address			. Kind of Agency		C. 0	Certificate No			
Big Valley Aviation	on Inc:		U	S. Certificated Mech	anic	WG	3R944L			
7535 S. Lindberg	h St.		F	oreign Certificated Me	chanic		-			
Stockton Ca. 952	:03	X	C	ertificated Repair Sta	lion					
			М	anufacturer						

hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. Date Signature of Authorized Individual 12-06-01 Paul McKenzie 7. Approval for Return To Service Pursuant to the authority given persons specified below, the unit identified In item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED FAA Fit. Standards Other (Specify) Manufacturer Inspection Authorization BY Inspector Person Appoved by Transport X FAA Designee Repair Station Canada Airworthiness Group Date of Approval or Rejection Certificate or Designation No. Signature of Authorized Individual 12-06-01 WG3R944L Paul McKenzie

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments

FAA Form 337 (12-88)

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.							
8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft netionality and registration mark and date work completed.)							
12-06-01 N510PD Actt. 9497.7							
Main Rotor Blade Installation.							
Installed replacement Main Rotor Blades P/N 500P2100-101 revision E. In accordance with Helicopter Technology Company (HTC) Installation and Maintenance Instructions HTCM-001, Rev. N/C dated 4-02-96 and Master Drawing List MDL-2100-01, Rev. B, dated 5-31-96 or later FAA approved revision and STC # SR09074RC.							
Weight and Balance Not Affected.							

U.S. Department of Transportation
Federal Aviation

FAA Form 337 (12-88)

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020

For FAA Use Only
Office Identification

7.29. WP-27

instruction	s and disposition of	this for	ntries. See FAR 43.9, FAR rm. This report is required on (Section 901 Federal Avi	by lav	w (49 U.S.C. 1421).	43.9-1 (or Failure to	subsequent re o report can res	vision the	nereof) for civil penalty n	ot to		
1. Aircra	oft Make	Make MDHC				Model 369E						
	Serial No.		0400E		Nationality and Registration Mark N510PD							
2. Owne	Name (As she City of Oa Police He		Address (As shown on registration certificate) 455 7TH Street Oakland Ca. 94607									
			´3.	For	FAA Use Only							
			4. Unit Identif	inatle					5. T	imo.		
	<u> </u>	• • •		Ican			Carial Na			1		
Unit		Ma	ike		Model		Serial No.		Repair	Alteration		
AIRFRAME		(As descri								x		
POWERPL	ANT					B						
, PROPELLI	ER			•								
APPLIANC	Type E Manufacturer											
			6. (Confo	rmity Statement	!.			<u>l</u>			
A. Agend	y's Name and Addre	ess			B. Kind of Agenc	у		C. C	ertificate No.			
Big Valle	y Aviation Inc.				U.S. Certificated Mechanic				WG3R944L			
	indbergh St. Ca. 95206				Foreign Certificated M							
Olocaton	Oa. 30200			X	Certificated Repair St	ation						
					Manufacturer							
have	been made in accord	dance '	eration made to the unit(s) with the requirements of Pa ect to the best of my knowle	art 43	of the U.S. Federal	and desc Aviation I	cribed on the re Regulations an	verse o	r attachments ne information	hereto		
Date 05-02-00			P	aul N		ul	Muse	4	7			
11 4.1					for Return 70 Sen							
Pur Adr	suant to the authority ministrator of the Fed	y giver leral A	n persons specified below, wation Administration and	the u is	nit identified in item X APPROVE	4 was ins	REJECT	TED	prescribed by	the		
ву	FAA Fit. Standards Inspector		Manufacturer		Inspection Author		Other (S	pecify)				
	FAA Designee	Х	Repair Station		Person Appoved by Canada Airworthine	Transport ss Group						
Date of Ap	proval or Rejection		Certificate or Designation	No.	Signature of Author	rized Indi	victual //		17			
05-02-00 WG3R944L					Paul McKenzle							

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished						
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)						
(······						
•						
05-02-00						
ACTT. 8769.2						
N510PD						
Pulse Light Installation						
Puise Light Installation						
Installed Precise Flight Pulselite Control Unit Model 1220/2410-2 in the Landing / Taxi Recognition Visual Control Light system in						
accordance with the Pulselite Installation Manual #PPRI-2000 Revision 13 and STC# SH3319NM. Weight and Balance not affected.						
Mounted control unit at station 62.5 under left hand floor panel.						

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The deliteration of the Atlantia d						
Additional Sheets Are Attached						
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RECEIVED OAKLAND F.S.D.O.

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U.S. Departm	ent
of Transports	tion
Federal Avia	tion

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

JUL 0 9 100MB No.2120-0020

For FAA Use Only

Office Identification

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Adminis										TUI DE	M-1880			
and	l disposition	n of this form. T	his re	entries. See FAR 4 eport is required by I Federal Aviation A	law (49 l	J.S.C			(or subsequent revi	sion thereo	f) for instructi	ons		
		Make		······································				Model						
1. 4	Aircraft	Serial No.	IDH	<u> </u>				Nationalite -	369E nd Registration Mark		.			
•••			400E					ічанопаіку а	na Registration Mark N510PD					
		1		registration certifcate))				shown on registration	certificate)				
2.	Owner	City of Oakla Police Helica		r I Init				455 7TH :	Street Ca. 94607					
r once rencopter one														
		······			3	3. F	or FAA Use C	nly			ı			
						. U	nit Identificat	ion			5. Туре			
	Unit	:	Ма	ke . ;	·		Model		Serial No	D.	Repair	Alteration		
AIRF	RAME				(As desc	cribe	d in item 1 abov	/e) ~~~~~		,	•			
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POWE	ERPLANT													
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PROP	ELLER .													
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APPLIANCE		Manufacturer						•						
					6.	Cor	nformity State	ement	•		<u> </u>	1		
		lame and Add	dres	s .			Kind of Agend	у		}	icate No.			
	/alley Avi										/G3R944L			
	South Li kton Ca. !	ndbergh St. 95206		•		Foreign Certificated Mechanic X Certificated Repair Station								
							Manufacturer	dir Otation		1				
				eration made to the										
				with the requirement ect to the best of my			of the U.S. Feder	al Aviation I	Regulations and tha	t the inform	ation			
Date		,		:			nature of Authoriz	ed Individu	al O					
	6-5-	98			•	P	winch	mz } ϵ	· Kaul	Mari	les i	,		
							al for Return			110	0			
Purs Adn	uant to the ninistrator	authority given of the Federal	perso Aviati	ons specified below ion Administration	, the unit and is	lden	tified in item 4 wa		d in the manner pred REJECTED	scribed by t	he	• •		
BY	1	Fit. Standards ector		Manufacturer		Insp	ection Authoriza	tion	Other (Specify)				
	FAA	Designee	х	Repair Station			son Approved by ada Airworthines	-				,		
Date		l or Rejection		Certificate or Designation No.		<u> </u>	nature of Authoriz			1n		*		
6-5-98 Designation No. WG3R944L Poul makene in the first in the fir														

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and regis	tration mark and date work completed.)
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	A STANDARD COMMENT
6-5-98 N510PD ACTT 7713.3	The state of the s
	,
Pc Safety Valve Kit	
the second secon	the state of the s
Installed Aeronautical Accessories Inc. Compressor Discharge F Allison 250-C20 Engine in accordance with Report # AA-92005 Revision	Pressure (Pc) Valve Kit P/N 250-951 Series for n Y dated July 9, 1997 and STC # SE5511NM.
Continued Airworthness Instructions provided in the Maintenance AA-92005 Revision Y dated July 9, 1997.	and Inspection Instructions Section 3 of Report
Weight and Balance not Affected. Adjusted equipment List to ref	lect this installation.
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☐ Additional Sheets Are A	ttached

OAKLAND F.S.D.O.

APR 3 0 1000 MAJOR REPAIR AND ALTERATION (Ainframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020

For FAA Use Only

Office Identification

Federal A Administ	ration							·	.1.9.	WAZ	7			
				entries. See FAR 43.9				subsequent revi	sion thereof) for Instructi	ons			
				port is required by law		J.S.C.1421). Failure	to report can r	esult in a civil per	nalty not to	exceed \$1,00	00			
for e	each suc	h violation (Sectio	n 901	Federal Aviation Act	1958)									
		Make				Model								
1. /	Aircraft	MDHC					369E							
/		Serial No. 0400E					Nationality and Registration Mark N510PD							
			n on r	egistration certificate)				own on registration	certificate)					
		City Of C		-			-	H. Street	,					
2. (Owner			pter Unit		-	Oakland	d CA. 94607						
					3	. For FAA Use C	Only							
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_					4	. Unit Identifica	tion			5. Type				
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,	Unit		Mal	ke		Model		Serial No	·,	Repair	Alteration			
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							1				-			
POWERPLANT														
									-					
PROPELLER														
														
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APPLI	IANCE	Manufacturer												
		Ivialiulaciulei		1										
_	٠				6.	Conformity State	ement			<u> </u>				
A. A	gency's	Name and Ad	dress	3		B. Kind of Agen			C. Certi	icate No.				
		y Aviation Inc.			_	U.S. Certificate		WG3R944L						
		Lindbergh ST.				Foreign Certific								
S	tockton	CA. 95206				X Certificated Re	ł							
<u> </u>				ration made to the unl	4/=\ 1d	Manufacturer	yo and describ	and on the reverse	or attachm	ents herete				
D. 1	ceruiy ti	iat the repair and/ n made in accorda	or alle	vith the requirements	ı(s) lui of Pari	43 of the U.S. Fede	ral Aviation Re	egulations and tha	t the inform	ation				
f	urnished	herein is true and	i corre	ect to the best of my ki	nowied	ige.		G aranette and an						
Date						Signature of Authori	zed Individual							
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<u> </u>	4-22-98) 				Sact 1111	Who i	<u>., 7.</u>	AU[//	ckenz	250			
				7.	App	oval for Return	Service			•				
Purs Adn	uant to t ninistrate	he authority given or of the Federal.	perso Aviati	ons specified below, the	e unit dis	X APPROV	as inspected in	n the manner pres	scnbed by t	ne				
	F/	A Fit. Standards	Г		Т			Other (Specify)	·····				
ву		spector		Manufacturer		Inspection Authoriza	tion							
 	F	AA Designee	Х	Repair Station		Person Approved by Transport					•			
						Canada Airworthine		L,						
Date	of Appro	val or Rejection		Certificate or Designation No.		Signature of Authori	zed Individual							
4	-22-98			WG3R944L		Kan Ulla	iki's	Pavi	mck	anzin				

8. Description of Work Accomplished of more space is required, attach additional sheets. Identity with electral nationality and registration mark and date work completed.) 4-22-98 N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc. rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance will Tech-Tool Plastics, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 5719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	3. Description of Work	Accomplished		
4-22-98 N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc. rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-4505-42	If more space is required, atta	ch additional sheets. Identify with aircraft nation	tality and registration mark and date work	K completed.)
4-22-98 N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc. rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-450				,
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4-22-98 N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc. rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-4505-42 and-450	•			
N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc.rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	Acquisition made in the angle of the specific sp	or, nan, unmadate un hij katini orini innin.		The second secon
N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc.rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	.			,
N510PD ACTT. 7613.4 Installed Tech-Tool Plastics Inc.rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505-42 and-43 inaccordance windows P/N 369-4505, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	4-22-98	•		
Installed Tech-Tool Plastics Inc.rubber mounted crew windows P/N 369-4505-42 and-43 inaccordance with Tech-Tool Plastics, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	N510PD	*****		
Tech-Tool Plastics, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	AC11. /613.4		/	
Tech-Tool Plastics, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	-			
Tech-Tool Plastics, Inc. Drawing 369-4505, Rev. B, dated 8-23-94, and STC# 8719SW. Weight and Balance change is negligible. Weight and Balance is not affected. Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	Installed Tech To	nol Plactice Inc rubber mounted crew	windows P/N 369-4505-42 and	-43 inaccordance with
Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	Tech-Tool Plastics, Inc.	. Drawing 369-4505, Rev. B, dated &	-23-94, and STC# 8719SW.	, massiannes min
Installed Rapid Removal Door Hinge System inaccordance with Certified Technology Installation Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	Weight and Bala	nce change is negligible. Weight and	Balance is not affected.	
Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	_	·		•
Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.				
Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.				
Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.				•
Instructions CT9602-1 Revision A, dated July 24, 1997 and STC# SR00409SE. Adjusted Weight and Balance data to reflect this installation. Revised Equipment list.	Installed Panid F	Removal Door Hinge System inaccor	dance with Certified Technology	√ Installation
Revised Equipment list.	Instructions CT9602-1	Revision A, dated July 24, 1997 and	STC# SR00409SE.	,
Revised Equipment list.			•	
	Adjusted Weight	and Balance data to reflect this insta	allation.	
**************************************	Revised Equipm	ent list.		

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☐ Additional Sheets Are Attached

OAKLAND FS.D.O.

Form Approved OMB No. 2120-0020

US Department of Transportation Federal Aviation Administration WAJOR REPAIR AND ALTERATION For FAA Use Of Transportation Federal Aviation Administration For FAA Use Of Transportation WP17 WA													
and disposition	of this form. Th	is for	entries. See FAR 43. m is required by law (Federal Aviation Act	49 U.S	S.C. 1	Appendix B, and 421). Failure to	AC 43.9-1 report can i	(or subseque result in a civi	ent revi	islon thereof) Ity not to exce	for instructi ed \$1,000	ons	
	Maka		nell Douglas		¥/		Model	369E					
1. Aircraft	Serial No.	RIA	NO. 0400E				Nationality	and Registra N510PD	ration Mark				
2. Owner			n registration certifica Oakland Police D	•	Address (As shown on registration certificate) 456 7th Street Helicopter Avn Oakland, CA 94607								
					3. For	FAA Use Only	.			· · · · · · · · · · · · · · · · · · ·			
			conto conto	ieb iden rements a rmity ins かートン DAI	end is a pection 09-	parein complies with opproved for the above by a person authorized of the Signature	the applicable described air din FAR 43, Se	creft, subject to ction 43.7	4				
					4. Uni	t Identification					5. Type		
Unit		Ma	ke		Model					l No.	Repair	Alteration	
AIRFRAME	~~	~~	*****	(As de	escrit	oed in item 1 a	bove) 🛹		***	·**		х	
POWERPLANT													
PROPELLER											·		
APPLIANCE	Туре												
	Manufacturer										<u> </u>		
A. Agency's Na	me and Addres	<u> </u>		6. (rmity Statement ind of Agency				C. Certificat	e No.		
Bay Avionic Bay Avionic Oakland Int' Oaklan, CA	s, Inc s, Inc I Airport				U.S. Certificated Mechanic Foreign Certificated Mechanic X Certificated Repair Station Manufacturer					WK3R948L Inst.Class 1,3,4 Radio Class 1,2,3 Limited A/F			
have been n	nade in accorda	nce v	eration made to the ur with the requirements ect to the best of my k	of Pari	t 43 o	ed in item 4 abov f the U.S. Feden	ve and desc al Aviation I	cribed on the Regulations a	revers	e or attachme It the informat	nts hereto ion		
Date 5/23	3/97				Sign	ature of Authoriz Bay Avionics,	62	Postere F	W	teore	}		
-	<i>/- / /</i>					r Return to Serv		7					
Pursuant to the Administrator o	authority given f the Federal Av	personation	ons specified below, to Administration and I	he unli s	t ident	ified in item 4 wa	as inspecte ED	d in the mann	ner pre ED	escribed by the			
FAA Insp	Fit. Standards Manufacturer				Inspection Authorization Other (Specify)								
1 1	Designee X Repair Station				Person Approved by Transport Canadian Alrworthiness Group								
Date of Approva	or Rejection (12-88)		Certificate or Designation No. WK3R948L		Sign	Hoslie	zed Individu	lal Central	() 3e))			

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

The existing II Morrow Apollo 618C Loran Receiver and Loran Antenna were removed from this aircraft.

A II Morrow Model 360 GPS Receiver and A-33 GPS Antenna were installed in this aircraft. The 360 GPS Receiver was installed in the instrument panel in place of the removed Loran Receiver. The A-33 GPS Antenna was mounted in place of the removed Loran Antenna using the existing doubler. The 360 GPS Receiver was installed as a stand alone unit with no navigation interface.

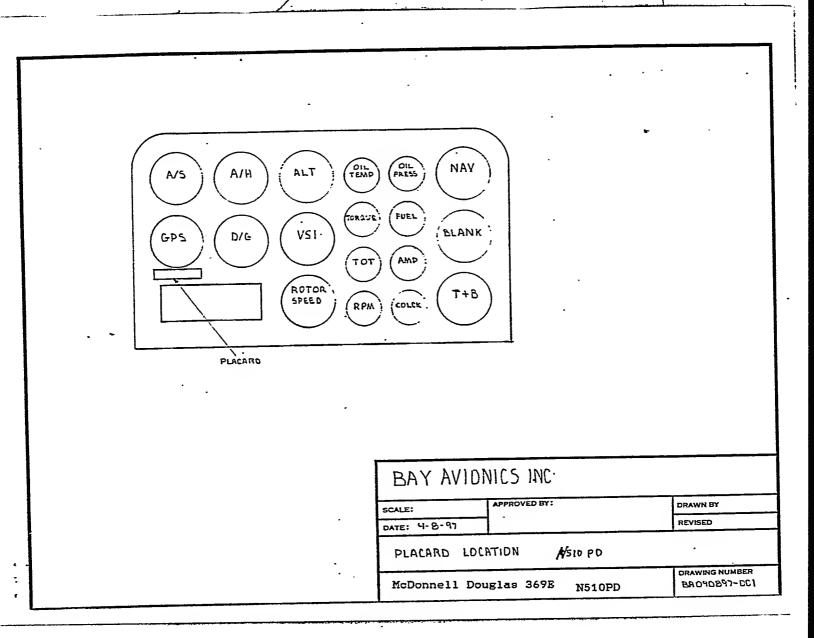
This installation was conducted in accordance with II Morrow Installation Guide No. 560-0124-00, rev. A, dated May 1995, and within the guidelines set forth by FAA A43.13-2A, Chapter 2 & 3 and AC 20-138, sections 7.b and 7.c (2).

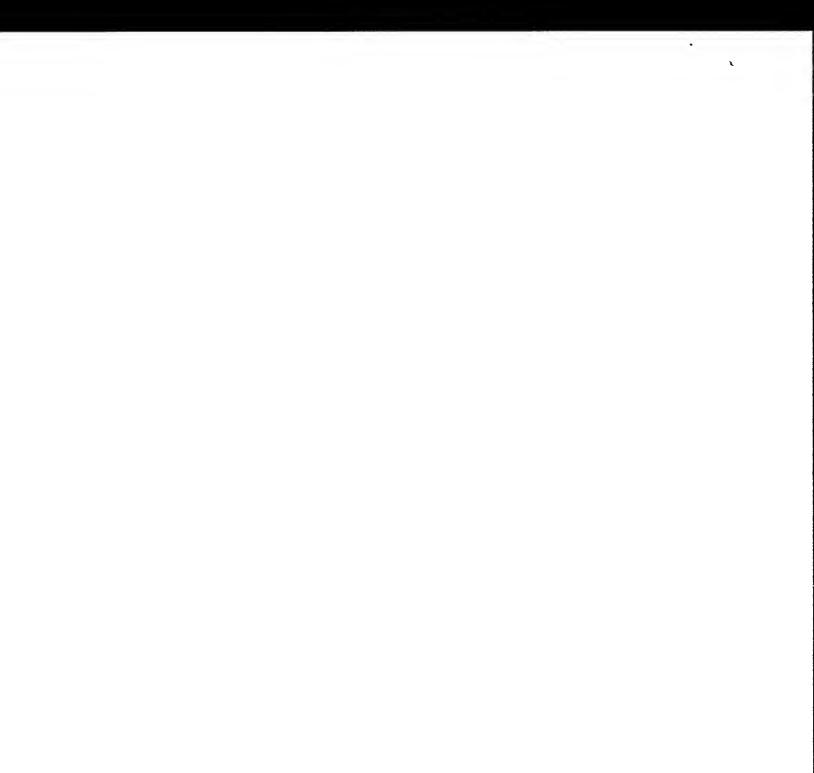
The II Morrow Apollo 360 GPS Receiver has been previously installed in a Partenavia P-68 series aircraft and approved for VFR use per Supplemental Type Certificate No. SA 00146SE.

The aircraft instrument panel was placarded "GPS Limited to VFR Use Only." This placard was installed below the II Morrow 360 GPS unit. See Bay Avionics Drawing BA040897-001 for installation details.

The pilot was provided a copy of the II Morrow Apollo 360 GPS Receiver User Guide, P/N 560-0123-00 dated October 1994.

The aircraft weight and balance and equipment list were amended to reflect this change.







MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification

WE-27

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000

for e	for each such violation (Section 901 Federal Aviation Act of 1958).													
		Make					Model							
1. Airc	raft			MONC				3690	<u> </u>					
i. Air	Jail	Serial No.				<u></u>	l .	ty and Registration	_					
				406 <i>E</i> on registration cer				N510						
		Name (As sho	own (on registration cer	rtificate,)		(As shown on regis						
2. O w	ner			if orkl			20	5 <i>5 71</i> 11 8	TREET					
		POLICE	<u> </u>	HELICOP	アディ	_ ULDIT	. UNIT ONCLAND, CA, 9460							
			ž			3. For FAA Use O	nly	•						
The data identified herein complies with the applicable airworthiness requirements and is approved for the above described aircraft, subject to conformity inspection by a person authorized in FAR-43, Section 43.7 SOUND TO MAKE SIGNATURE OAK-FSOO														
			7000	as F		4. Unit Identificati	on			5. Type				
Unit Make						Model		Serial No.	·	Repair	Alteration			
AIRFR	AME		••••	······································	As desc	ribed in Item 1 abo	ve)				×			
POWE	RPLANT						š							
PROPELLER														
		Туре												
APPLIA	ANCE	Manufturi												
		Manufacturer			İ									
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Α. Δα	encv's N	lame and Addre	ess			B. Kind of Agenc		-	C. Certi	ficate No.				
				VACTION 1		U.S. Certificate		;	3. 33.11					
91	ラグ	iene Li	מ ממי	NIATIONS CT RO		Foreign Certific	cated Mecha	anic	1	/	,			
					,	X Certificated Re	pair Station		WE30944L					
ON	レビネド	WA , C	-1	, 9462		Manufacturer								
ı	nave bee	n made in acco	ordar		rements	_	S. Federal	Aviation Regulation						
Date						Signature of Auth	orized Indi	ividual						
	\$	5 MAY	77			tout-		ngerfor	الم	%				
					7. Ap	proval for Return T	o Service							
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED IREJECTED										ed by the				
BY		A Fit. Standards pector		Manufacturer		Inspection Authoriz	ation	Other (Specify)						
		Designee	×	Repair Station		Person Approved by Transport Canada Airworthiness Group								
Date of Approval or Rejection Certificate or Designation No. Signature of Authorized Individual								. /		-				
	8.111	1497		NG3094	44	Foul A	lunge	Worl						
FAA F	orm 337	(12-88)			C									

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8.	Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
	INSTALLED PRECISION AVIATION INC. VERTICLE CARD.
	COMPASS HOBEL NUMBER PAIL-700 IN ACCURDANCE
	WITH PAIL INSTALLATION BULLETIN DATEL 23 RPRIL 90.
	ALL WORK PERFORMED IN ACCORDANCE WITH AC 43.13-
	IA, CHANGE TRIPLES CHAPTER 2. SECTION 3. PRIGE SI,
	PARA. 99(c)(d)(1) AND FIGURE 12.29, PAGE 65. WEIGHT
	GUS BALANCE NOT AFFECTED. WEIGHT OF OLD UNIT IS 14
	DZ'S , WEIGHT OF NEW WIT IS II OZ'S MOUNTING IS
	AT FUSELAGE STATION SQLS (MOTORT LOCATION), LIGHTING
	SYSTEM WIRING WAS NOT DETERED. LEDUIPHENT LIST WAS
	ip-dated.
	NOTAING FOILOWS



US Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification
W-27

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

for each such violation (Section 901 Federal Aviation Act of 1958).													
	Make		*1 k . ! r		Model 369E								
1. Aircraft	Serial No.		MDHC_			Nationalii	ty and Registration	Mark					
	Seliai No.		4000				N516P						
•	Name (As sho	own	on registration cer	tificate)	 	Address (As shown on regi	stration c	ertificate)				
2. Owner	City	0	F OAKLAN	۵۱		455 7TH STREET							
z. Owner	POLICE	7	XEPT. HEL	(COP)	ER UNIT	OAKLAND, CA, 94607							
	_l			Dnly									
					4. Unit Identifica	tion			5. Type				
Unit		Ма	ke		Model		Serial No		Repair	Alteration			
AIRFRAME	•••	••••		As desci	ribed in Item 1 ab	ove) •••••				×			
POWERPLANT													
PROPELLER													
APPLIANCE	Type Manufacture	r											
		-		6.	Conformity Stat	ement			1	<u> </u>			
A. Agency's I	Name and Addr	ess			B. Kind of Agen			C. Certi	ificate No.				
Rig Valley A	viation, Inc.				U.S. Certifica	ted Mechanic							
EAA Bonair	Station No. WG3	D944	L		Foreign Certi	****			Valley Aviation, Inc. A Repair Station No. WG3D944L				
9625 5	PRNART	RD	, CAKLAN.	s, ca	✓ Certificated F	•		FAA NE	pail Sialoit Inc	, 11CDDJ-1-E			
D. I certify have be	that the repair a en made in acco	nd/o ordar	r alteration made t	o the uni	it(s) identified in it of Part 43 of the I	em 4 above a	and described on t Aviation Regulation	he reverse ons and ti	e or attachme hat the inform	nts hereto nation			
Date					Signature of Aut	horized Indi	ividual ,						
18	SEPT S	36			1.4	unçe	upu/@)					
					roval for Return	77	<i>y</i>						
Pursuant to the authority given persons specified below, the unit identified if item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED Contact Co										ed by the			
	A Flt. Standards pector				Inspection Author	ization							
FA	A Designee	<u> </u>	Repair Station		Person Approved Canada Airworthi	néss Group							
	Date of Approval or Rejection Certificate or Signature of Authorized Individual Big Valley Aviation No. IESEPT 94 FAA Repair Station No. WG3D944.												
18 SEP	176	FAA	nepair Station No.	.03034	ten	-7 Yu	neufore	<u> </u>					

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
INSTALLED TECH-TOOL PLASTICS, INC.
RUBBER MOUNTED PROSENGER WINDOW ITAK
TECH TOOL TRASTICS, INC. DRAWING 369-3506
REV. 'E DAVED 11-09-93 PER STC SUBG395W
EQUIPMENT LAST UP-BATES. WEIGHT AND BALANCE.
NOT AFFIECTED.
NOTHING FOLLOWS
·
,

United States of America

Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

Number SH8639SW

This certificate, issued to Tech-Tool Plastics, Inc. 7800 Skyline Park Dr. Fort Worth, TX 76108

corlifus that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the sirus thiness requirements of Part 6 of the Civil Air Regulations.

Original Groduct - Type Certificate Number: H3WE

Make: McDonnell Douglas Helicopter Company Mudel: 369A, D, E, F, FF, H, HM, HS, HE, and 500N

Description of Type Design Change:

Installation of an aft door window in accordance with Tech-Tool Plastics, Inc., drawing 369-3506, Rev. E, dated 11-09-93, or later FAA approved revision.

Limitations and Conditions:
Compatibility of this modification with previously installed equipment must be determined by installer.

This cortificate and the supporting date which is the basis for approval shall remain in effect until surrendoved, suspended, rowked, or a termination date is otherwise established by the Administrator of the Gedoral Aviation Administration.

Dale of application: May 8, 1992

Glabofissiance: May 13, 1993

Sule reissued:

Jule unrended 12/21/93 Rev.

By direction of the Stageinistrator

Larry M. Kelly, Manager

Rotorcraft Certification of

Any pletation of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding & cars, or both.

This certificate may be transferred in accordance with FAR 21.47

FAX FOLK 8110-2(10-68)

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US Departr of Transpor	

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification INSTRUCTIONS: Print or type ell entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to record one could be added to the country of the

1. Aircraft	h violation (Section 901 Federal Aviation Make		Model 367£								
I. Aureran	Serial No.		Nationality and Registr	ation Mark -	n Mark						
and the second	Name (As frawn on registration certification certification)	instal	Address (As about as a said to be a said to be								
2. Owner	The gray on registration certific	ate) Address (As shown on registration certificate)									
2 Owner	O.P.D., DAVIAGO T	over Der	Oakland Air	nort		;					
\$172.5		3. For FAA Use O	aly	N. N							
				· 6	Ž						
Project Contract		-4. Unit identificati	on in the second		5. Type	·					
Unit	Make	Model	Sérial	No.	Repair	Alteratio					
AIRFRAME	(AB	described in Item 1 about	(6)	••	10. Zeg	\$ <u>\</u>					
POWERPLANT					0.	ECEN AND E					
PROPELLER				0.07	≨ W	S D.C					
APPLIANCE	Type Manufacturer				- 13						
with the second	i sera grainara are la la casa la c	6. Conformity States	nest 2 to 1 terrent	30		<u> </u>					
A	ame and Address	B. Kind of Agency	range anger i light dia ag i i sig a dilit i	C. Certifi	cate No.						
	ty Aviation	U.S. Certificated		2							
P.ö.B.x	HIIZ	Y Certificated Reg	ated Mechanic		Valley Aviation, Inc.						
OALAN	Co. 94614	Manufacturer :-		FAA Rep	air Station No	W03D9441					
	of the repair and/or elteration made to the mede in eccordance with the requirement herein is true and correct to the best of		o. receral Aviation Regul	n the reverse of	or attachme it the inform	nts hereto netion					
Date 7-/5	-9 2	Signature of Aution	11th								
-1 (7.	Approval for Return To	Segrice :	1-, .							
Administrator	he authority given persons specified b of the Federal Aviation Administration	elow, the unit identified and is STAPPROVE	I in Item 4 was inspected	in the mann	er prescrib	ed by the					
	it. Standards	Inspection Authorize	Other (Speci	(y) .		<u></u>					
	Designee X Repeir Station	Person Approved by Canada Airworthines	S Group								
Date of Approva	Or Rejection Certificate or Designation N	Signal fre of Autro	ized Individual		<i>y</i> =						

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more spece is required, attech additional sheets, identify with aircraft nationality and registration mark and data work completed.) Installed YEA Door openers I.A.W. Aeronautical Accessories S.T.C. SH307650. Revised A/c weight & BALANCE. SEE AlbAT MANUAL

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1	A DESCRIPTION OF AIRCRAFT		* 1					
۲	REGISTERED OWNER		ADDRESS					
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.]	A. Operating Limitations and Markin	nos in Compliance						
X	91.31 as Applicable	The state of the same of the control			FAA Form 8130-9 (Arta			
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B. Current Operating Limitations Attached			'	Tribute and to the second of t				
7	C. Data, Drawings, Photographs, etc. (Attach when required)			Previous Airworthiness Certificate Issued in Accordance with				
·	D. Current Weight and Raisons into	ormation Available in A		. 1		7		
X	D. Current Weight and Balance Info	rmation Available in Aircraft	FAR _	4. 4 37.4	CAR	(Original Attache	d)	
X X	D. Current Weight and Balance Info E. Major Repair and Alteration, FAA F. This Inspection Recorded in Aircr	ermation Available in Aircraft Form 337 (Attach when required)	FAR _	4. 4 37.4	CAR	(Original Attache	dj	

\$ U.S. Government Printing Office: 1995-491-818/5235

DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION

'STANDARD AIRWORTHINESS CERTIFICATE

2 MANUFACTURER AND MODEL McDONNELL DOUGLAS

3 AIRCRAFT SERIAL NUMBER

CATEGORY

N510PD

HELICOPTER COMPANY 369E

0400E

AUTHORITY AND BASIS FOR ISSUANCE

This airworthmess certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which assued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthmess code as Exceptions.

NONE

TERMS AND CONDITIONS

Unless sooner surrendered, suspending airworthiness certificate is effective a accordance with Paris 21, 43, and 91 ct.

Shake

ATE OF ISSUANCE AUG 1 0 1990

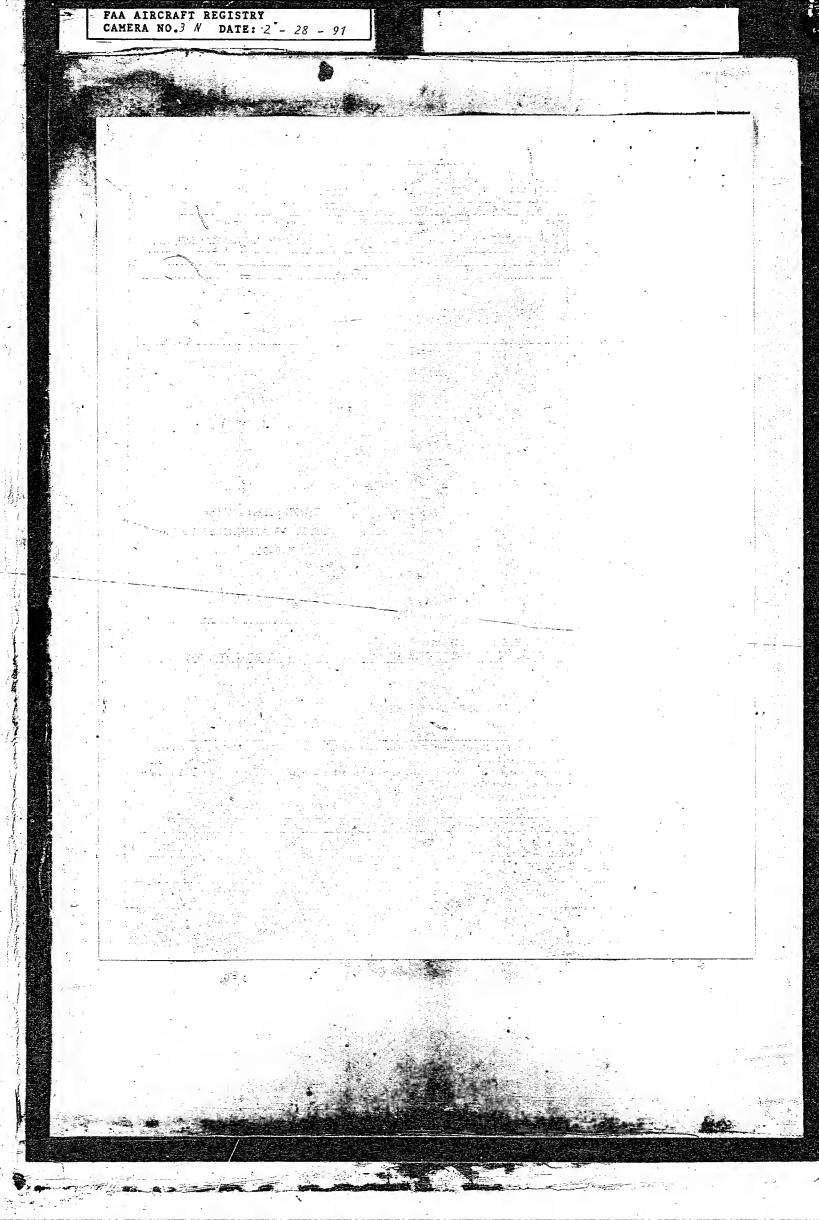
ny alteration, reproduction, or misuse of this certificate per De punishable by America exceeding \$1,000, or imprisonment not exceeding \$1,000, or AA Form 8100-2 (8-82)

FAA AIRCRAFT REGISTRY CAMERA NO. 2N DATE: 3-11-92

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1]				MODEL	<u> </u>				
1. AIRCRAFT	MCDO:	NNELL DOUGLAS 1	HELĂCO:	PTERS	1					
1 4	SEKIAL NO.					9-E				
	0400-	_ 	•	-	INATIONAL	IIY AND I	REGISTRATION	MARK		
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STATEMENT OF COMPL	DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION LIANCE WITH THE FEDERAL A	VIATION REGULAT	TIONS JULY	29, 1990
	AIRCRAFT OR AIRCRAFT CO	DMPONENT IDENTIFE	CATION	
AKE	MODEL NO. TYPE (Aimle	ne, Radio, Helicopter,	NAME OF APPLICAN	Ī
ICDONNELL-DOUGLAS		HELICOPTER	CUSTOM AIRCRA	FT INTERIORS
	LISTOF	DATA		
IDENTIFICATION		TITLE		
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1-10060 REV. "B"	FLAMMABILITY TEST P	LAN/REPORT		
DATED: 7-29-90				
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APPLICABLE REQUIPEMENT		E IN SUPPORT OF	APPLICATION 1	OK SIC
A TEICHEL REGULATION	3 (2.0)		•	•
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FAR	25.853(a)(1) AMENDMENT 2	5-32		***************************************
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CSSTIFICATION Bedans	whority vested by direction of the	Administrator and in ac	cordance with condi-	ions and limitations
of appointment under Part !	83 of the Federal Aviation Regulati	ions, data listed above	and on attached she	ets numbered
h	lave been examined in accordance w	ith established proced	ures and lound to co	aply with applicable
requirements of the Federal	Aviation Regulations.	•	:	
	Recommend approval of these data Approve these data		1	
	ED ENGINE ERÎNG REPRESENTATIVEIS	DESIGNATION NUMBE	RS(S) CLAS	SIFICATION(S)
al Pr	Bullue	NM-814		& EQUIPMENT
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FAA Form 8110-3 (11-70)	SUPERSEDES PREVIOUS EDITION		• .	
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FLAMMABILITY TEST PLAN/REPORT NO. 1-10060

REV. B

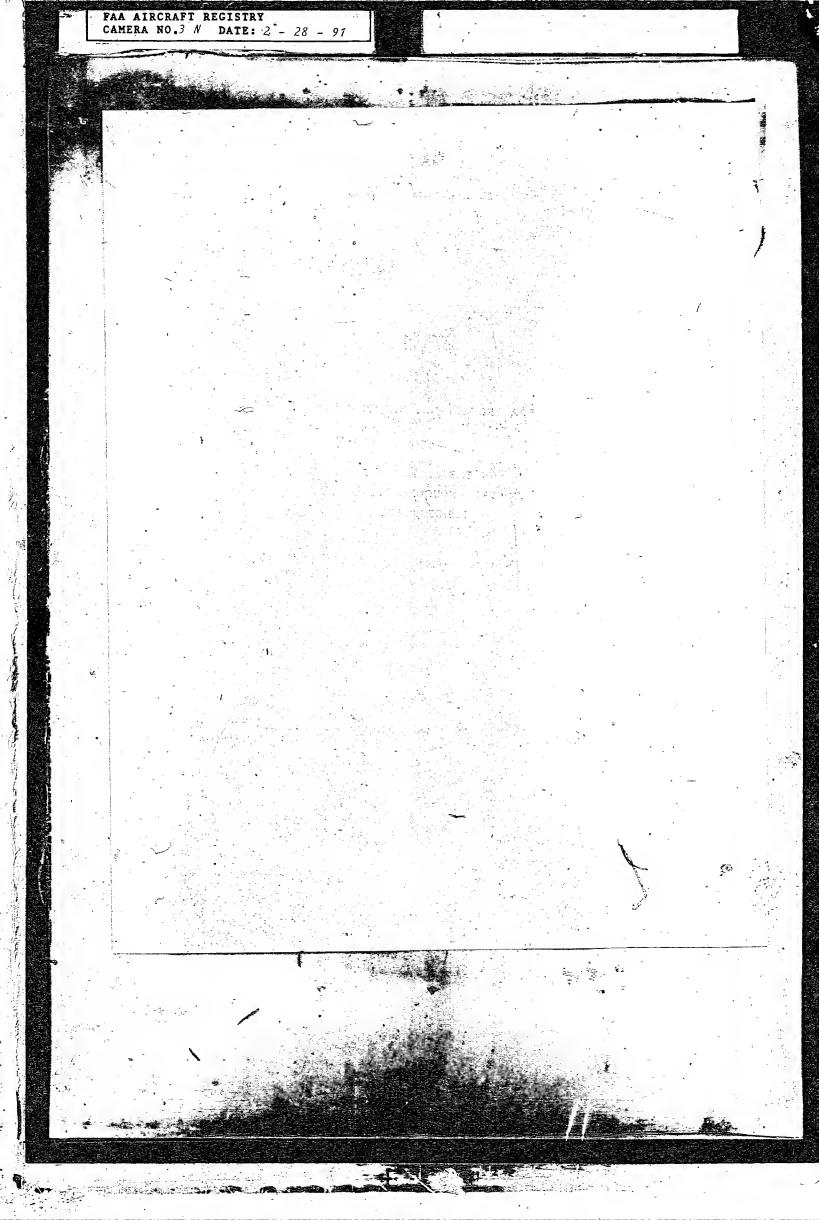
FAA PROJECT NO. NM100L-1710

FOR INSTALLATION IN:
MCDONNELL-DOUGLAS 369 SERIES
HELICOPTERS

PREPARED BY

3-15-9

3701 Industry Ave • Lakewood, CA 90712 • (213) 426-5098



REVISIONS							
REV	DESCRIPTION	DATE	APPROVAL				
A	ITEM 1 PAGE 2 AND MATERIAL DESCRIPTION PAGE 4: (PAINT WAS) AMERON 220	·7-5 90	ga .				
В	ADDED TEST RESULTS TO PAGES 4 THRU 16	7-29 90	13				
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PAGE



SCOPE 5

This Document sets forth Test Procedures to demonstrate compliance with requirements of FAR 29.853(a)(1).

REFERENCE

FAR 29.853 FAR 25 Appendix F, PART 1 SCD H5-5000 Interior Trim-Crew and Aft cabin Details.

TEST SPECIMENS

Three test specimens 3 x 12 inches of each fabric or material in each direction(WARP & FILL), if applicable, will be vertically burned per FAR 29.853(a)(1). Each specimen will be identified.

TEST WITNESS

All tests will be witnessed by a duly authorized FAA Representitive.

TEST REPORT

This Test Plan will become a Test Report upon completion of all testing and the addition of the test results.

The name and location of the test facility, the person to be contacted, phone number and requested test date are:

BUCKLEY ENTERPRISES 23428 VIA GAYO VALENCIA, CA 91355

JOHN BUCKLEY, (805)255-9842 Approx. test date: 3-26-90

•	DATA	
	TEST *	
SATION AND USAGE	USAGE	TRIM PANELS FILLER RETAINER DECOR INSULATION ATTACHMENTS RETAINER INSULATOR INSULATOR
MATERIALS IDENTIFICATION AND USAGE	MATERIAL IDENTIFICATION	PANEL: 1 PLY L501-7781 PRE- BREG, 1/8" KLEGECELL FGD55E- 03.& 1 PLY L501-7781 W/POLY- ESTER FILLER #4, ACRYLIC PRIMER, 2 COATS NASON ACRYLIC ENAMEL #419-19 WHITE; 1 COAT TEXTURE 419-19 WHITE; 1 COAT TEXTURE 419-19 WHITE; CUSTOM AIRCRAFT INTERIORS #4101 ETHAFOAM, 1/2" THICK DOW CHEMICAL, MIDLAND, MICH. 6/6 NATURAL NYLON BLOCK REGAL PLASTICS, GARDENA, CA FABRIC BACKED VINYL NAUGA- HYDE-COLOR: WHITE, UNIROYAL, MISHAWAKA, INDIANA SAME AS #4, COLOR: BLACK 2.0# POLYESTER FOAM, 1/2" THK., CHARCOAL BLOCKSOM, LOS ANGELES, CA BARXFOL/BM-1AY H.L. BLACKFORD, CORONA, CA SJ3418 2" LOOP VELCRO SCOTCH-MATE, 3M SJ3418 2" LOOP VELCRO SCOTCH-MATE, 3M SJ3418 2" PILE VELCRO IVI ENSOLITE UNIROYAL PLASTICS, MISHAWAKA, INDIANA
	IDENT. CODE	4 3 2 7 9 6 1 1 1 0 1 1 0 1 1 1 0 1 1 1 0 1 1 1 1
	FIGURE	

Page 2 REV.A

* CODE

F1-80 SECOND VERTICAL (FAR 25.853 (A) F2-12 SECOND VERTICAL (FAR 25.853 (B) F3-15 SECOND HORIZONTAL (FAR 25.853 (B-2) F4-15 SECOND HORIZONTAL (FAR 25.863 (B-3) F5-30 SECOND 46* (FAR 25.855 (A-1) F-8 30 SECOND 80* (FAR 25.1359) (D) FAA AIRCRAFT REGISTRY CAMERA NO.3 N DATE: -2

CODE F1- F2-		FIGURE NUMBER
80 SECOND	1 2	IDENT. CODE
F1-80 SECOND VERTICAL (FAR 25.853 (A) F2-12 SECOND VERTICAL (FAR 25.853 (B) F3-15 SECOND HORIZONTAL (FAR 25.853 (B-2)	CLEAR LEXAN REGAL PLASTICS, GARDENA, CA SANTOPRENE 251-80 MONSANTO CHEMICAL	MATERIAL IDENTIFICATION
F4-15 SECOND HORIZONTAL (FAR 25.853 (B-3) F5-30 SECOND 46" (FAR 25.855 (A-1) F-6 30 SECOND 60" (FAR 25.1359) (D)	EDGE TRIM	USAGE
		TEST *
		DATA SHEET

Buckley Enterprises 23428 Via Gayo Valencia, CA 91355 (805) 255-9842

PAGE 4
REVISION B

PANEL: 1- PLY L501-7781 PREPREG, 1/8" KLEGECELL FGD55E03 & 1 PLY L501-7781 WITH POLYESTER FILLER #4, ACRYLIC PRIMER 2 COATS NASON ACRYLIC ENAMEL #419-19 WHITE, 1 COAT TESTREPORTNO. MATERIAL DESCRIPTION* 1-10060 TESTNO. 549a TEXTURE 419-19 WHITE SAMPLENO. USAGE 549 TRIM PANELS MODEL/ PROGRAM MCDONNELL-DOUGLAS 369 SERIES HELICOPT MANUFACTURER SUPPLIER CUSTOM AIRCRAFT INTERIORS TEST METHOD SKETCH: IGNITION TIME-MATERIAL POSITION 60 SEC - VERT. 12 SEC - VERT. 15 SEC - HORIZ 15 SEC - HORIZ FAR 2583(a) AMENDMENT 25-32 (b) F1 FI. FI. FI. FI. FAR 25.855(a-1) FAR 25853(a) FLAME TEMP: TEST LOCATION: Valencia, CA 91355 SOAK BEGAN TEST DATE 1857°f 7-26-90 7-29-90

TEST VALU	JES .			14			•
SAMPLE NO.	TEST METHOD	EXTING TIME (SECS) .	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1 -2 -3	F1	Ø Ø Ø	5.50 5.12 5.75 5.46	Ø Ø Ø			
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PASSED PASSED	BUCKLEY X ial Type, Construction,	FAILED Specification, Thickness	Color, Exc	COMMEN	ne e	P.	

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FLAMMABILITY DATA SHEET

Buckley Enterprises 23428 Via Gayo Valencia, CA 91355 (805) 255-9842

PAGE 5
REVISION 8

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MATERIAL DESCRIPTION	1/2" MUTCK			TEST REPORT	
#4101	ETHAFOAM 1/2" THICK			TEST NO.	550a
USAGE FILLE	ER		MODEL/MCDON	SAMPLENO.	550 TIGT.AS 365
	CHEMICAL AND, MICH.			ES HELIC	
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178 12,5EL-VE	18,240-1	1		FLAME .	105786
TEST LOCATION: Valencia, CA 91355	SOAK 7-26-90 BEGAN	TEST 7-29 DATE:	-90 	TEMP:	1857°£
TEST VALUES		DRIP	TIDN' B	URN	AFTER

TEST VALL	JES	•		. · · ·			
SAMPLE NO.	TEST METHOD	EXTING TIME- (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN' TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1 -2 -3	F1	0 0	4.25 4.75 4.50	Ø Ø Ø	S- 3		
	P						
BY PASSED	JOHN BUCKLE	FAILED , Specification, Thickness	7	COMME	VIX.		e

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· PAGE REVISION TEST REPORTNO. MATERIAL DESCRIPTION* 6/6_NATURAL NYLON BLOCK 1-10060 TESTNO. 551a SAMPLENO. 551 USĄGE RETAINER MODEL MCDONNELL-DOUGLAS 369
PROGRAM
SERIES HELICOPTER MANUFACTURER SUPPLIER REGAL PLASTICS GARDENA, CA SKETCH: TEST METHOD IGNITION TIME - MATERIAL POSITION Fl. FAR 25853(a) AMENDMENT 25-32 USEC - VERT.
USEC - HORIZ.
USEC - HORIZ.
USEC - 45°
30 SEC - 45°
20 SEC - 60° ELE.
USEC - VERT. (b-3) FAR 25.855(a-1) FAR 25.1359(d) FAR 25.853(a) F4. FS FG F7. TEST DATE FLAME TEST LOCATION Valencia, CA 91355 1857°F SOAK 7-26-90 7-29-90 TEMP:

SAMPLE NO.	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1 -2	F1	Ø Ø	1.38 1.28	ø ø			
-3 .		Ø	1.12	Ø			
		, y	1.20	, ,			

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REVISION

MATERIAL DESCRIPTION* TEST REPORT NO. FABRIC BACKED VINYL, COLOR: WHITE 1-10060 TEST NO. <u>55</u>2a USAGE DECOR: SAMPLE NO. 552 MANUFACTURER SUPPLIER UNIROYAL PLASTICS MODEL/ PROGRAM MCDONNELL-DOUGLAS 369 SERIES HELICOPTER MISHAWAKA, INDIANA TESTMETHOD SKETCH: IGNITION TIME - MATERIAL POSITION 60 SEC-VERT. FAR 25853(2) AMENDMENT 25-32 USEC - VERI.
USEC - HORIZ
USEC - HORIZ
SEC - 45° FAR 25.855(a-1) FAR 25.1359(d) 30 SEC - 60" EL E TEST SOAK BEGAN TEST OCATION: Valencia, CA 91355 7-26-90 FLAME 7-29-90 1857°F

TEST VAL	UES .	_:		Θ		. ·	
SAMPLE NO.	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1 -2 -3	F1	Ø Ø	5.87 6.25 5.50	Ø Ø			
AVER.	2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	ø	5.87	Ø			
PASSED	HIN BUCKTEA	FAILED Decilication, Thickness, Co	iolor, Euc	COMMENT		• * * * * * * * * * * * * * * * * * * *	

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PAGE 8
REVISION B

MATERIAL DESCRIPTION*	FABRIC BA	CKED VIN	· // .	YDE, COLOR: B	LACK	TEST REPO	
	,			•		TESTNO.	553a
USAGE	DECOR	3.6.	4.14.2			SAMPLEN	Ю. <u>553</u>
MANUFACTURER SUPPLIER		PLASTICS A, INDIAN			PROGRAM	CDONNELL-I	OUGLAS HELICOPTER
TESTMETHOD		-		- SKETCH:	,		111111111111111111111111111111111111111
I	IGNITIONTIME-M	ATERIAL POSITIO	NC				
F2 7	60 SEC - VERT. 12 SEC - VERT. 15 SEC - HORIZ 15 SEC - HORIZ. 20 SEC - 45	FAR 25.853(a) AM (b) (b-2) (b-3) FAR 25.855(a-1)	ENDMENT 25-32				
	30 SEC - 60° ELEC 12 SEC - VERT	FAR 25.1359 (d) FAR 25.853(a)	* -15				· ,
TEST LOCATION: Valencia	.CA 91355	SOAK -	7-26-90	TEST 7-29-	90	FLAME TEMP:	1857°F

TEST VAL	UES "			-			
SAMPLE NO.	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1 -2 -3	F1 ***	Ø Ø Ø	4.25 5.25 6.38	Ø Ø Ø		-	-
AVER.		Ø	5.29	0		•••	
PASSED	OHN BUCKLEY OHN SUCKLEY Type, Construction,	FAILEIST Specification, Thickness,	Color, Eic	COMMEN	TS:		

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TEST REPORT NO. 1-10060 2.0# POLYESTER FOAM, 1/2" THICK MATERIAL DESCRIPTION TESTNO. COLOR: CHARCOAL SAMPLENO. 554 INSULATION USAGE MODEL MCDONNELL-DOUGLAS
PROGRAM 360 SERIES HELICO BLOCKSOM 369 SERIES HELICOPTER MANUFACTURER LOS ANGELES, CA SUPPLIER .. SKETCH: TESTMETHOD IGNITION TIME-MATERIAL POSITION FAR 2583(a) AMENDMENT 25-32 60 SEC - VERT. FL 12 SEC - VERT. 15 SEC - HORIZ 15 SEC - HORIZ 15 SEC - HORIZ 20 SEC - 45° 20 SEC - 60° ELL FAR 25.855 (a-1) FAR 25.1359 (d) 1857°F TEST DATE 7-29-90 SOAK < 7-26-90 TEST LOCATION: Valencia, CA 91355

TEST VALU	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCIIES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN./MIN.)	AFTER GLOW (SECS)
-1 -2 -3	F1	Ø	2.50 1.87 2.38	Ø Ø Ø			
AVER.		0					

TESTER FAILED PASSED XX

Induce Material Type, Construction, Specification, Thickness, Color, Etc.

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REVISION . TEST REPORTNO. MATERIAL DESCRIPTION* 1-10060 BARYFOL/BM-1AY TESTNO. SAMPLENO. USAGE INSULĀTION MANUFACTURER H.L. BLACKFORD CORONA, CA MODEL/ PROGRAM MCDONNELL-DOUGLAS 369 SERIES HELICOPT SKETCH: **TEST METHOD** IGNITION TIME-MATERIAL POSITION FAR 25.853(a) AMENDMENT 25-32 (b) FL 60 SEC-VERT. 12 SEC - VERT. 15 SEC - HORIZ. 15 SEC - HORIZ. 30 SEC - 45° 30 SEC - 60° ELE FA. FS. FZ. FAR 25.855(2-1) FAR 25.1359(d) FAR 25.853(a) -VERT F8 TEST DATE FLAME TEST LOCATION: Valencia, CA 91355 1857°F 7-29-90 7-26-90

TEST VAL	JES		· · · · · · · · · · · · · · · · · · ·				
SAMPLE NO.	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCIIES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1		12.3	7.5	3.5			
-2 -3	F1	3.1	4.75 5.25	1.0 ø		-	
AVER.		5.13	5.83	1.5			
BY WINESSED	OHN BUCKLEY	3.00	لها .	COMMEN	ITS:		
PASSICD *Include Mater	XX fal Type, Construction,	FAILED Specification, Thickness	11 -				, ;

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REVISION -TEST REPORT NO. 1-10060 MATERIAL DESCRIPTION TESTNO. 556a SJ3418 2" LOOP VELCRO SAMPLENO. 556 USAGE ATTACHMENTS MCDONNELL-DOUGLAS MODEL/ PROGRAM 369 SERIES HELICOPTER MANUFACTURER SUPPLIER SCOTCH-MATE, 3M SKETCH TESTMETHOD IGNITION TIME-MATERIAL POSITION FAR 2583(a) AMENDMENT 25-32 60 SEC - VERT FI IZSEC - VERT. ISSEC - HORIZ ISSEC - HORIZ (b-3) FAR 25.855(a-1) FAR 25.853(a): 30 SEC - 45° 30 SEC - 60° EL -VEX FLAME 1857°F TEST DATE: 7-29-90 SOAK BEGAN 7-26-90 TEST LOCATION: Valencia, CA 91355

SAMPLE NO.	JES TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN.MIN.)	AFTER GLOW (SECS)
-1 -2 -3 AVER.	F1	Ø Ø Ø	4.38 4.06 5.0 4.48	Ø Ø Ø	<i>A</i> 4		
TESTED BY WINESSE BY PASSED		FAILEI	//	COMME	NIS.	-	

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				-		REVISION	
				· .	4 - 2 4		
MATERIAL			····	· · · · · · · · · · · · · · · · · · ·		TESTREPO	
DESCRIPTION*	072410	? 2" PILE VI	PT.CRO	•			060
• ,	203410	5 5111P 41	3110110			TESTNO.	557a
	· .					SAMPLEN	ю. 557
USAGE ~	ATTACHM	MENTS		1. 12	MODEL/		
MANUFACTURE	R SCOTCH-	-MATE, 3M		7.7	PROGRAM	MCDONNELL-	HELICOPTER
SUPPLIER			. *			309 SEKTES	Inancor III
TEST METHO	D			SKETCH:			
	•			1	•		
- ' '	IGNITION TO	VE-MATERIAL POS	TION .				
X Fl.	60 SECVEK		AMENDMENT 25-32	┧			• .
+ FL -₹ FL	12 SEC - VER 15 SEC - HOR) -2)	-	· ·		•
F4.	15 SEC - HOR	IZ. 6	-3)	_	• • •		
175 164	30 SEC - 45	FAR 25.555(a- LEC FAR 25.1359)	d)	₫*			
F7	12 SEC - VER	T. FAR 25853(a))	<u>ប </u>			
TEST		SOAK	7-26-90	TEST	7-29-90	FLAME TEMP:	1857°F
LOCATION: Val	eneia, CA 91355	BEGAN		DATE:			
TEST VALUE	2						· · · · · · · · · · · · · · · · · · ·
TEST VALUE	- - 1			DRIP			AFTER
		EXTING	BURNED LENGTH	EXTING TIME	BURN TIME	BURN RATE	GLOW
SAMPLE NO.	TEST METHOD	TIME (SECS)	(INCHES)	(SECS)	.(SECS)	('NIM'NI)	(SECS)
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-3	1	ø	3.56	Ø		•	
AVER.	:	ø	4.02	Ø	:		
				1.00	, ~		,
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FAILED X *Include Material Type, Construction, Specification, Thickness, Color, Euc. COMMENTS:

FAA AIRCRAFT REGISTRY CAMERA NO.3 N DATE: 2 - 28 e entre

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REVISION

FLAME

7-29-90

1857°F

TEST REPORT NO. 1-10060 MATERIAL DESCRIPTION TESTNO. 558a .06 ABS PLASTIC-FR SAMPLENO. 558 USAGE RETAINER MODEL/ PROGRAM MCDONNELL-DOUGLAS MANUFACTURER REGAL PLASTICS SUPPLIER GARDENA, CA 369 SERIES HELICOPTER GARDENA, CA SKETCH TEST METHOD IGNITION TIME-MATERIAL POSITION FAR 25.853(a) AMENDMENT 25-32 60 SEC - VERT FI USEC - VERT. USEC - HORIZ. USEC - HORIZ. 30 SEC - 45° SIEC 30 SEC - 60° ELEC 12 SEC - VERT. F3. F4. FAR 25.1359 (d) FAR 25.853(a)

LOCATION	Valencia, CA 91355	BEGAN	7-20-90	DATE	7-25-50	1EMF.	
TEST VALU	JES						
SAMPLE NO.	TEST METHOD	EXTING TIME (SECS)	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN/MIN.)	AFTER GLOW (SECS)
-1	÷	12.6	3.81	ø			
-3	F1	15.8 10.0	5.48 5.18	Ø			
AVER.		12:8	4.82	. 0		, e	
TESTED BY WINESSED BY PASSED	POHN BÜCKLEN	FAILED	Oup S	∞мме	VIS.		

*Include Material Type, Construction, Specification, Thickness, Color, Etc.

SOAK BEGAN

7-26-90

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PAGE.

TEST REPORT NO. 1-10060 MATERIAL DESCRIPTION . TESTNO. 559a IVI ENSOLITE SAMPLENO: 559 INSULATION USAGE MODEL/
PROGRAM MCDONNELL-DOUGLAS MANUFACTURER UNIROYAL PLASTICS SUPPLIER MISHAWAKA INDIAN 369 SERIES HELICOPTER MISHAWAKA, INDIANA - SKETCH: TEST METHOD IGNITION TIME - MATERIAL POSITION FAR 25 853 (a) AMENDMENT 25-32 WSEC - VERT.
12 SEC - VERT.
15 SEC - HORIZ
15 SEC - HORIZ
30 SEC - 45°
30 SEC - 60° ELI FL FAR 25.855(2-1) FAR 25.1359(d) 1857°F

7-26-90

EST VALL	TEST	EXTING TIME	BURNED LENGTH (INCHES)	DRIP EXTING TIME (SECS)	BURN TIME (SECS)	BURN RATE (IN./MIN.)	AFTER GLOW (SECS)
NO.	METHOD	(SECS)	(11.0.122)				
		d	4.18	ø			·
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-2	F1	Ø	13.	ø	·		-
- 3		Ø	5.75				
		0	4.68	Ø	' '		1
AVER.				√th on the gr			1
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		H. A. A. S.		COMM	TIE.	·	`
TESTED BY	JOHN BUCKLE	Y	<u> </u>	OMM	2412.		
WINESSE	0 ()	Buly	bee				
PASSED		FAILE	D A				

TEST DATE 7-29-90

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PAGE REVISION TEST REPORT NO. 1-10060 MATERIAL DESCRIPTION TESTNO. 560a CLEAR LEXAN SAMPLENO: 560 MCDONNELL-DOUGLAS WINDOW. MODEL USAGE 369 SERIES HELICOPTER PROGRAM REGAL PLASTICS GARDENA, CA MANUFACTURER SUPPLIER SKETCH TEST METHOD IGNITION TIME - MATERIAL POSITION. OSEC VERT LISEC VERT LISEC HORIZ ISEC HORIZ 30SEC 45° 30SEC 40° ELEC LISEC VERT FAR 25.853(a) AMENDMENT 25-32 (b) X FI. F F3. FAR 25.855(a-1 FAR 25.1359 (d) FLAME 1857°F TEST TEMP: 7<u>-29-90</u> SOAK BEGAN: DATE 7-26-90 LOCATION: Valencia CA 91355 AFTER GLOW (SECS) TEST VALUES DRIP BURN RATE (INJMIN.) BURN TIME EXTING TIME (SECS) BURNED LENGTII EXTING (SECS) TIME (SECS) TEST METHOD SAMPLE NO. (INCHES) Ø 4.38 Ø -1 Ø 4.38 ø F1 -2 Ø 3.50 Ø _3 4.09 Ø AVER. COMMITME BUCKLEY BY BY FAILED PASSED XX *Include Material Type, Construction, Specification, Thickness, Color, Etc.

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Buckley Enterprises 23428 Via Gayo Valencia, CA 91355 (805) 255-9842

PAGE 16 REVISION B

MATERIAL DESCRIPTION*	- CANTY	PRENE 251-80	flar flar		i site i si			TEST REPO	ORTNO.	
	SANIC									561a
USAGE	FDGE	TRIM							SAMPLEN	561
MANUFACTUR SUPPLIER	LIP	TO CHEMICAL					MODEL/ PROGRAM	MCDO	NNELL-D	OUGLAS
SOFFLIER							<u> </u>	59 SE	RIES HE	LICOPTER
TEST METH	OD	·			SKETCH:					
	IGNITIONTI	ME-MATERIAL POS	TTION		A					
X FL	60 SECVER		AMENDMENT 25-32						• •	
F2.	12 SEC - VER 12 SEC - HOI 15 SEC - HOR	UZ. 6	2)	7		· ·	•			200
FS FS	30 SEC45°	FAR25855(2-	1)	-					7	entre in Asi
F7.	12 SEC - VER			5					'	
TEST		SOAK BEGAN:	7–26–90		TEST DATE:	7–29-	-90		FLAME TEMP:	1857°F
LOCATION	/akmcia, CA 91355	1 BOOALC								
TEST VALU	TES				- tn			1		1
SAMPLE NO.	TEST METIIOD	EXTING TIME (SECS)	BURNED LENGTH (INCHIS)	EX	RIP TING IME ECS)	T	URN IME ECS)	R	URN ATE /MIN.)	AFTER GLOW (SECS)
		ø	1.75	1.	Ø				ı	
-1 -2	F1	Ø	2.12		ø					
-3	•	ø	2.38		ø	. د				
AVER.		ø	2.08-		Ø .					
							· ·	-	. •	
						-				
BY I	OHN BUCKTE	3.00	w		COMME	strs:				
PASSED	\boxtimes	FAILED				•				
*Include Materi	ial Type, Construction	Specification, Thickness	, Color, Each							

DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY OFFICE DENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. McDonnell Douglas Helicopter Co. 369E 1. AIRCRAFT NATIONALITY AND REGISTRATION MARK SERIAL NO. F0400E) N510PD NAME (As shown on registration certificate)
McDonnell Douglas Helicopter Co. ADDRESS (As shown on registration certificate) 2. OWNER 5000 E. McDowell Road 85205 Mesa, AZ 3. FOR FAA USE ONLY

	4. UNIT IDENTIFICATION.							
UNIT	MAKE	MODEL	MODEL : SERIAL NO.					
AIRFRAME .	***************************************	Nees (As described in item 1 above)	····		х			
POWERPLANT								
PROPELLER	a .		• ٤	-				
APPLIANCE	MANUFACTURER (177)							

	CONFORMITY STATEMENT	10
A. AGENCY'S NAME AND ADDRESS	B. KIND OF AGENCY	C. CERTIFICATE NO.
Flight Trails Helicopters, Inc.	U.S. CERTIFICATED MECHANIC	
4805 E. Falcon Drive	FOREIGN CERTIFICATED MECHANIC	
Mesa, AZ 85205	X CERTIFICATED REPAIR STATION	407-40
	MANUFACTURER	407-40

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Sep		mber 5, 19	90		ELLEND N. BREM	enerthon
	<u> </u>				ROVAL FOR RETURN TO SERVICE	
Pui the	Adr	nt to the authority ministrator of the F	gi ver edera	n persons specified hel al Aviation Administra	ow, the unit identified in item tion and is X APPROVED	4 was inspected in the manner prescribed by REJECTED
ВУ	FAA FLT. STANDARDS INSPECTOR		MANUFACTURER	INSPECTION AUTHORIZATION	OTHER (Specify)	
		FAA DESIGNEE	X	REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE REJEC		APPROVAL OR		CERTIFICATE OR DESIGNATION NO.	SIGNATURE OF AUTHORIZ	ED INDIVIDUAL merthon
		/90		407-40	ELLEND N. BREM	ERTHON
		007	. —			

FAA Form 337 (7-67)

(8320)

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) Installed Aeronautical Accessories, Inc. Rain Gutters per STC SH1626SO dated 6/13/84, revised 6/1/88 and drawings and installation instructions dated 3/29/88, see 337. Installed Facet Enterprises Scavenge Oil Filter per STC SH401GL dated 4/18/80, amended 6/10/88 and installation instructions No. 1741050 dated 1/1/83, revised 5/24/88.

dated 1/1/83, revised 5/24/88.

Installed WECO Starter Generator Kit per STC SH3195NW dated 12/12/85, and WECC Master Drawings No. WE369 Rev. C dated 12/3/85. All work done in accordance with AC 43.13-1 & -2, No weight change.

Installed Wire Strike Protection per STC SH1713SO dated 12/12/84, revised June 1, 1988 and ER84650 installation instruction dated 2/24/84, revised 10/4/86 and 10/4/89 and drawing 44183053 and 44183048 and 44185049.

Installed Fargo Aux. Tank per STC SH656GL dated 11/12/82, revised 7/30/85 and Fargo Drawings A1066-1082 dated 10/25/82 and installation manual data. 11/5/82, amended 6/2/86.

All work completed in accordance with AC 43.13-1A & 2A.

Weight & balance revised. See Pilot's Flight Manual.

Weight & balance revised, See Pilot's Flight Manual.

ADDITIONAL SHEETS ARE ATTACHED

1975-G.P.O.-1703-M/673-900/175